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(19) **United States**(12) **Patent Application Publication**
WATANABE(10) **Pub. No.: US 2017/0292388 A1**(43) **Pub. Date: Oct. 12, 2017**(54) **CERAMIC MATRIX COMPOSITE MEMBER
AND METHOD OF MANUFACTURING THE
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(57)

ABSTRACT

A ceramic matrix composite member used as a turbine blade includes a principal part forming a blade part and a dovetail part, and a subordinate part forming a platform part. A principal fiber in a ceramic fiber fabric forming the principal part is a continuous fiber. An extension direction of the principal fiber is in parallel with a direction in which stress is applied. The ceramic fiber fabrics respectively forming the principal part and the subordinate part are joined together and formed into an integrated three-pronged fiber fabric. The ceramic fiber fabric forming the principal part and the ceramic fiber fabric forming the subordinate part are integrated together by being set into a mold with the ceramic fiber fabric forming the subordinate part folded at a desired angle to the ceramic fiber fabric forming the principal part. Then, a ceramic matrix is formed in the obtained molded body.

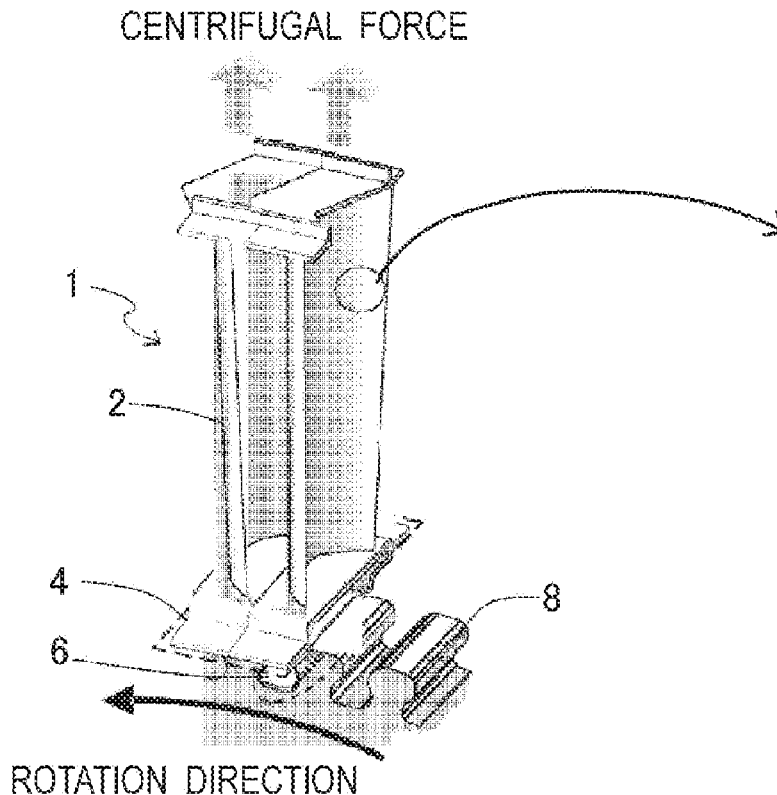


FIG. 1A

CENTRIFUGAL FORCE

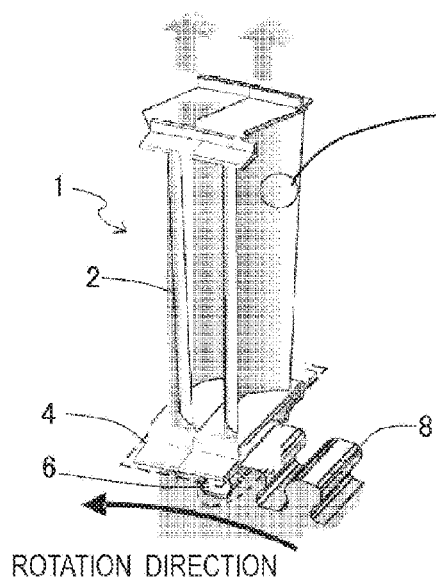


FIG. 1B

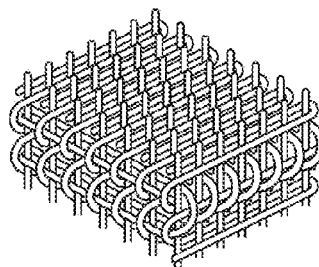


FIG. 2A

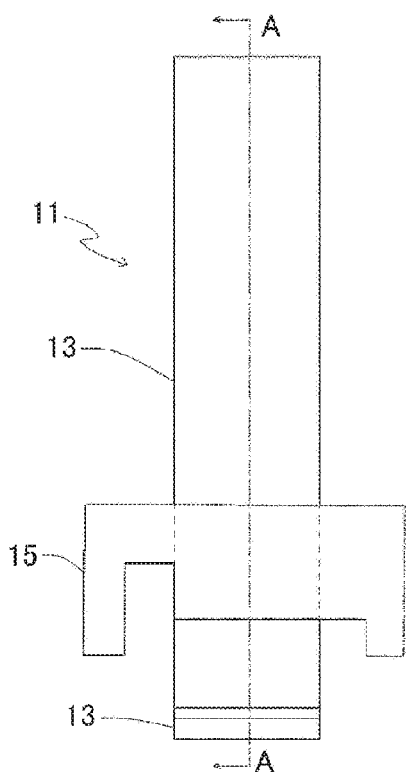


FIG. 2B

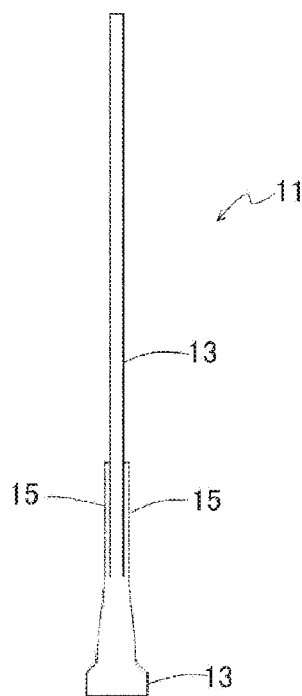


FIG. 3A

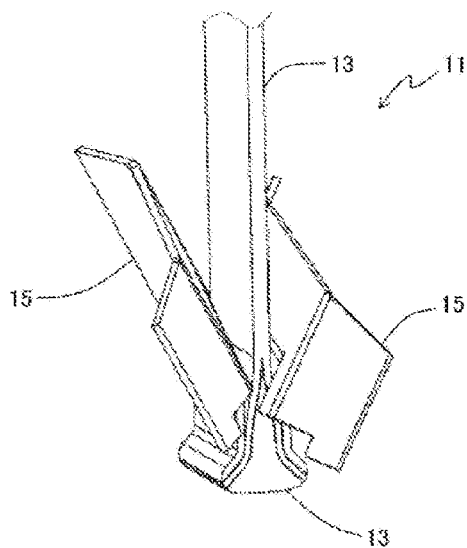


FIG. 3B

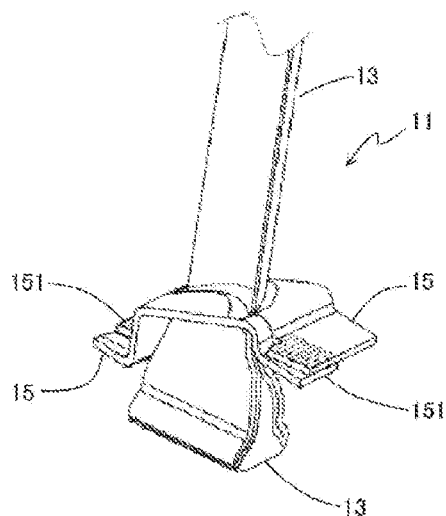


FIG. 4A

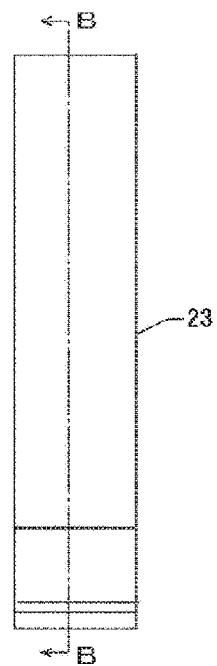


FIG. 4B

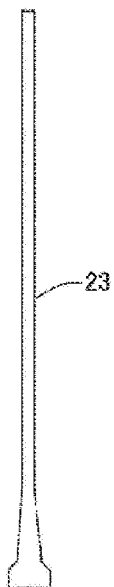


FIG. 4C

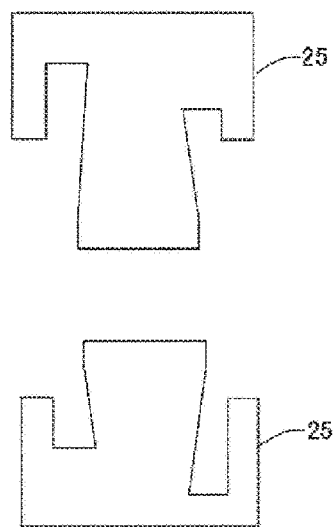


FIG. 5A

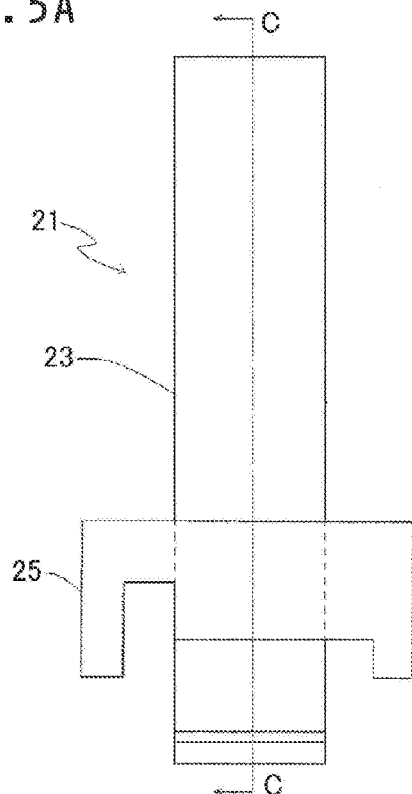


FIG. 5B

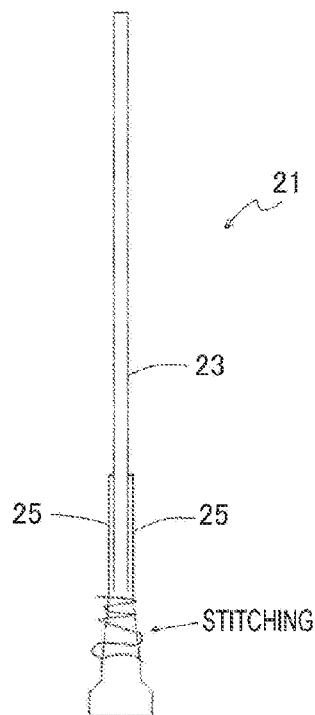


FIG. 6A

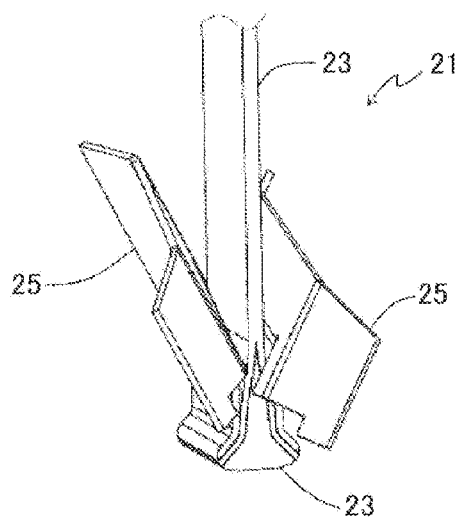


FIG. 6B

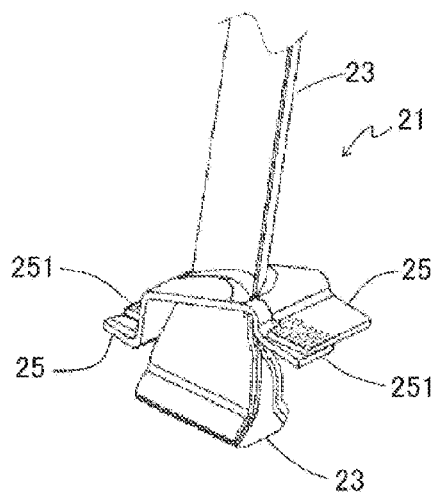


FIG. 7A

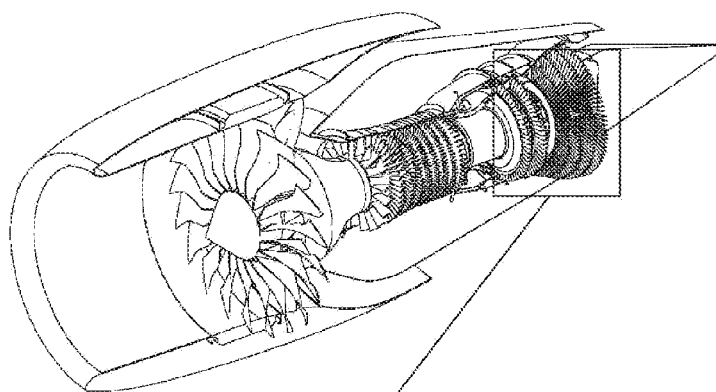
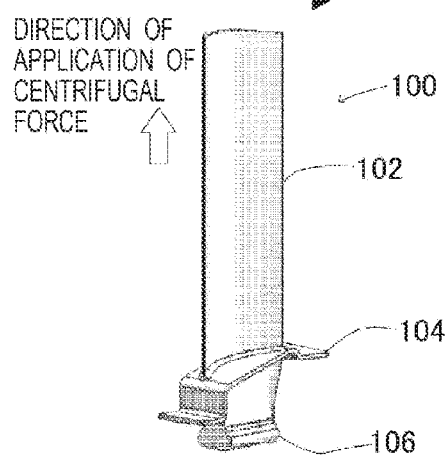


FIG. 7B



CERAMIC MATRIX COMPOSITE MEMBER AND METHOD OF MANUFACTURING THE SAME

CROSS REFERENCE TO RELATED APPLICATIONS

[0001] This application is a divisional of U.S. application Ser. No. 14/250,044 filed Apr. 10, 2014, which is a continuation application of International Application No. PCT/JP2012/076160 filed on Oct. 10, 2012, which claims priority to Japanese Patent Application No. 2011-227221 filed on Oct. 14, 2011, the entire contents of each of which are incorporated by reference herein.

BACKGROUND OF THE INVENTION

1. Field of the Invention

[0002] The present invention relates to a ceramic matrix composite member and a method of manufacturing the same.

2. Description of the Related Art

[0003] Jet engines and the like include components which are subjected to large stress applied in a particular direction, such as centrifugal force, while in use. Requirements for such components include particularly high strength as well as heat resistance. For this reason, the components are usually made from metallic materials. FIG. 7A is a schematic cross-sectional view of a generally-used aircraft turbo-fan engine. FIG. 7B is a schematic view magnifying one of turbine blades of the turbo-fan engine shown in FIG. 7A. While the engine is in operation, large centrifugal force is applied to the turbine blade in a longitudinal direction (radial direction) of a blade part. For this reason, the turbine blades are usually made from a Ni-based alloy or the like. As shown in FIG. 7B, a turbine blade **100** includes: a blade part **102**; a platform part **104** extending in a direction vertical to a principal surface of the blade part **102**; and a dovetail part **106** placed at one end portion of the blade part **102**. In this manner, the turbine blade **100** has a relatively complicated shape. Nevertheless, turbine blades can be manufactured with relative ease by casting a metallic material such as a Ni-based alloy.

[0004] Ceramic matrix composite members are formed from ceramic matrix composites (CMCs) each containing ceramic fibers and a ceramic matrix. In recent years, there are growing expectations that ceramic matrix composite members will be applied to jet engine components. Ceramic matrix composite members are light in weight and excellent in heat resistance. For this reason, if ceramic matrix composite members can be used as jet engine components, it is expected that the weight and fuel consumption rate of an engine will be reduced.

[0005] U.S. Pat. No. 7,510,379, U.S. Pat. No. 7,094,021 and U.S. Pat. No. 4,650,399 have proposed a jet engine component to which ceramic matrix composites are applied, and a method of manufacturing the component. To put it specifically, these patent documents describe turbine blades formed from ceramic matrix composite members. In these turbine blades, the blade part is joined to the platform part by use of a matrix, a brazing material or the like.

SUMMARY OF THE INVENTION

[0006] The conventional turbine blade using the ceramic matrix composite member break down in some cases, because the turbine blade cannot withstand large stress (centrifugal force or the like) applied in a particular direction. The breakdown tends to start, particularly, at a portion where the parts are joined.

[0007] An object of the present invention is to solve the foregoing problem. To put it specifically, an object of the present invention is to provide a ceramic matrix composite member, which withstands large stress (centrifugal force or the like) applied in a particular direction and is accordingly less likely to break down when used as a turbine blade.

[0008] A first aspect of the present invention provides a ceramic matrix composite member to be used as a turbine blade, which includes: a principal part forming a blade part and a dovetail part; and a subordinate part forming a platform part. Here, a principal fiber in a ceramic fiber fabric forming the principal part is a continuous fiber. A direction of the principal fiber is in parallel with a direction in which stress is applied. The ceramic fiber fabric forming the principal part and a ceramic fiber fabric forming the subordinate part are joined together to form an integrated three-pronged fiber fabric. The ceramic fiber fabric forming the principal part and the ceramic fiber fabric forming the subordinate part are integrated together by being set into a mold with the ceramic fiber fabric forming the subordinate part folded at a desired angle to the ceramic fiber fabric forming the principal part. Then, a ceramic matrix is formed in the obtained molded body.

[0009] A second aspect of the present invention provides a ceramic matrix composite member to be used as a turbine blade, which includes: a principal part forming a blade part and a dovetail part; and a subordinate part forming a platform part. Here, a principal fiber in a ceramic fiber fabric forming the principal part is a continuous fiber. A direction of the principal fiber is in parallel with a direction in which stress is applied. The ceramic fiber fabric forming the principal part and a ceramic fiber fabric forming the subordinate part are joined together by stitching. Thereafter, the joined ceramic fiber fabrics are integrated together by being set into a mold with the ceramic fiber fabric forming the subordinate part folded at a desired angle to the ceramic fiber fabric forming the principal part. Then, a ceramic matrix is formed in the obtained molded body.

[0010] The present invention can provide a ceramic matrix composite member, which withstands large stress (centrifugal force or the like) applied in a particular direction and is accordingly less likely to break down when used as a turbine blade.

BRIEF DESCRIPTION OF THE DRAWINGS

[0011] FIG. 1A is a schematic perspective view of a turbine blade of first and second embodiments of the present invention, and FIG. 1B is a schematic perspective view showing an example of a structure of a ceramic fiber fabric forming the turbine blade.

[0012] FIGS. 2A and 2B show an integrated three-pronged fiber fabric of the first embodiment of the present invention, in which FIG. 2A is a schematic side view of the integrated three-pronged fiber fabric and FIG. 2B is a cross-sectional view taken along the A-A line of FIG. 2A.

[0013] FIGS. 3A and 3B are schematic perspective views for explaining how the integrated three-pronged fiber fabric of the first embodiment of the present invention is deformed.

[0014] FIG. 4A is a schematic side view of a fiber fabric in a principal part of a second embodiment of the present invention, FIG. 4B is a cross-sectional view taken along the B-B line of FIG. 4A, and FIG. 4C is a schematic front view of two platform parts.

[0015] FIG. 5A is a schematic side view of the fiber fabric in the principal part of the second embodiment of the present invention, and FIG. 5B is a cross-sectional view taken along the C-C line of FIG. 5A.

[0016] FIGS. 6A and 6B are schematic perspective views for explaining how the fiber fabric of the second embodiment of the present invention is deformed.

[0017] FIG. 7A is a schematic perspective view of a generally-used aircraft turbo-fan engine, and FIG. 7B is a schematic diagram magnifying one of turbine blades of the turbo-fan engine.

DESCRIPTION OF THE PREFERRED EMBODIMENTS

[0018] The present invention has been accomplished based on the following findings. A ceramic matrix composite member includes a ceramic fiber fabric forming a principal part of the ceramic matrix composite member. A principal fiber in the ceramic fiber fabric extends in a predetermined direction (the direction will be referred to as an extension direction for the sake of explanatory convenience). When the ceramic matrix composite member is used as a turbine blade, the ceramic matrix composite member is subjected to stress, such as centrifugal force associated with its rotation. In this regard, it has been found that even if the stress is large, the ceramic matrix composite member is less likely to break down when the extension direction of the principal fiber is in parallel with the direction of the stress. Furthermore, in a case where multiple parts, which will later form the ceramic matrix composite member as the turbine blade, are molded individually and joined together after a matrix is formed with the multiple parts, the ceramic matrix composite member is more likely to break down from joining portions thus formed. In contrast, it has been found that: when the multiple parts are formed integrally and the matrix is formed in the integrally molded body, bonds among fibers of the parts become stronger and the strength becomes larger as a whole; and accordingly, the ceramic matrix composite member becomes less likely to break down.

[0019] Descriptions will be hereinbelow provided for embodiments of the present invention. Each embodiment represents a ceramic matrix composite member to be used as a turbine blade. FIG. 1A is a schematic perspective view of a turbine blade 1 of the embodiments of the present invention. FIG. 1B is a schematic perspective view showing an example of a structure of a ceramic fiber fabric forming the turbine blade. As shown in FIG. 1A, the external shape of the turbine blade 1 may be the same as that of a generally-used turbine blade (shown in FIGS. 7A and 7B, for example).

[0020] The turbine blade 1 includes: a blade part 2 and a dovetail part 6 collectively forming a principal part; and platform parts 4 forming a subordinate part to the principal part. Each platform part 4 extends in a direction perpendicular to a principal surface of the blade part 2. The dovetail part 6 is placed at one of two end portions of the blade part 2. As shown in FIG. 1A, the dovetail part 6 is placed in an

end portion closer to the center of rotation of the turbine blade 1, for example. In this case, as shown in FIG. 1A, the dovetail part 6 is fitted in a disk part 8. While the turbine blade 1 is in use, the disk part 8 rotates. This rotation applies strong centrifugal force to the turbine blade 1 in a longitudinal direction (radial direction) of the blade part 2.

[0021] As shown in FIG. 1B, a ceramic fiber fabric forming the turbine blade 1 has a three-dimensional structure. The fiber fabric having the three-dimensional structure can be obtained by: tying hundreds to thousands of ceramic fibers into a fiber bundle; and then weaving such fiber bundles in the X-, Y-, and Z-directions. To put it specifically, the fiber fabric having the three-dimensional structure can be obtained by: stacking multiple layers each obtained by arranging fiber bundles in the X-direction and in the Y-direction perpendicular to the X-direction; and stitching the multiple layers with other fiber bundles in the thickness direction of the multiple layers (in the Z-direction). It should be noted that the ceramic fiber fabric of the embodiments does not have to adopt the three-dimensional structure, or may partially adopt the three-dimensional structure.

[0022] The ceramic fiber fabric forming the principal part (i.e., the blade part 2 and the dovetail part 6) includes a principal fiber. In the embodiments, the extension direction of the principal fiber is almost in parallel with the direction of application of the centrifugal force. In this respect, the principal fiber in the ceramic fiber fabric means a group of fibers extending in a particular direction which are among the fibers included in the fiber fabric. In addition, the particular direction means the X-direction, the Y-direction or the Z-direction, for example, if the ceramic fiber fabric has the three-dimensional structure. For this reason, in this case, the principal fiber is any one of the fiber bundles in the X-direction, the fiber bundles in the Y-direction and the bundles in the Z-direction. As described above, when the direction of the centrifugal force applied to the turbine blade is almost in parallel with the direction of the principal fiber (the X-direction, the Y-direction or the Z-direction), the turbine blade is less likely to break down even if the turbine blade is subjected to large centrifugal force.

[0023] Furthermore, the principal fiber in the ceramic fiber fabric forming the principal part may be formed from a continuous fiber. In other words, the principal fiber is not cut out from one end portion to another end portion of the principal part. In this case, the principal fiber may be a monofilament formed only from one continuous fiber, or a multifilament formed from a bundle of continuous fibers. When the principal fiber extends in the direction almost in parallel with that of the centrifugal force, and concurrently when the principal fiber is formed from the continuous fiber, the principal fiber is less likely to break down even if the principal fiber is subjected to large centrifugal force. As a consequence, the principal part as a whole becomes less likely to break down and durable for use.

[0024] A first embodiment of the present invention provides a ceramic matrix composite member to be used as a turbine blade, which includes: a principal part forming a blade part and a dovetail part; and a subordinate part forming platform parts. A principal fiber in a ceramic fiber fabric forming the principal part is a continuous fiber. The direction of the principal fiber is in parallel with a direction in which centrifugal force is applied. The ceramic fiber fabric forming the principal part and the ceramic fiber fabric forming the subordinate part are unified into an integrated three-pronged

fiber fabric. The ceramic matrix composite member is manufactured by: integrating the two ceramic fiber fabrics together by setting the ceramic fiber fabric forming the principal part and the ceramic fiber fabric forming the subordinate part into a mold with the ceramic fiber fabric forming the subordinate part folded at a desired angle to the ceramic fiber fabric forming the principal part; and forming a ceramic matrix in the obtained molded body.

[0025] In addition, a second embodiment of the present invention is a ceramic matrix composite member to be used as a turbine blade, which includes: a principal part forming a blade part and a dovetail part; and a subordinate part forming platform parts. A principal fiber in a ceramic fiber fabric forming the principal part is a continuous fiber. The direction of the principal fiber is in parallel with a direction in which centrifugal force is applied. The ceramic matrix composite member is manufactured by: joining the ceramic fiber fabric forming the principal part and the ceramic fiber fabric forming the subordinate part by stitching; then integrating the joined ceramic fiber fabrics together by setting the joined ceramic fiber fabrics into a mold with the ceramic fiber fabric forming the subordinate part folded at a desired angle to the ceramic fiber fabric forming the principal part; and forming a ceramic matrix in the obtained molded body.

[0026] Descriptions will be provided for the first embodiment by use of FIGS. 2A to 3B. FIGS. 2A and 2B are diagrams illustrating an integrated three-pronged fiber fabric **11** in which a ceramic fiber fabric **13** forming the blade part and the dovetail part (the principal part), and ceramic fiber fabrics **15** forming the platform parts (the subordinate part) are joined together. FIG. 2A is a schematic side view of the integrated three-pronged fiber fabric **11** and FIG. 2B is a cross-sectional view taken along the A-A line of FIG. 2A.

[0027] In the first embodiment, after the integrated three-pronged fiber fabric **11** is obtained, as shown in FIG. 3A, the ceramic fiber fabrics **15** forming platform parts **4** (see FIG. 1A) are folded at a desired angle to the ceramic fiber fabric **13** forming a blade part **2** (see FIG. 1A) and a dovetail part **6** (see FIG. 1A) (at almost 90 degrees in the case of the turbine blade). Thus, the fiber fabric in the aspect shown in FIG. 3B is obtained. Thereafter, overlapping portions **151** of the two fiber fabrics **15** each forming the platform part **4** are stitched to each other by use of other fiber bundles. This stitching makes it possible to further increase strength of the ceramic fiber fabric.

[0028] Here, the integrated three-pronged fiber fabric **11** can be manufactured by a conventional publicly-known method, for example. For instance, the integrated three-pronged fiber fabric in a desired shape can be obtained by: tying hundreds to thousands of ceramic fibers into a fiber bundle; and then weaving such fiber bundles in the X-, Y-, and Z-directions.

[0029] Furthermore, no specific restriction is imposed on the material quality, thickness or the like of the ceramic fibers. For example, ceramic fibers made of SiC, C, Si₃N₄, Al₂O₃, BN and the like may be used. Moreover, the thickness of the ceramic fibers may be the same as that of conventional publicly-known fibers. The thickness may be in a range from several micrometers to several tens of micrometers, for example.

[0030] After the fiber fabric in the aspect shown in FIG. 3B is obtained, the obtained fiber fabric is set into a mold (not illustrated) and integrated together. The mold has an internal shape corresponding to the shape of the desired

molded body, and is dividable into an appropriate number (6, for example) of pieces. The fiber fabric is set into the mold while changing its shape to the mold. Thus, the fiber fabric can be integrated together in the mold. Thereafter, the ceramic matrix is formed in the obtained molded body. The forming of the ceramic matrix can be achieved, for example, by a method using a chemical reaction of a gas, a method using sintering of solid power, and the like. For example, the matrix is precipitated in a surface of the molded body through a chemical reaction by exposing the molded body, integrated in the mold, to a material gas in a chamber. Alternatively, the integrated molded body is dipped into slurry of material power solids, and is then subjected to sintering. Other methods are also applicable.

[0031] Next, descriptions will be provided for the second embodiment by use of FIGS. 4A to 6B.

[0032] FIGS. 4A, 4B and 4C are diagrams illustrating a fiber fabric **23** forming the blade part and the dovetail part (the principal part), as well as fiber fabrics **25** forming the platform parts (the subordinate part). FIG. 4A is a schematic side view of the fiber fabrics **23**, **25**. FIG. 4B is a cross-sectional view taken along the B-B line of FIG. 4A. FIG. 4C is a schematic front view of two platform parts.

[0033] In the second embodiment, after the fiber fabric **23** and the fiber fabrics **25** as shown in FIGS. 4A to 4C are obtained, the fiber fabric **21** in a shape shown in FIGS. 5A and 5B is obtained by joining the fiber fabric **23** and the fiber fabrics **25** together by stitching. FIG. 5A is a schematic side view of the fiber fabric **21** obtained by stitching. FIG. 5B is a cross-sectional view taken along the C-C line of FIG. 5A. In the aspect shown in FIGS. 5A and 5B, the fiber fabric **23** and the fiber fabrics **25** are joined together by being stitched in the vicinity of the dovetail part, as shown in FIG. 5B.

[0034] Subsequently, as shown in FIG. 6A, the fiber fabrics **25** forming the platform parts are folded at the desired angle to the fiber fabric **23** forming the blade part and the dovetail part (at 90 degrees in the case of the turbine blade). Thus, the fiber fabric in the aspect shown in FIG. 6A is obtained.

[0035] After the fiber fabric in the aspect shown in FIG. 6B is obtained, it is desirable that overlapping portions **251** of the two fiber fabrics **25** each forming the platform part be stitched together by use of other fiber bundles, because the strength of the obtained turbine blade of the present invention is further increased.

[0036] Here, no specific restriction is imposed on the method of manufacturing the fiber fabric **23** and the fiber fabrics **25**. The fiber fabric **23** and the fiber fabrics **25** may be manufactured by use of a conventional, publicly-known method, for example. For instance, each fiber fabric in the desired shape can be obtained by: tying hundreds to thousands of ceramic fibers into a fiber bundle; and then weaving such fiber bundles in the X-, Y-, and Z-directions.

[0037] Furthermore, no specific restriction is imposed on the material quality, thickness or the like of the ceramic fibers. For example, ceramic fibers made of SiC, C, Si₃N₄, Al₂O₃, BN and the like may be used. Moreover, the thickness of the ceramic fibers may be the same as that of conventional publicly-known fibers. The thickness may be in a range from several micrometers to several tens of micrometers, for example.

[0038] After the fiber fabric in the aspect shown in FIG. 6B is obtained as described above, the obtained fiber fabric is set into the mold and integrated together. The shaping and

the subsequent forming of the ceramic matrix are the same as the process for forming the ceramic matrix composite member of the first embodiment. To put it specifically, the ceramic matrix composite member is similarly manufactured by: setting the fiber fabrics into the divided mold integrating the fiber fabrics together; and then forming the ceramic matrix in the surface of the molded body by use of the foregoing method.

[0039] It should be noted that: the present invention is not limited to the above-described embodiments; the present invention is specified by the description in the scope of claims; and the present invention further includes all modifications which fall within the range and meanings equivalent to the description of the scope of claims.

What is claimed is:

1. A ceramic matrix composite member to be used as a turbine blade, comprising:

a principal part forming a blade part and a dovetail part; and

a subordinate part forming a platform part, wherein

a principal fiber in a ceramic fiber fabric forming the principal part is a continuous fiber,

a direction of the principal fiber is in parallel with a direction in which stress is applied,

the ceramic fiber fabric forming the principal part and a ceramic fiber fabric forming the subordinate part are joined together to form an integrated three-pronged fiber fabric,

the ceramic fiber fabric forming the principal part and the ceramic fiber fabric forming the subordinate part are integrated together by being set into a mold with the ceramic fiber fabric forming the subordinate part folded at a desired angle to the ceramic fiber fabric forming the principal part, and

a ceramic matrix is formed in the obtained molded body.

* * * * *