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(54) **SYSTEM AND METHOD FOR CONTROLLING THE COMBUSTION PROCESS IN A GAS TURBINE OPERATING WITH EXHAUST GAS RECIRCULATION**

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(57) **ABSTRACT**

A system includes a fuel control system configured to control a fuel flow to one or more combustors and an oxidant control system configured to control an oxidant flow to each combustor of the one or more combustors, wherein the oxidant flow is configured to at least partially react with the fuel flow within the one or more combustors to form an exhaust gas flow. The system also includes an exhaust gas system configured to direct a recirculation flow of the exhaust gas flow to each combustor of the one or more combustors; and a controller coupled to the fuel control system, the oxidant control system, and the exhaust gas system. The controller is configured to independently control a fuel-to-oxidant ratio and an exhaust gas-to-oxidant ratio. The FOR is the fuel flow divided by the oxidant flow, and the EGOR is the recirculation flow divided by the oxidant flow.

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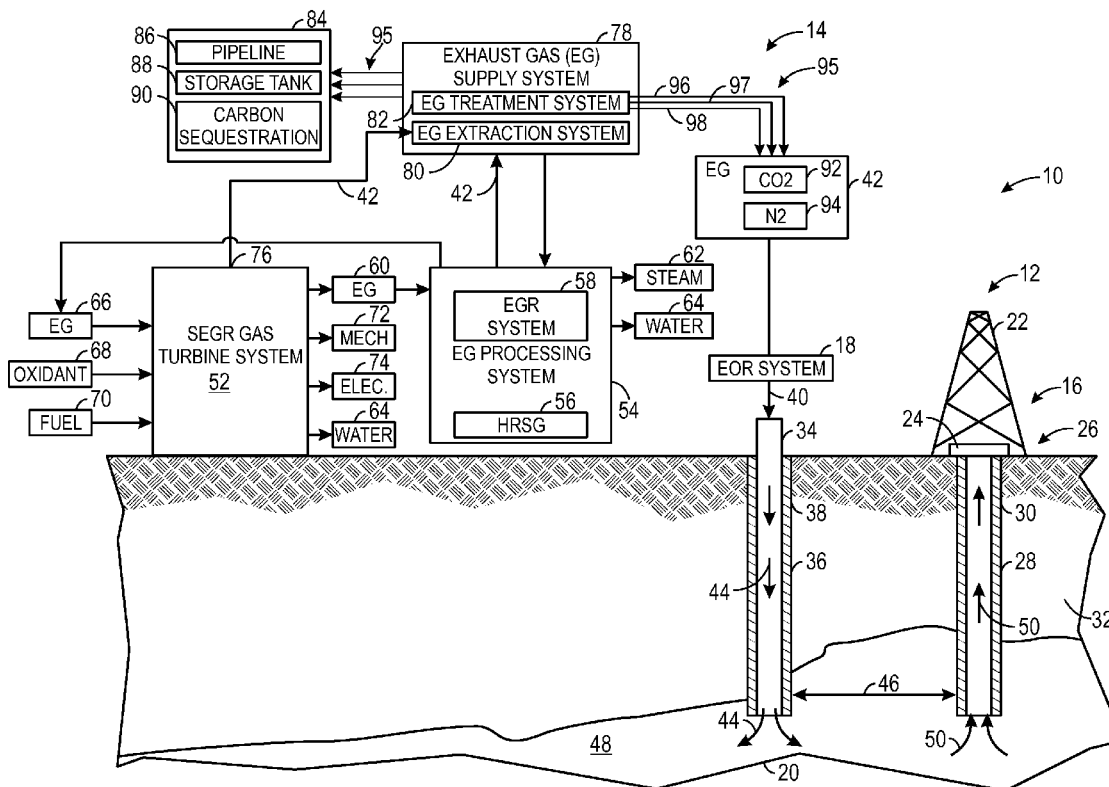
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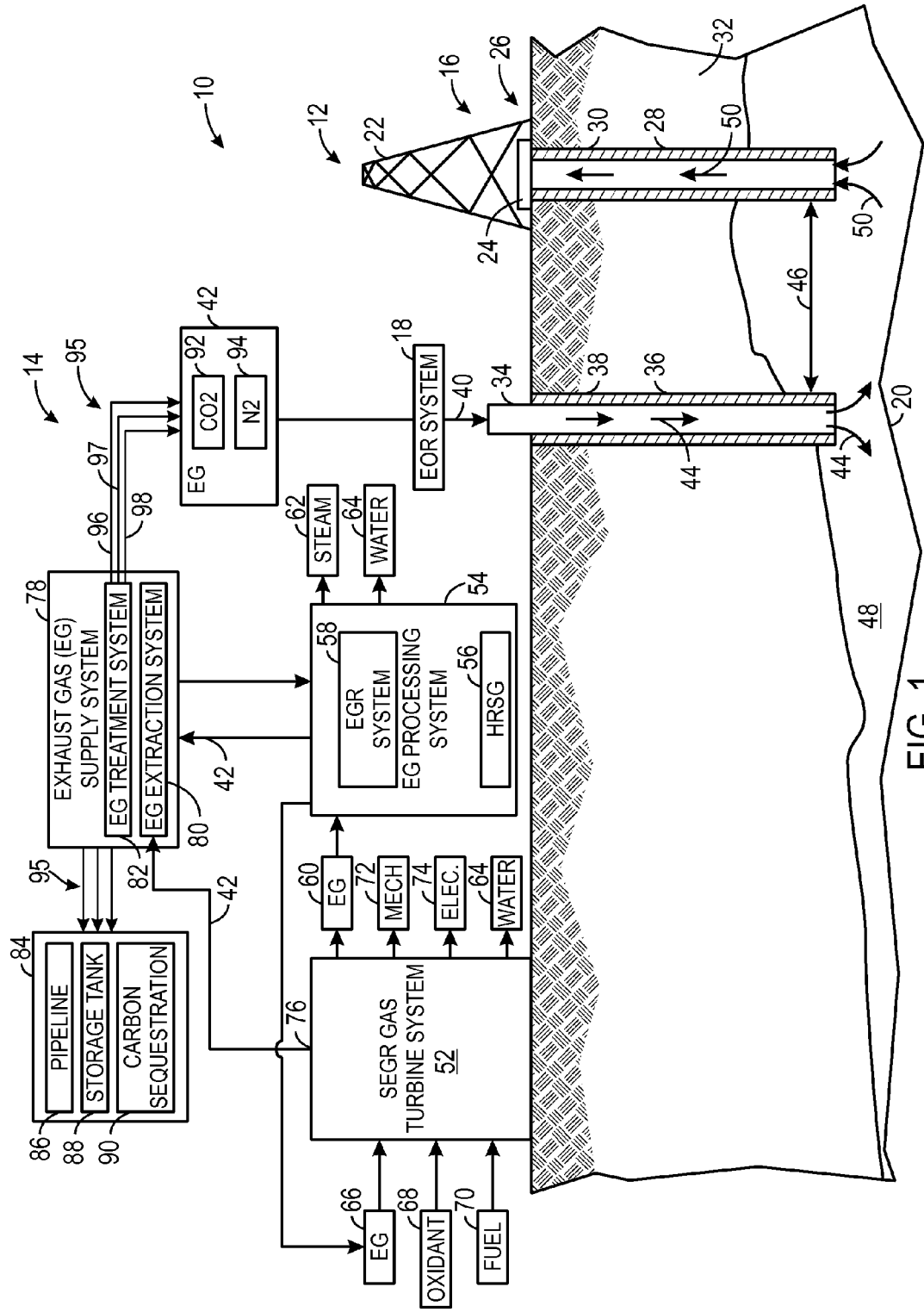


FIG. 1

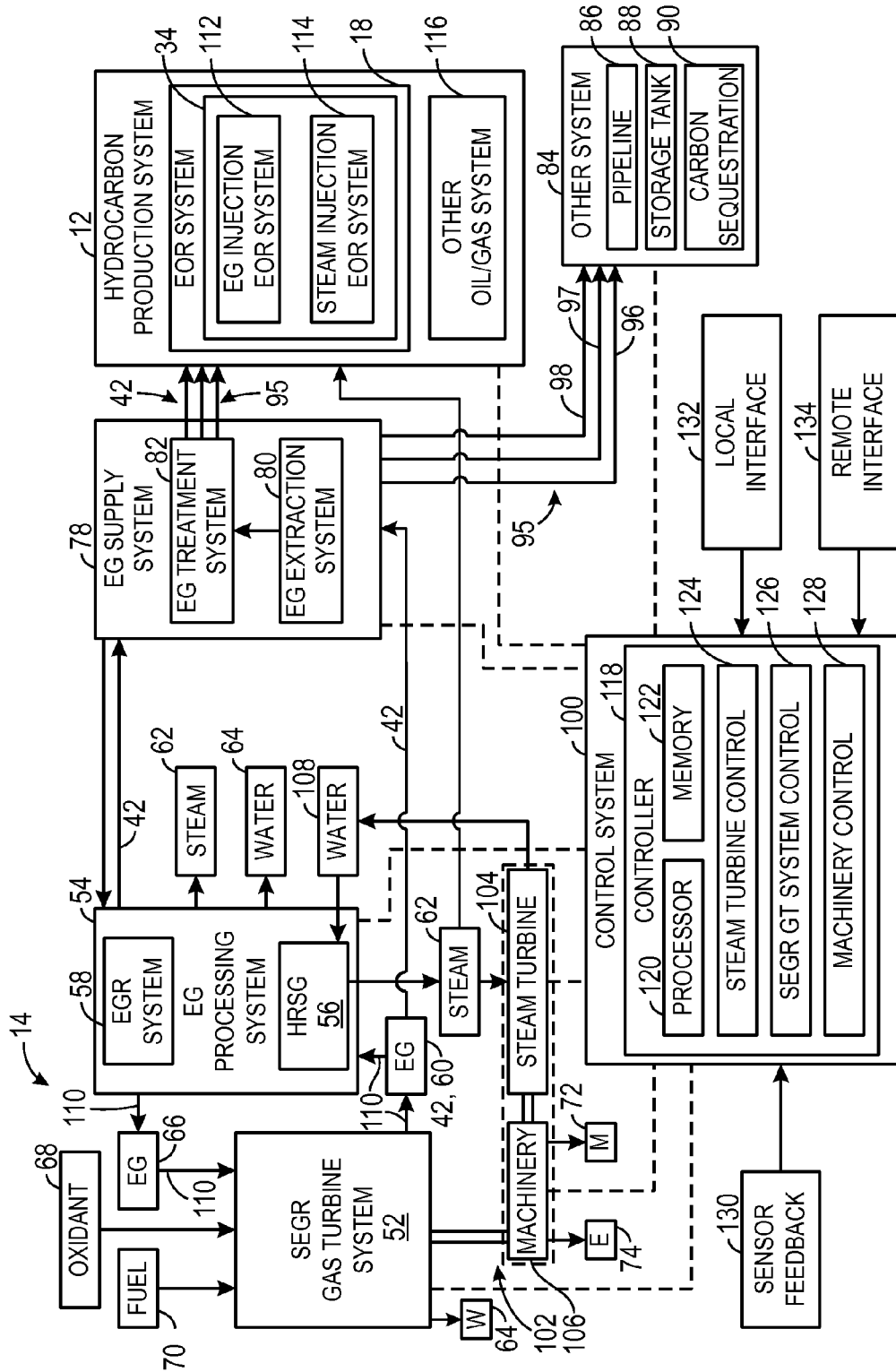


FIG. 2

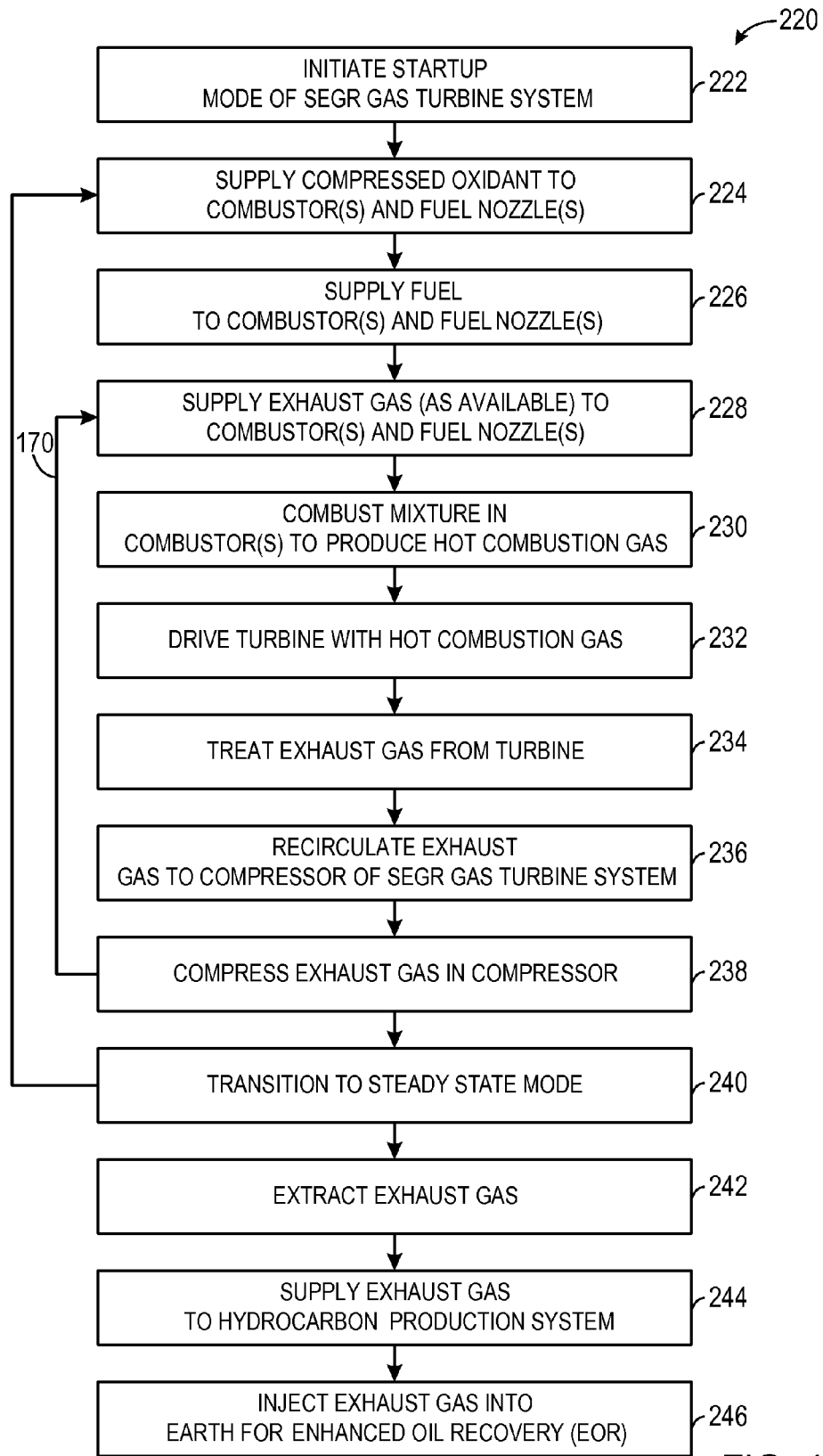


FIG. 4

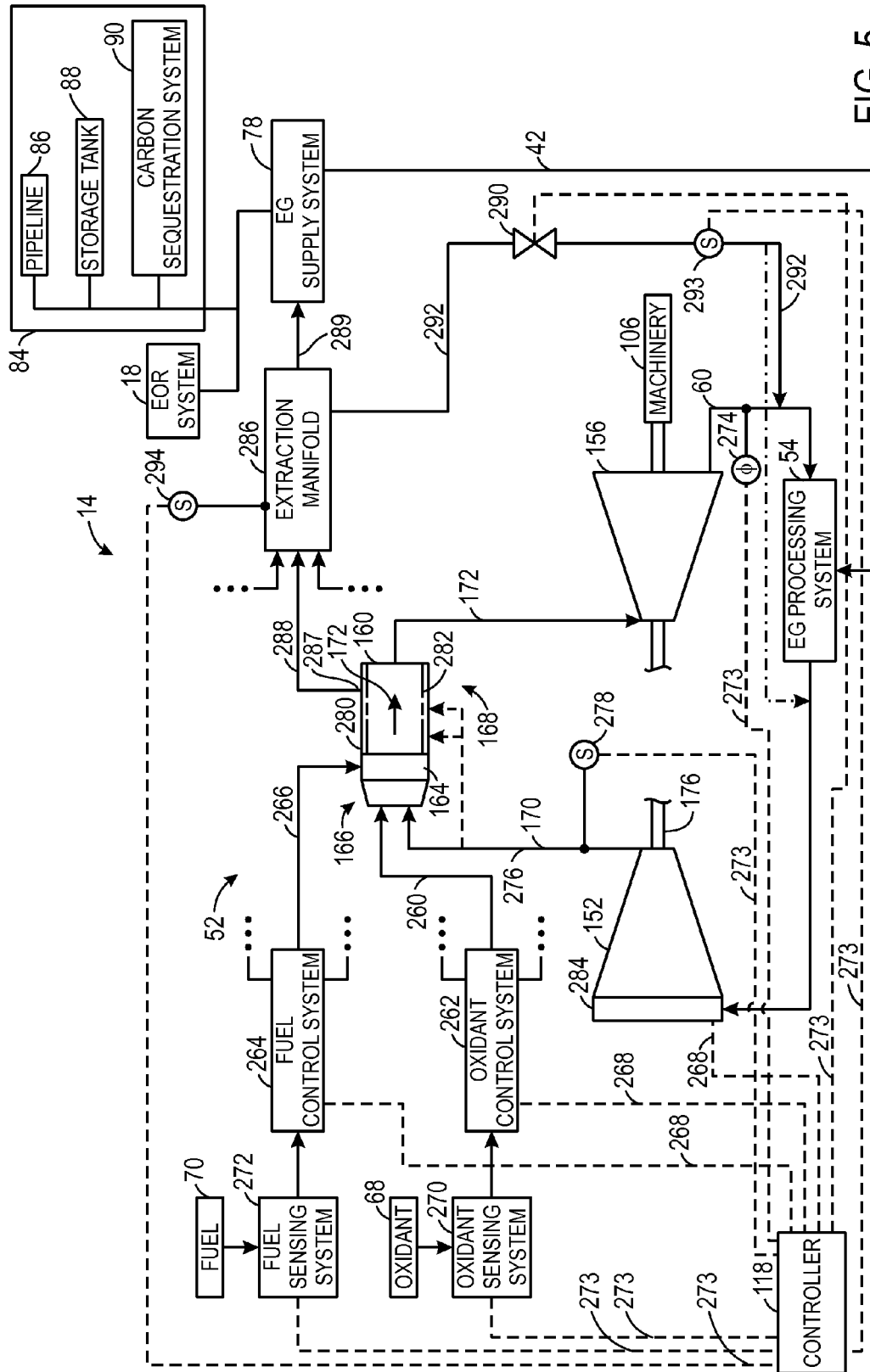


FIG. 5

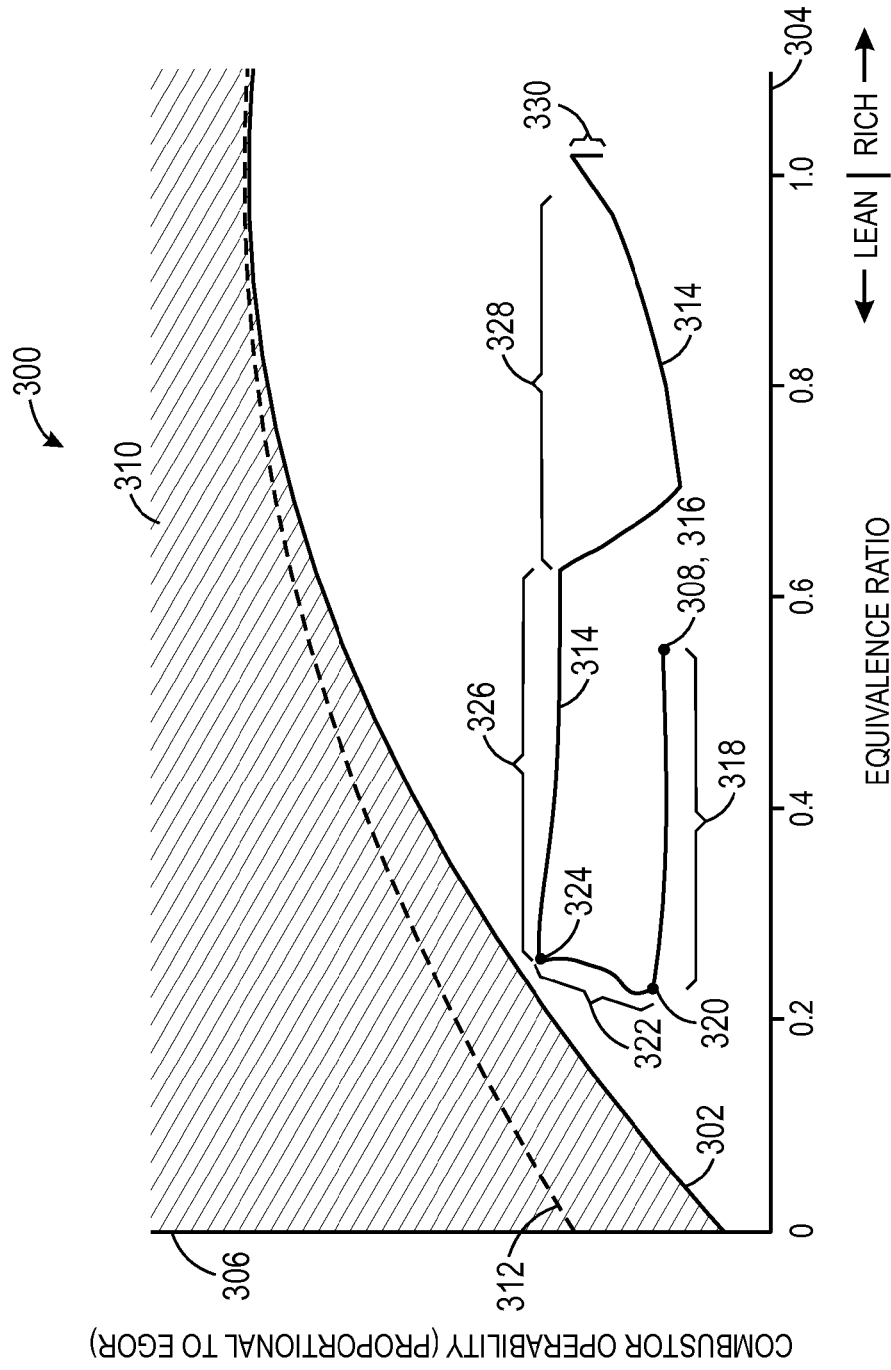


FIG. 6

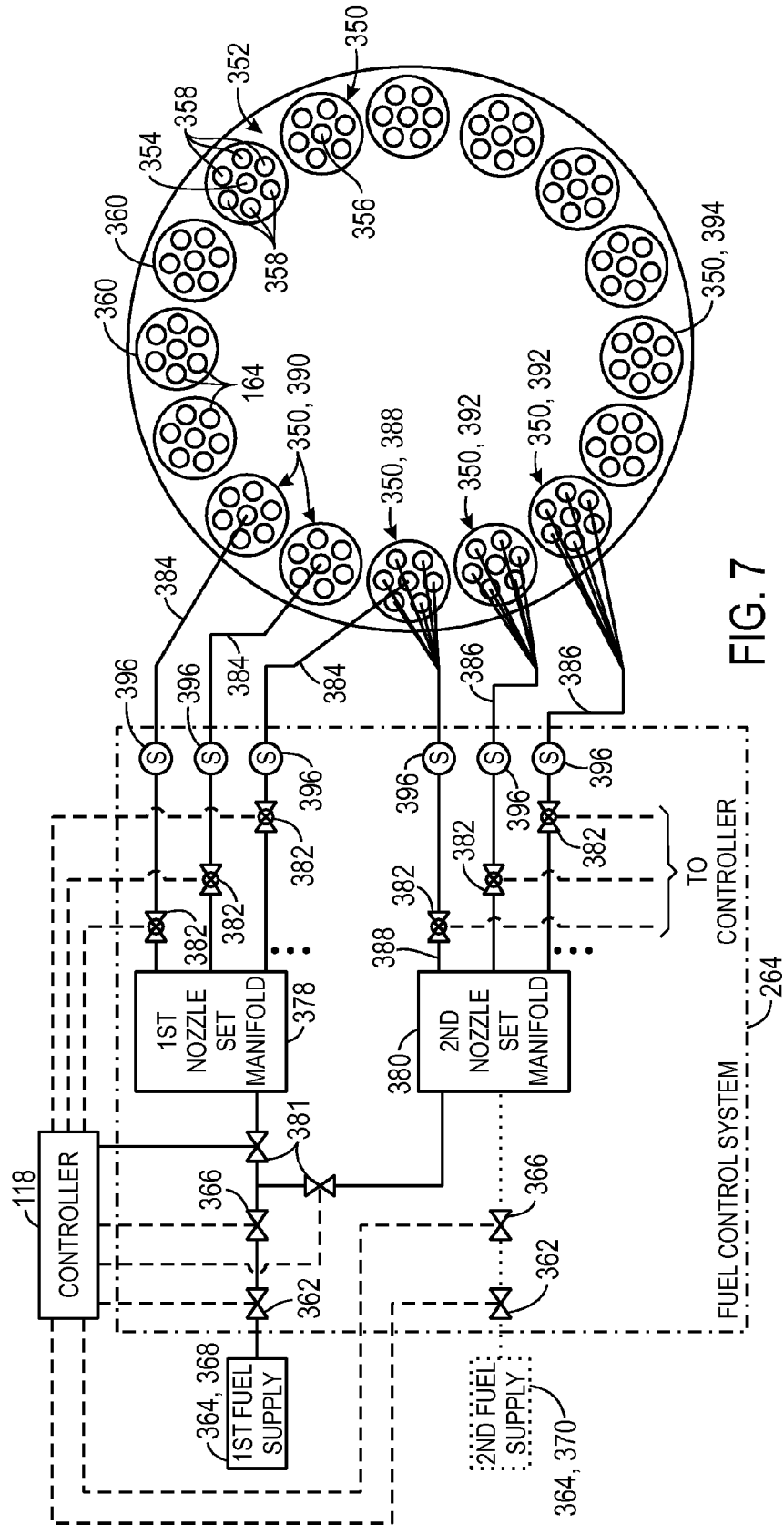
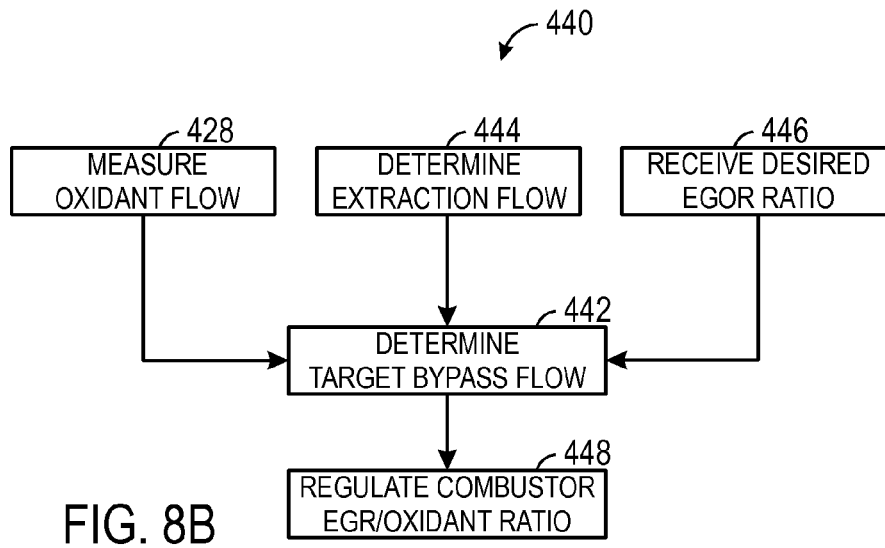
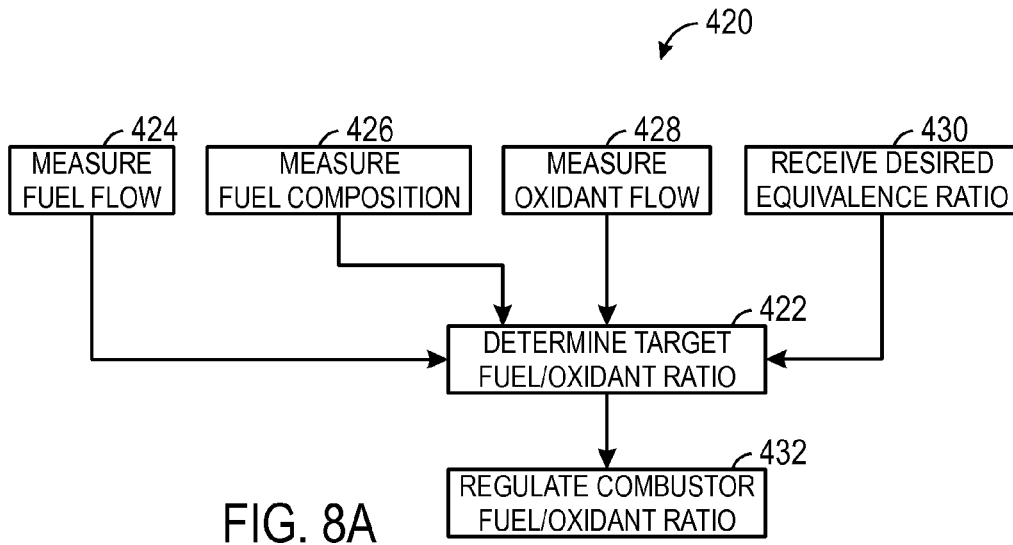


FIG. 7



**SYSTEM AND METHOD FOR
CONTROLLING THE COMBUSTION
PROCESS IN A GAS TURBINE OPERATING
WITH EXHAUST GAS RECIRCULATION**

CROSS-REFERENCE TO RELATED
APPLICATIONS

[0001] This application claims priority to and benefit of U.S. Provisional Patent Application No. 61/929,868, entitled "SYSTEM AND METHOD FOR CONTROLLING THE COMBUSTION PROCESS IN A GAS TURBINE OPERATING WITH EXHAUST GAS RECIRCULATION," filed Jan. 21, 2014, which is hereby incorporated by reference in its entirety for all purposes.

BACKGROUND

[0002] The subject matter disclosed herein relates to gas turbine engines with recirculation of exhaust gas.

[0003] Gas turbine engines are used in a wide variety of applications, such as power generation, aircraft, and various machinery. Gas turbine engines generally combust a fuel with an oxidant (e.g., air) in a combustor section to generate hot combustion products, which then drive one or more turbine stages of a turbine section. In turn, the turbine section drives one or more compressor stages of a compressor section, thereby compressing oxidant for intake into the combustor section along with the fuel. Again, the fuel and oxidant mix in the combustor section, and then combust to produce the hot combustion products. The combustion products may be recirculated back to the combustor section. Generally, the nature of the combustion products, such as the relative levels of particular gases in the combustion products (e.g., nitrogen oxide (NO_x), and carbon dioxide (CO₂)) can be affected by the ratio of fuel to oxidant used in the combustion process. Unfortunately, certain ratios of fuel to oxidant or recirculated combustion products to oxidant may decrease the stability of combustion within the combustor section.

BRIEF DESCRIPTION

[0004] Certain embodiments commensurate in scope with the originally claimed invention are summarized below. These embodiments are not intended to limit the scope of the claimed invention, but rather these embodiments are intended only to provide a brief summary of possible forms of the invention. Indeed, the invention may encompass a variety of forms that may be similar to or different from the embodiments set forth below.

[0005] In a first embodiment, a system includes a fuel control system configured to control a fuel flow to one or more combustors and an oxidant control system configured to control an oxidant flow to each combustor of the one or more combustors, wherein the oxidant flow is configured to at least partially react with the fuel flow within the one or more combustors to form an exhaust gas flow. The system also includes an exhaust gas system configured to direct a recirculation flow of the exhaust gas flow to each combustor of the one or more combustors; and a controller coupled to the fuel control system, the oxidant control system, and the exhaust gas system. The controller is configured to independently control a fuel-to-oxidant ratio (FOR) and an exhaust gas-to-oxidant ratio (EGOR). The FOR is the fuel flow divided by the oxidant flow, and the EGOR is the recirculation flow divided by the oxidant flow.

[0006] In a second embodiment, a system includes a combustor, a turbine, and a turbine bypass valve. The combustor includes an oxidant inlet configured to receive an oxidant flow, a plurality of fuel nozzles, wherein each fuel nozzle of the plurality of fuel nozzles is configured to receive a fuel flow, and a first fuel trim valve configured to control the fuel flow to at least one fuel nozzle of the plurality of fuel nozzles. The first fuel trim valve is controlled based at least in part on a fuel-to-oxidant ratio (FOR). The combustor also includes a recirculation inlet configured to receive a recirculation flow from a recirculation system, wherein at least the oxidant flow and the fuel flow are configured to combust within the combustor and form an exhaust gas flow, and one or more extraction ports configured to extract a first portion of the recirculation flow to an extraction manifold. The turbine is configured to receive the exhaust gas flow and a second portion of the recirculation flow from the combustor and the exhaust gas flow, to drive a load, and to direct the second portion of the exhaust gas flow to the recirculation system. The turbine bypass valve is configured to extract a bypass flow from the first portion of the recirculation flow to the extraction manifold, wherein the turbine bypass valve is configured to direct the bypass flow to the recirculation system based at least in part on an exhaust-gas-to-oxidant ratio (EGOR), wherein the recirculation flow comprises the second portion of the exhaust gas flow and the bypass flow, and the turbine bypass valve is controlled independently of the first fuel trim valve.

[0007] In a third embodiment, a method of operating an exhaust gas recirculation (EGR) gas turbine engine includes controlling a fuel-to-oxidant ratio (FOR) to a combustor based at least in part on a desired equivalence ratio and a load on the EGR gas turbine engine, combusting fuel and oxidant in the combustor to form exhaust gas, recirculating a recirculation portion of the exhaust gas to the combustor, and controlling an exhaust-gas-to-oxidant ratio (EGOR) based at least in part on an operability limit corresponding to the FOR.

BRIEF DESCRIPTION OF THE DRAWINGS

[0008] These and other features, aspects, and advantages of the present invention will become better understood when the following detailed description is read with reference to the accompanying drawings in which like characters represent like parts throughout the drawings, wherein:

[0009] FIG. 1 is a diagram of an embodiment of a system having a turbine-based service system coupled to a hydrocarbon production system;

[0010] FIG. 2 is a diagram of an embodiment of the system of FIG. 1, further illustrating a control system and a combined cycle system;

[0011] FIG. 3 is a diagram of an embodiment of the system of FIGS. 1 and 2, further illustrating details of a gas turbine engine, exhaust gas supply system, and exhaust gas processing system;

[0012] FIG. 4 is a flow chart of an embodiment of a process for operating the system of FIGS. 1-3;

[0013] FIG. 5 is a diagram of an embodiment of the system of FIGS. 1-3, further illustrating a control system, sensing systems for fuel and oxidant flows, and an exhaust gas extraction system with a turbine bypass valve;

[0014] FIG. 6 is chart of the equivalence ratio and the combustor operability during operation of an embodiment of the system of FIG. 5;

[0015] FIG. 7 is a diagram of an embodiment of a fuel control system of the system of FIG. 5; and

[0016] FIGS. 8A and 8B are flow charts of embodiments of a process for independently controlling the fuel-to-oxidant ratio and the exhaust-gas-to-oxidant ratio.

DETAILED DESCRIPTION

[0017] One or more specific embodiments of the present invention will be described below. In an effort to provide a concise description of these embodiments, all features of an actual implementation may not be described in the specification. It should be appreciated that in the development of any such actual implementation, as in any engineering or design project, numerous implementation-specific decisions must be made to achieve the developers' specific goals, such as compliance with system-related and business-related constraints, which may vary from one implementation to another. Moreover, it should be appreciated that such a development effort might be complex and time consuming, but would nevertheless be a routine undertaking of design, fabrication, and manufacture for those of ordinary skill having the benefit of this disclosure.

[0018] Accordingly, while example embodiments are capable of various modifications and alternative forms, embodiments thereof are illustrated by way of example in the figures and will herein be described in detail. It should be understood, however, that there is no intent to limit example embodiments to the particular forms disclosed, but to the contrary, example embodiments are to cover all modifications, equivalents, and alternatives falling within the scope of the present invention.

[0019] The terminology used herein is for describing particular embodiments only and is not intended to be limiting of example embodiments. As used herein, the singular forms "a", "an" and "the" are intended to include the plural forms as well, unless the context clearly indicates otherwise. The terms "comprises", "comprising", "includes" and/or "including", when used herein, specify the presence of stated features, integers, steps, operations, elements, and/or components, but do not preclude the presence or addition of one or more other features, integers, steps, operations, elements, components, and/or groups thereof.

[0020] Although the terms first, second, primary, secondary, etc. may be used herein to describe various elements, these elements should not be limited by these terms. These terms are only used to distinguish one element from another. For example, but not limiting to, a first element could be termed a second element, and, similarly, a second element could be termed a first element, without departing from the scope of example embodiments. As used herein, the term "and/or" includes any, and all, combinations of one or more of the associated listed items.

[0021] Certain terminology may be used herein for the convenience of the reader only and is not to be taken as a limitation on the scope of the invention. For example, words such as "upper", "lower", "left", "right", "front", "rear", "top", "bottom", "horizontal", "vertical", "upstream", "downstream", "fore", "aft", and the like; merely describe the configuration shown in the figures. Indeed, the element or elements of an embodiment of the present invention may be oriented in any direction and the terminology, therefore, should be understood as encompassing such variations unless specified otherwise.

[0022] As discussed in detail below, the disclosed embodiments relate generally to gas turbine systems with exhaust gas recirculation (EGR), and particularly stoichiometric operation of the gas turbine systems using EGR. For example, the gas turbine systems may be configured to recirculate the exhaust gas along an exhaust recirculation path, stoichiometrically combust fuel and oxidant along with at least some of the recirculated exhaust gas, and capture the exhaust gas for use in various target systems. The recirculation of the exhaust gas along with stoichiometric combustion may help to increase the concentration level of carbon dioxide (CO₂) in the exhaust gas, which can then be post treated to separate and purify the CO₂ and nitrogen (N₂) for use in various target systems. The gas turbine systems also may employ various exhaust gas processing (e.g., heat recovery, catalyst reactions, etc.) along the exhaust recirculation path, thereby increasing the concentration level of CO₂, reducing concentration levels of other emissions (e.g., carbon monoxide, nitrogen oxides, and unburnt hydrocarbons), and increasing energy recovery (e.g., with heat recovery units). Furthermore, the gas turbine engines may be configured to combust the fuel and oxidant with one or more diffusion flames (e.g., using diffusion fuel nozzles), premix flames (e.g., using premix fuel nozzles), or any combination thereof. In certain embodiments, the diffusion flames may help to maintain stability and operation within certain limits for stoichiometric combustion, which in turn helps to increase production of CO₂. For example, a gas turbine system operating with diffusion flames may enable a greater quantity of EGR, as compared to a gas turbine system operating with premix flames. In turn, the increased quantity of EGR helps to increase CO₂ production. Possible target systems include pipelines, storage tanks, carbon sequestration systems, and hydrocarbon production systems, such as enhanced oil recovery (EOR) systems.

[0023] Systems and methods described herein may independently control the fuel-to-oxidant ratio (FOR) of the mixture and the exhaust-gas-to-oxidant ratio (EGOR) of the mixture provided to the one or more combustors of the gas turbine system. The FOR and the EGOR may be independently controlled during a start-up sequence of the gas turbine system, during steady-state operation (e.g., driving a load, supplying exhaust gas flow to exhaust gas extraction system for enhanced oil recovery), or during a shutdown of the gas turbine system, or any combination thereof. In some embodiments, the FOR is controlled based at least in part on a load on the gas turbine system and/or an operating speed of the gas turbine system, and the EGOR is controlled based at least in part on an operability limit of a combustor of the gas turbine system. The operability limit of the combustor may correspond to a set of values for the FOR and the EGOR, wherein values of the FOR and the EGOR below the operability limit may describe combustion within the combustor that may be sustained with reduced likelihood of undesired flame blow-out, flashback, auto-ignition, or harmonic vibrations, or any combination thereof.

[0024] The FOR may be controlled by adjusting the oxidant flow and/or the fuel flow to the combustor. For example, the FOR may be controlled based on feed forward control and/or feedback control. The feed forward control, in certain embodiments, may be based on the composition of the fuel and the oxidant, current flow rates of the fuel and/or oxidant to the gas turbine system, and a desired stoichiometric fuel-to-air ratio. The feedback control, in some embodiments, may be based on a measured composition of the recirculated

exhaust gas. A control system (e.g., an industrial controller having a processor, memory, and instructions stored on memory and executable by the processor to perform control functions) may utilize the feed forward control, the feedback control, or any combination thereof, to control the flow of the fuel and/or oxidant to the gas turbine system. The FOR may be controlled to drive a load, to startup the gas turbine system, and/or to achieve a desired composition for the exhaust, such as to achieve higher levels of carbon dioxide (CO₂) in the exhaust gas. The control system may separately control the FOR to one or more combustors of a gas turbine system by adjusting the relative fuel and/or oxidant flows to each combustor. In some embodiments, the control system differentially controls a fuel flow to a first (e.g., center) set of fuel nozzles of a combustor relative to a fuel flow to a second (e.g., outer) set of fuel nozzles.

[0025] The EGOR may be controlled by adjusting the exhaust gas flow (e.g., recirculated exhaust gas) that mixes with the oxidant and fuel flows. The exhaust gas flow may be supplied to the combustor to cool the combustion gases, to dilute the oxidant, or to reduce the amounts of residual oxidant and/or fuel in the exhaust gas flow, or any combination thereof. The control system controls the exhaust gas to adjust the EGOR in order to maintain sufficient oxidant flow for a stable reaction with the fuel flow. A portion of the exhaust gas flow supplied to the combustor may be extracted to an extraction manifold for transmission to an exhaust gas supply system (e.g., enhanced oil recovery, storage tank, pipeline) and/or recirculation to the exhaust gas compressor. The control system may control a turbine bypass valve to control a bypass portion of the exhaust gas that bypasses the turbine section and flows from the extraction manifold to the exhaust gas compressor for recirculation to the combustor. The control system may control the turbine bypass valve to adjust a quantity of exhaust gas extracted to the extraction manifold. In some embodiments, adjustments to the turbine bypass valve may not reduce the exhaust gas supplied from the extraction manifold to an exhaust gas supply system. Accordingly, the control system may control the EGOR by controlling the quantity of exhaust gas that mixes with the combustion gases. Additionally or in the alternative, the control system may control inlet guide vanes to the exhaust gas compressor, thereby adjusting the quantity of exhaust gas recirculated from the turbine section to the exhaust gas compressor. The control system may control the exhaust gas supplied to the combustor via control of the inlet guide vanes, and the control system may control the exhaust gas extracted from the combustor via control of the turbine bypass valve.

[0026] FIG. 1 is a diagram of an embodiment of a system 10 having a hydrocarbon production system 12 associated with a turbine-based service system 14. As discussed in further detail below, various embodiments of the turbine-based service system 14 are configured to provide various services, such as electrical power, mechanical power, and fluids (e.g., exhaust gas), to the hydrocarbon production system 12 to facilitate the production or retrieval of oil and/or gas. In the illustrated embodiment, the hydrocarbon production system 12 includes an oil/gas extraction system 16 and an enhanced oil recovery (EOR) system 18, which are coupled to a subterranean reservoir 20 (e.g., an oil, gas, or hydrocarbon reservoir). The oil/gas extraction system 16 includes a variety of surface equipment 22, such as a Christmas tree or production tree 24, coupled to an oil/gas well 26. Furthermore, the well 26 may include one or more tubulars 28 extending through a

drilled bore 30 in the earth 32 to the subterranean reservoir 20. The tree 24 includes one or more valves, chokes, isolation sleeves, blowout preventers, and various flow control devices, which regulate pressures and control flows to and from the subterranean reservoir 20. While the tree 24 is generally used to control the flow of the production fluid (e.g., oil or gas) out of the subterranean reservoir 20, the EOR system 18 may increase the production of oil or gas by injecting one or more fluids into the subterranean reservoir 20.

[0027] Accordingly, the EOR system 18 may include a fluid injection system 34, which has one or more tubulars 36 extending through a bore 38 in the earth 32 to the subterranean reservoir 20. For example, the EOR system 18 may route one or more fluids 40, such as gas, steam, water, chemicals, or any combination thereof, into the fluid injection system 34. For example, as discussed in further detail below, the EOR system 18 may be coupled to the turbine-based service system 14, such that the system 14 routes an exhaust gas 42 (e.g., substantially or entirely free of oxygen) to the EOR system 18 for use as the injection fluid 40. The fluid injection system 34 routes the fluid 40 (e.g., the exhaust gas 42) through the one or more tubulars 36 into the subterranean reservoir 20, as indicated by arrows 44. The injection fluid 40 enters the subterranean reservoir 20 through the tubular 36 at an offset distance 46 away from the tubular 28 of the oil/gas well 26. Accordingly, the injection fluid 40 displaces the oil/gas 48 disposed in the subterranean reservoir 20, and drives the oil/gas 48 up through the one or more tubulars 28 of the hydrocarbon production system 12, as indicated by arrows 50. As discussed in further detail below, the injection fluid 40 may include the exhaust gas 42 originating from the turbine-based service system 14, which is able to generate the exhaust gas 42 on-site as needed by the hydrocarbon production system 12. In other words, the turbine-based system 14 may simultaneously generate one or more services (e.g., electrical power, mechanical power, steam, water (e.g., desalinated water), and exhaust gas (e.g., substantially free of oxygen)) for use by the hydrocarbon production system 12, thereby reducing or eliminating the reliance on external sources of such services.

[0028] In the illustrated embodiment, the turbine-based service system 14 includes a stoichiometric exhaust gas recirculation (SEGR) gas turbine system 52 and an exhaust gas (EG) processing system 54. The gas turbine system 52 may be configured to operate in a stoichiometric combustion mode of operation (e.g., a stoichiometric control mode) and a non-stoichiometric combustion mode of operation (e.g., a non-stoichiometric control mode), such as a fuel-lean control mode or a fuel-rich control mode. In the stoichiometric control mode, the combustion generally occurs in a substantially stoichiometric ratio of a fuel and oxidant, thereby resulting in substantially stoichiometric combustion. In particular, stoichiometric combustion generally involves consuming substantially all of the fuel and oxidant in the combustion reaction, such that the products of combustion are substantially or entirely free of unburnt fuel and oxidant. One measure of stoichiometric combustion is the equivalence ratio, or phi (Φ), which is the ratio of the actual fuel/oxidant ratio relative to the stoichiometric fuel/oxidant ratio. An equivalence ratio of greater than 1.0 results in a fuel-rich combustion of the fuel and oxidant, whereas an equivalence ratio of less than 1.0 results in a fuel-lean combustion of the fuel and oxidant. In contrast, an equivalence ratio of 1.0 results in combustion that is neither fuel-rich nor fuel-lean, thereby substantially con-

suming all of the fuel and oxidant in the combustion reaction. In context of the disclosed embodiments, the term stoichiometric or substantially stoichiometric may refer to an equivalence ratio of approximately 0.95 to approximately 1.05. However, the disclosed embodiments may also include an equivalence ratio of 1.0 plus or minus 0.01, 0.02, 0.03, 0.04, 0.05, or more. Again, the stoichiometric combustion of fuel and oxidant in the turbine-based service system 14 may result in products of combustion or exhaust gas (e.g., 42) with substantially no unburnt fuel or oxidant remaining. For example, the exhaust gas 42 may have less than 1, 2, 3, 4, or 5 percent by volume of oxidant (e.g., oxygen), unburnt fuel or hydrocarbons (e.g., HCs), nitrogen oxides (e.g., NO_x), carbon monoxide (CO), sulfur oxides (e.g., SO_x), hydrogen, and other products of incomplete combustion. By further example, the exhaust gas 42 may have less than approximately 10, 20, 30, 40, 50, 60, 70, 80, 90, 100, 200, 300, 400, 500, 1000, 2000, 3000, 4000, or 5000 parts per million by volume (ppmv) of oxidant (e.g., oxygen), unburnt fuel or hydrocarbons (e.g., HCs), nitrogen oxides (e.g., NO_x), carbon monoxide (CO), sulfur oxides (e.g., SO_x), hydrogen, and other products of incomplete combustion. However, the disclosed embodiments also may produce other ranges of residual fuel, oxidant, and other emissions levels in the exhaust gas 42. As used herein, the terms emissions, emissions levels, and emissions targets may refer to concentration levels of certain products of combustion (e.g., NO_x, CO, SO_x, O₂, N₂, H₂, HCs, etc.), which may be present in recirculated gas streams, vented gas streams (e.g., exhausted into the atmosphere), and gas streams used in various target systems (e.g., the hydrocarbon production system 12).

[0029] Although the SEGR gas turbine system 52 and the EG processing system 54 may include a variety of components in different embodiments, the illustrated EG processing system 54 includes a heat recovery steam generator (HRSG) 56 and an exhaust gas recirculation (EGR) system 58, which receive and process an exhaust gas 60 originating from the SEGR gas turbine system 52. The HRSG 56 may include one or more heat exchangers, condensers, and various heat recovery equipment, which collectively function to transfer heat from the exhaust gas 60 to a stream of water, thereby generating steam 62. The steam 62 may be used in one or more steam turbines, the EOR system 18, or any other portion of the hydrocarbon production system 12. For example, the HRSG 56 may generate low pressure, medium pressure, and/or high pressure steam 62, which may be selectively applied to low, medium, and high pressure steam turbine stages, or different applications of the EOR system 18. In addition to the steam 62, a treated water 64, such as a desalinated water, may be generated by the HRSG 56, the EGR system 58, and/or another portion of the EG processing system 54 or the SEGR gas turbine system 52. The treated water 64 (e.g., desalinated water) may be particularly useful in areas with water shortages, such as inland or desert regions. The treated water 64 may be generated, at least in part, due to the large volume of air driving combustion of fuel within the SEGR gas turbine system 52. While the on-site generation of steam 62 and water 64 may be beneficial in many applications (including the hydrocarbon production system 12), the on-site generation of exhaust gas 42, 60 may be particularly beneficial for the EOR system 18, due to its low oxygen content, high pressure, and heat derived from the SEGR gas turbine system 52. Accordingly, the HRSG 56, the EGR system 58, and/or another portion of the EG processing system 54 may output or recir-

culate an exhaust gas 66 into the SEGR gas turbine system 52, while also routing the exhaust gas 42 to the EOR system 18 for use with the hydrocarbon production system 12. Likewise, the exhaust gas 42 may be extracted directly from the SEGR gas turbine system 52 (i.e., without passing through the EG processing system 54) for use in the EOR system 18 of the hydrocarbon production system 12.

[0030] The exhaust gas recirculation is handled by the EGR system 58 of the EG processing system 54. For example, the EGR system 58 includes one or more conduits, valves, blowers, exhaust gas treatment systems (e.g., filters, particulate removal units, gas separation units, gas purification units, heat exchangers, heat recovery units, moisture removal units, catalyst units, chemical injection units, or any combination thereof), and controls to recirculate the exhaust gas along an exhaust gas circulation path from an output (e.g., discharged exhaust gas 60) to an input (e.g., intake exhaust gas 66) of the SEGR gas turbine system 52. In the illustrated embodiment, the SEGR gas turbine system 52 intakes the exhaust gas 66 into a compressor section having one or more compressors, thereby compressing the exhaust gas 66 for use in a combustor section along with an intake of an oxidant 68 and one or more fuels 70. The oxidant 68 may include ambient air, pure oxygen, oxygen-enriched air, oxygen-reduced air, oxygen-nitrogen mixtures, or any suitable oxidant that facilitates combustion of the fuel 70. The fuel 70 may include one or more gas fuels, liquid fuels, or any combination thereof. For example, the fuel 70 may include natural gas, liquefied natural gas (LNG), syngas, methane, ethane, propane, butane, naphtha, kerosene, diesel fuel, ethanol, methanol, biofuel, or any combination thereof.

[0031] The SEGR gas turbine system 52 mixes and combusts the exhaust gas 66, the oxidant 68, and the fuel 70 in the combustor section, thereby generating hot combustion gases or exhaust gas 60 to drive one or more turbine stages in a turbine section. In certain embodiments, each combustor in the combustor section includes one or more premix fuel nozzles, one or more diffusion fuel nozzles, or any combination thereof. For example, each premix fuel nozzle may be configured to mix the oxidant 68 and the fuel 70 internally within the fuel nozzle and/or partially upstream of the fuel nozzle, thereby injecting an oxidant-fuel mixture from the fuel nozzle into the combustion zone for a premixed combustion (e.g., a premixed flame). By further example, each diffusion fuel nozzle may be configured to isolate the flows of oxidant 68 and fuel 70 within the fuel nozzle, thereby separately injecting the oxidant 68 and the fuel 70 from the fuel nozzle into the combustion zone for diffusion combustion (e.g., a diffusion flame). In particular, the diffusion combustion provided by the diffusion fuel nozzles delays mixing of the oxidant 68 and the fuel 70 until the point of initial combustion, i.e., the flame region. In embodiments employing the diffusion fuel nozzles, the diffusion flame may provide increased flame stability, because the diffusion flame generally forms at the point of stoichiometry between the separate streams of oxidant 68 and fuel 70 (i.e., as the oxidant 68 and fuel 70 are mixing). In certain embodiments, one or more diluents (e.g., the exhaust gas 60, steam, nitrogen, or another inert gas) may be pre-mixed with the oxidant 68, the fuel 70, or both, in either the diffusion fuel nozzle or the premix fuel nozzle. In addition, one or more diluents (e.g., the exhaust gas 60, steam, nitrogen, or another inert gas) may be injected into the combustor at or downstream from the point of combustion within each combustor. The use of these diluents may help

temper the flame (e.g., premix flame or diffusion flame), thereby helping to reduce NO_x emissions, such as nitrogen monoxide (NO) and nitrogen dioxide (NO_2). Regardless of the type of flame, the combustion produces hot combustion gases or exhaust gas 60 to drive one or more turbine stages. As each turbine stage is driven by the exhaust gas 60, the SEGR gas turbine system 52 generates a mechanical power 72 and/or an electrical power 74 (e.g., via an electrical generator). The system 52 also outputs the exhaust gas 60, and may further output water 64. Again, the water 64 may be a treated water, such as a desalinated water, which may be useful in a variety of applications on-site or off-site.

[0032] Exhaust extraction is also provided by the SEGR gas turbine system 52 using one or more extraction points 76. For example, the illustrated embodiment includes an exhaust gas (EG) supply system 78 having an exhaust gas (EG) extraction system 80 and an exhaust gas (EG) treatment system 82, which receive exhaust gas 42 from the extraction points 76, treat the exhaust gas 42, and then supply or distribute the exhaust gas 42 to various target systems. The target systems may include the EOR system 18 and/or other systems, such as a pipeline 86, a storage tank 88, or a carbon sequestration system 90. The EG extraction system 80 may include one or more conduits, valves, controls, and flow separations, which facilitate isolation of the exhaust gas 42 from the oxidant 68, the fuel 70, and other contaminants, while also controlling the temperature, pressure, and flow rate of the extracted exhaust gas 42. The EG treatment system 82 may include one or more heat exchangers (e.g., heat recovery units such as heat recovery steam generators, condensers, coolers, or heaters), catalyst systems (e.g., oxidation catalyst systems), particulate and/or water removal systems (e.g., gas dehydration units, inertial separators, coalescing filters, water impermeable filters, and other filters), chemical injection systems, solvent based treatment systems (e.g., absorbers, flash tanks, etc.), carbon capture systems, gas separation systems, gas purification systems, and/or a solvent based treatment system, exhaust gas compressors, any combination thereof. These subsystems of the EG treatment system 82 enable control of the temperature, pressure, flow rate, moisture content (e.g., amount of water removal), particulate content (e.g., amount of particulate removal), and gas composition (e.g., percentage of CO_2 , N_2 , etc.).

[0033] The extracted exhaust gas 42 is treated by one or more subsystems of the EG treatment system 82, depending on the target system. For example, the EG treatment system 82 may direct all or part of the exhaust gas 42 through a carbon capture system, a gas separation system, a gas purification system, and/or a solvent based treatment system, which is controlled to separate and purify a carbonaceous gas (e.g., carbon dioxide) 92 and/or nitrogen (N_2) 94 for use in the various target systems. For example, embodiments of the EG treatment system 82 may perform gas separation and purification to produce a plurality of different streams 95 of exhaust gas 42, such as a first stream 96, a second stream 97, and a third stream 98. The first stream 96 may have a first composition that is rich in carbon dioxide and/or lean in nitrogen (e.g., a CO_2 rich, N_2 lean stream). The second stream 97 may have a second composition that has intermediate concentration levels of carbon dioxide and/or nitrogen (e.g., intermediate concentration CO_2 , N_2 stream). The third stream 98 may have a third composition that is lean in carbon dioxide and/or rich in nitrogen (e.g., a CO_2 lean, N_2 rich stream). Each stream 95 (e.g., 96, 97, and 98) may include a gas dehydration unit,

a filter, a gas compressor, or any combination thereof, to facilitate delivery of the stream 95 to a target system. In certain embodiments, the CO_2 rich, N_2 lean stream 96 may have a CO_2 purity or concentration level of greater than approximately 70, 75, 80, 85, 90, 95, 96, 97, 98, or 99 percent by volume, and a N_2 purity or concentration level of less than approximately 1, 2, 3, 4, 5, 10, 15, 20, 25, or 30 percent by volume. In contrast, the CO_2 lean, N_2 rich stream 98 may have a CO_2 purity or concentration level of less than approximately 1, 2, 3, 4, 5, 10, 15, 20, 25, or 30 percent by volume, and an N_2 purity or concentration level of greater than approximately 70, 75, 80, 85, 90, 95, 96, 97, 98, or 99 percent by volume. The intermediate concentration CO_2 , N_2 stream 97 may have a CO_2 purity or concentration level and/or a N_2 purity or concentration level of between approximately 30 to 70, 35 to 65, 40 to 60, or 45 to 55 percent by volume. Although the foregoing ranges are merely non-limiting examples, the CO_2 rich, N_2 lean stream 96 and the CO_2 lean, N_2 rich stream 98 may be particularly well suited for use with the EOR system 18 and the other systems 84. However, any of these rich, lean, or intermediate concentration CO_2 streams 95 may be used, alone or in various combinations, with the EOR system 18 and the other systems 84. For example, the EOR system 18 and the other systems 84 (e.g., the pipeline 86, storage tank 88, and the carbon sequestration system 90) each may receive one or more CO_2 rich, N_2 lean streams 96, one or more CO_2 lean, N_2 rich streams 98, one or more intermediate concentration CO_2 , N_2 streams 97, and one or more untreated exhaust gas 42 streams (i.e., bypassing the EG treatment system 82).

[0034] The EG extraction system 80 extracts the exhaust gas 42 at one or more extraction points 76 along the compressor section, the combustor section, and/or the turbine section, such that the exhaust gas 42 may be used in the EOR system 18 and other systems 84 at suitable temperatures and pressures. The EG extraction system 80 and/or the EG treatment system 82 also may circulate fluid flows (e.g., exhaust gas 42) to and from the EG processing system 54. For example, a portion of the exhaust gas 42 passing through the EG processing system 54 may be extracted by the EG extraction system 80 for use in the EOR system 18 and the other systems 84. In certain embodiments, the EG supply system 78 and the EG processing system 54 may be independent or integral with one another, and thus may use independent or common subsystems. For example, the EG treatment system 82 may be used by both the EG supply system 78 and the EG processing system 54. Exhaust gas 42 extracted from the EG processing system 54 may undergo multiple stages of gas treatment, such as one or more stages of gas treatment in the EG processing system 54 followed by one or more additional stages of gas treatment in the EG treatment system 82.

[0035] At each extraction point 76, the extracted exhaust gas 42 may be substantially free of oxidant 68 and fuel 70 (e.g., unburnt fuel or hydrocarbons) due to substantially stoichiometric combustion and/or gas treatment in the EG processing system 54. Furthermore, depending on the target system, the extracted exhaust gas 42 may undergo further treatment in the EG treatment system 82 of the EG supply system 78, thereby further reducing any residual oxidant 68, fuel 70, or other undesirable products of combustion. For example, either before or after treatment in the EG treatment system 82, the extracted exhaust gas 42 may have less than 1, 2, 3, 4, or 5 percent by volume of oxidant (e.g., oxygen), unburnt fuel or hydrocarbons (e.g., HCs), nitrogen oxides

(e.g., NO_x), carbon monoxide (CO), sulfur oxides (e.g., SO_x), hydrogen, and other products of incomplete combustion. By further example, either before or after treatment in the EG treatment system 82, the extracted exhaust gas 42 may have less than approximately 10, 20, 30, 40, 50, 60, 70, 80, 90, 100, 200, 300, 400, 500, 1000, 2000, 3000, 4000, or 5000 parts per million by volume (ppmv) of oxidant (e.g., oxygen), unburnt fuel or hydrocarbons (e.g., HCs), nitrogen oxides (e.g., NO_x), carbon monoxide (CO), sulfur oxides (e.g., SO_x), hydrogen, and other products of incomplete combustion. Thus, the exhaust gas 42 is particularly well suited for use with the EOR system 18.

[0036] The EGR operation of the turbine system 52 specifically enables the exhaust extraction at a multitude of locations 76. For example, the compressor section of the system 52 may be used to compress the exhaust gas 66 without any oxidant 68 (i.e., only compression of the exhaust gas 66), such that a substantially oxygen-free exhaust gas 42 may be extracted from the compressor section and/or the combustor section prior to entry of the oxidant 68 and the fuel 70. The extraction points 76 may be located at interstage ports between adjacent compressor stages, at ports along the compressor discharge casing, at ports along each combustor in the combustor section, or any combination thereof. In certain embodiments, the exhaust gas 66 may not mix with the oxidant 68 and fuel 70 until it reaches the head end portion and/or fuel nozzles of each combustor in the combustor section. Furthermore, one or more flow separators (e.g., walls, dividers, baffles, or the like) may be used to isolate the oxidant 68 and the fuel 70 from the extraction points 76. With these flow separators, the extraction points 76 may be disposed directly along a wall of each combustor in the combustor section.

[0037] Once the exhaust gas 66, oxidant 68, and fuel 70 flow through the head end portion (e.g., through fuel nozzles) into the combustion portion (e.g., combustion chamber) of each combustor, the SEGR gas turbine system 52 is controlled to provide a substantially stoichiometric combustion of the exhaust gas 66, oxidant 68, and fuel 70. For example, the system 52 may maintain an equivalence ratio of approximately 0.95 to approximately 1.05. As a result, the products of combustion of the mixture of exhaust gas 66, oxidant 68, and fuel 70 in each combustor is substantially free of oxygen and unburnt fuel. Thus, the products of combustion (or exhaust gas) may be extracted from the turbine section of the SEGR gas turbine system 52 for use as the exhaust gas 42 routed to the EOR system 18. Along the turbine section, the extraction points 76 may be located at any turbine stage, such as interstage ports between adjacent turbine stages. Thus, using any of the foregoing extraction points 76, the turbine-based service system 14 may generate, extract, and deliver the exhaust gas 42 to the hydrocarbon production system 12 (e.g., the EOR system 18) for use in the production of oil/gas 48 from the subterranean reservoir 20.

[0038] FIG. 2 is a diagram of an embodiment of the system 10 of FIG. 1, illustrating a control system 100 coupled to the turbine-based service system 14 and the hydrocarbon production system 12. In the illustrated embodiment, the turbine-based service system 14 includes a combined cycle system 102, which includes the SEGR gas turbine system 52 as a topping cycle, a steam turbine 104 as a bottoming cycle, and the HRSG 56 to recover heat from the exhaust gas 60 to generate the steam 62 for driving the steam turbine 104. Again, the SEGR gas turbine system 52 receives, mixes, and stoichiometrically combusts the exhaust gas 66, the oxidant

68, and the fuel 70 (e.g., premix and/or diffusion flames), thereby producing the exhaust gas 60, the mechanical power 72, the electrical power 74, and/or the water 64. For example, the SEGR gas turbine system 52 may drive one or more loads or machinery 106, such as an electrical generator, an oxidant compressor (e.g., a main air compressor), a gear box, a pump, equipment of the hydrocarbon production system 12, or any combination thereof. In some embodiments, the machinery 106 may include other drives, such as electrical motors or steam turbines (e.g., the steam turbine 104), in tandem with the SEGR gas turbine system 52. Accordingly, an output of the machinery 106 driven by the SEGR gas turbines system 52 (and any additional drives) may include the mechanical power 72 and the electrical power 74. The mechanical power 72 and/or the electrical power 74 may be used on-site for powering the hydrocarbon production system 12, the electrical power 74 may be distributed to the power grid, or any combination thereof. The output of the machinery 106 also may include a compressed fluid, such as a compressed oxidant 68 (e.g., air or oxygen), for intake into the combustion section of the SEGR gas turbine system 52. Each of these outputs (e.g., the exhaust gas 60, the mechanical power 72, the electrical power 74, and/or the water 64) may be considered a service of the turbine-based service system 14.

[0039] The SEGR gas turbine system 52 produces the exhaust gas 42, 60, which may be substantially free of oxygen, and routes this exhaust gas 42, 60 to the EG processing system 54 and/or the EG supply system 78. The EG supply system 78 may treat and deliver the exhaust gas 42 (e.g., streams 95) to the hydrocarbon production system 12 and/or the other systems 84. As discussed above, the EG processing system 54 may include the HRSG 56 and the EGR system 58. The HRSG 56 may include one or more heat exchangers, condensers, and various heat recovery equipment, which may be used to recover or transfer heat from the exhaust gas 60 to water 108 to generate the steam 62 for driving the steam turbine 104. Similar to the SEGR gas turbine system 52, the steam turbine 104 may drive one or more loads or machinery 106, thereby generating the mechanical power 72 and the electrical power 74. In the illustrated embodiment, the SEGR gas turbine system 52 and the steam turbine 104 are arranged in tandem to drive the same machinery 106. However, in other embodiments, the SEGR gas turbine system 52 and the steam turbine 104 may separately drive different machinery 106 to independently generate mechanical power 72 and/or electrical power 74. As the steam turbine 104 is driven by the steam 62 from the HRSG 56, the steam 62 gradually decreases in temperature and pressure. Accordingly, the steam turbine 104 recirculates the used steam 62 and/or water 108 back into the HRSG 56 for additional steam generation via heat recovery from the exhaust gas 60. In addition to steam generation, the HRSG 56, the EGR system 58, and/or another portion of the EG processing system 54 may produce the water 64, the exhaust gas 42 for use with the hydrocarbon production system 12, and the exhaust gas 66 for use as an input into the SEGR gas turbine system 52. For example, the water 64 may be a treated water 64, such as a desalinated water for use in other applications. The desalinated water may be particularly useful in regions of low water availability. Regarding the exhaust gas 60, embodiments of the EG processing system 54 may be configured to recirculate the exhaust gas 60 through the EGR system 58 with or without passing the exhaust gas 60 through the HRSG 56.

[0040] In the illustrated embodiment, the SEGR gas turbine system 52 has an exhaust recirculation path 110, which extends from an exhaust outlet to an exhaust inlet of the system 52. Along the path 110, the exhaust gas 60 passes through the EG processing system 54, which includes the HRSG 56 and the EGR system 58 in the illustrated embodiment. The EGR system 58 may include one or more conduits, valves, blowers, gas treatment systems (e.g., filters, particulate removal units, gas separation units, gas purification units, heat exchangers, heat recovery units such as heat recovery steam generators, moisture removal units, catalyst units, chemical injection units, or any combination thereof) in series and/or parallel arrangements along the path 110. In other words, the EGR system 58 may include any flow control components, pressure control components, temperature control components, moisture control components, and gas composition control components along the exhaust recirculation path 110 between the exhaust outlet and the exhaust inlet of the system 52. Accordingly, in embodiments with the HRSG 56 along the path 110, the HRSG 56 may be considered a component of the EGR system 58. However, in certain embodiments, the HRSG 56 may be disposed along an exhaust path independent from the exhaust recirculation path 110. Regardless of whether the HRSG 56 is along a separate path or a common path with the EGR system 58, the HRSG 56 and the EGR system 58 intake the exhaust gas 60 and output either the recirculated exhaust gas 66, the exhaust gas 42 for use with the EG supply system 78 (e.g., for the hydrocarbon production system 12 and/or other systems 84), or another output of exhaust gas. Again, the SEGR gas turbine system 52 intakes, mixes, and stoichiometrically combusts the exhaust gas 66, the oxidant 68, and the fuel 70 (e.g., premixed and/or diffusion flames) to produce a substantially oxygen-free and fuel-free exhaust gas 60 for distribution to the EG processing system 54, the hydrocarbon production system 12, or other systems 84.

[0041] As noted above with reference to FIG. 1, the hydrocarbon production system 12 may include a variety of equipment to facilitate the recovery or production of oil/gas 48 from a subterranean reservoir 20 through an oil/gas well 26. For example, the hydrocarbon production system 12 may include the EOR system 18 having the fluid injection system 34. In the illustrated embodiment, the fluid injection system 34 includes an exhaust gas injection EOR system 112 and a steam injection EOR system 114. Although the fluid injection system 34 may receive fluids from a variety of sources, the illustrated embodiment may receive the exhaust gas 42 and the steam 62 from the turbine-based service system 14. The exhaust gas 42 and/or the steam 62 produced by the turbine-based service system 14 also may be routed to the hydrocarbon production system 12 for use in other oil/gas systems 116.

[0042] The quantity, quality, and flow of the exhaust gas 42 and/or the steam 62 may be controlled by the control system 100. The control system 100 may be dedicated entirely to the turbine-based service system 14, or the control system 100 may optionally also provide control (or at least some data to facilitate control) for the hydrocarbon production system 12 and/or other systems 84. In the illustrated embodiment, the control system 100 includes a controller 118 having a processor 120, a memory 122, a steam turbine control 124, a SEGR gas turbine system control 126, and a machinery control 128. The processor 120 may include a single processor or two or more redundant processors, such as triple redundant proces-

sors for control of the turbine-based service system 14. The memory 122 may include volatile and/or non-volatile memory. For example, the memory 122 may include one or more hard drives, flash memory, read-only memory, random access memory, or any combination thereof. The controls 124, 126, and 128 may include software and/or hardware controls. For example, the controls 124, 126, and 128 may include various instructions or code stored on the memory 122 and executable by the processor 120. The control 124 is configured to control operation of the steam turbine 104, the SEGR gas turbine system control 126 is configured to control the system 52, and the machinery control 128 is configured to control the machinery 106. Thus, the controller 118 (e.g., controls 124, 126, and 128) may be configured to coordinate various sub-systems of the turbine-based service system 14 to provide a suitable stream of the exhaust gas 42 to the hydrocarbon production system 12.

[0043] In certain embodiments of the control system 100, each element (e.g., system, subsystem, and component) illustrated in the drawings or described herein includes (e.g., directly within, upstream, or downstream of such element) one or more industrial control features, such as sensors and control devices, which are communicatively coupled with one another over an industrial control network along with the controller 118. For example, the control devices associated with each element may include a dedicated device controller (e.g., including a processor, memory, and control instructions), one or more actuators, valves, switches, and industrial control equipment, which enable control based on sensor feedback 130, control signals from the controller 118, control signals from a user, or any combination thereof. Thus, any of the control functionality described herein may be implemented with control instructions stored and/or executable by the controller 118, dedicated device controllers associated with each element, or a combination thereof.

[0044] In order to facilitate such control functionality, the control system 100 includes one or more sensors distributed throughout the system 10 to obtain the sensor feedback 130 for use in execution of the various controls, e.g., the controls 124, 126, and 128. For example, the sensor feedback 130 may be obtained from sensors distributed throughout the SEGR gas turbine system 52, the machinery 106, the EG processing system 54, the steam turbine 104, the hydrocarbon production system 12, or any other components throughout the turbine-based service system 14 or the hydrocarbon production system 12. For example, the sensor feedback 130 may include temperature feedback, pressure feedback, flow rate feedback, flame temperature feedback, combustion dynamics feedback, intake oxidant composition feedback, intake fuel composition feedback, exhaust composition feedback, the output level of mechanical power 72, the output level of electrical power 74, the output quantity of the exhaust gas 42, 60, the output quantity or quality of the water 64, or any combination thereof. For example, the sensor feedback 130 may include a composition of the exhaust gas 42, 60 to facilitate stoichiometric combustion in the SEGR gas turbine system 52. For example, the sensor feedback 130 may include feedback from one or more intake oxidant sensors along an oxidant supply path of the oxidant 68, one or more intake fuel sensors along a fuel supply path of the fuel 70, and one or more exhaust emissions sensors disposed along the exhaust recirculation path 110 and/or within the SEGR gas turbine system 52. The intake oxidant sensors, intake fuel sensors, and exhaust emissions sensors may include temperature sensors, pressure sen-

sors, flow rate sensors, and composition sensors. The emissions sensors may include sensors for nitrogen oxides (e.g., NO_x sensors), carbon oxides (e.g., CO sensors and CO₂ sensors), sulfur oxides (e.g., SO_x sensors), hydrogen (e.g., H₂ sensors), oxygen (e.g., O₂ sensors), unburnt hydrocarbons (e.g., HC sensors), or other products of incomplete combustion, or any combination thereof.

[0045] Using this feedback **130**, the control system **100** may adjust (e.g., increase, decrease, or maintain) the intake flow of exhaust gas **66**, oxidant **68**, and/or fuel **70** into the SEGR gas turbine system **52** (among other operational parameters) to maintain the equivalence ratio within a suitable range, e.g., between approximately 0.95 to approximately 1.05, between approximately 0.95 to approximately 1.0, between approximately 1.0 to approximately 1.05, or substantially at 1.0. For example, the control system **100** may analyze the feedback **130** to monitor the exhaust emissions (e.g., concentration levels of nitrogen oxides, carbon oxides such as CO and CO₂, sulfur oxides, hydrogen, oxygen, unburnt hydrocarbons, and other products of incomplete combustion) and/or determine the equivalence ratio, and then control one or more components to adjust the exhaust emissions (e.g., concentration levels in the exhaust gas **42**) and/or the equivalence ratio. The controlled components may include any of the components illustrated and described with reference to the drawings, including but not limited to, valves along the supply paths for the oxidant **68**, the fuel **70**, and the exhaust gas **66**; an oxidant compressor, a fuel pump, or any components in the EG processing system **54**; any components of the SEGR gas turbine system **52**, or any combination thereof. The controlled components may adjust (e.g., increase, decrease, or maintain) the flow rates, temperatures, pressures, or percentages (e.g., equivalence ratio) of the oxidant **68**, the fuel **70**, and the exhaust gas **66** that combust within the SEGR gas turbine system **52**. The controlled components also may include one or more gas treatment systems, such as catalyst units (e.g., oxidation catalyst units), supplies for the catalyst units (e.g., oxidation fuel, heat, electricity, etc.), gas purification and/or separation units (e.g., solvent based separators, absorbers, flash tanks, etc.), and filtration units. The gas treatment systems may help reduce various exhaust emissions along the exhaust recirculation path **110**, a vent path (e.g., exhausted into the atmosphere), or an extraction path to the EG supply system **78**.

[0046] In certain embodiments, the control system **100** may analyze the feedback **130** and control one or more components to maintain or reduce emissions levels (e.g., concentration levels in the exhaust gas **42**, **60**, **95**) to a target range, such as less than approximately 10, 20, 30, 40, 50, 100, 200, 300, 400, 500, 1000, 2000, 3000, 4000, 5000, or 10000 parts per million by volume (ppmv). These target ranges may be the same or different for each of the exhaust emissions, e.g., concentration levels of nitrogen oxides, carbon monoxide, sulfur oxides, hydrogen, oxygen, unburnt hydrocarbons, and other products of incomplete combustion. For example, depending on the equivalence ratio, the control system **100** may selectively control exhaust emissions (e.g., concentration levels) of oxidant (e.g., oxygen) within a target range of less than approximately 10, 20, 30, 40, 50, 60, 70, 80, 90, 100, 250, 500, 750, or 1000 ppmv; carbon monoxide (CO) within a target range of less than approximately 20, 50, 100, 200, 500, 1000, 2500, or 5000 ppmv; and nitrogen oxides (NO_x) within a target range of less than approximately 50, 100, 200, 300, 400, or 500 ppmv. In certain embodiments operating

with a substantially stoichiometric equivalence ratio, the control system **100** may selectively control exhaust emissions (e.g., concentration levels) of oxidant (e.g., oxygen) within a target range of less than approximately 10, 20, 30, 40, 50, 60, 70, 80, 90, or 100 ppmv; and carbon monoxide (CO) within a target range of less than approximately 500, 1000, 2000, 3000, 4000, or 5000 ppmv. In certain embodiments operating with a fuel-lean equivalence ratio (e.g., between approximately 0.95 to 1.0), the control system **100** may selectively control exhaust emissions (e.g., concentration levels) of oxidant (e.g., oxygen) within a target range of less than approximately 500, 600, 700, 800, 900, 1000, 1100, 1200, 1300, 1400, or 1500 ppmv; carbon monoxide (CO) within a target range of less than approximately 10, 20, 30, 40, 50, 60, 70, 80, 90, 100, 150, or 200 ppmv; and nitrogen oxides (e.g., NO_x) within a target range of less than approximately 50, 100, 150, 200, 250, 300, 350, or 400 ppmv. The foregoing target ranges are merely examples, and are not intended to limit the scope of the disclosed embodiments.

[0047] The control system **100** also may be coupled to a local interface **132** and a remote interface **134**. For example, the local interface **132** may include a computer workstation disposed on-site at the turbine-based service system **14** and/or the hydrocarbon production system **12**. In contrast, the remote interface **134** may include a computer workstation disposed off-site from the turbine-based service system **14** and the hydrocarbon production system **12**, such as through an internet connection. These interfaces **132** and **134** facilitate monitoring and control of the turbine-based service system **14**, such as through one or more graphical displays of sensor feedback **130**, operational parameters, and so forth.

[0048] Again, as noted above, the controller **118** includes a variety of controls **124**, **126**, and **128** to facilitate control of the turbine-based service system **14**. The steam turbine control **124** may receive the sensor feedback **130** and output control commands to facilitate operation of the steam turbine **104**. For example, the steam turbine control **124** may receive the sensor feedback **130** from the HRSG **56**, the machinery **106**, temperature and pressure sensors along a path of the steam **62**, temperature and pressure sensors along a path of the water **108**, and various sensors indicative of the mechanical power **72** and the electrical power **74**. Likewise, the SEGR gas turbine system control **126** may receive sensor feedback **130** from one or more sensors disposed along the SEGR gas turbine system **52**, the machinery **106**, the EG processing system **54**, or any combination thereof. For example, the sensor feedback **130** may be obtained from temperature sensors, pressure sensors, clearance sensors, vibration sensors, flame sensors, fuel composition sensors, exhaust gas composition sensors, or any combination thereof, disposed within or external to the SEGR gas turbine system **52**. Finally, the machinery control **128** may receive sensor feedback **130** from various sensors associated with the mechanical power **72** and the electrical power **74**, as well as sensors disposed within the machinery **106**. Each of these controls **124**, **126**, and **128** uses the sensor feedback **130** to improve operation of the turbine-based service system **14**.

[0049] In the illustrated embodiment, the SEGR gas turbine system control **126** may execute instructions to control the quantity and quality of the exhaust gas **42**, **60**, **95** in the EG processing system **54**, the EG supply system **78**, the hydrocarbon production system **12**, and/or the other systems **84**. For example, the SEGR gas turbine system control **126** may maintain a level of oxidant (e.g., oxygen) and/or unburnt fuel

in the exhaust gas **60** below a threshold suitable for use with the exhaust gas injection EOR system **112**. In certain embodiments, the threshold levels may be less than 1, 2, 3, 4, or 5 percent of oxidant (e.g., oxygen) and/or unburnt fuel by volume of the exhaust gas **42, 60**; or the threshold levels of oxidant (e.g., oxygen) and/or unburnt fuel (and other exhaust emissions) may be less than approximately 10, 20, 30, 40, 50, 60, 70, 80, 90, 100, 200, 300, 400, 500, 1000, 2000, 3000, 4000, or 5000 parts per million by volume (ppmv) in the exhaust gas **42, 60**. By further example, in order to achieve these low levels of oxidant (e.g., oxygen) and/or unburnt fuel, the SEGR gas turbine system control **126** may maintain an equivalence ratio for combustion in the SEGR gas turbine system **52** between approximately 0.95 and approximately 1.05. The SEGR gas turbine system control **126** also may control the EG extraction system **80** and the EG treatment system **82** to maintain the temperature, pressure, flow rate, and gas composition of the exhaust gas **42, 60, 95** within suitable ranges for the exhaust gas injection EOR system **112**, the pipeline **86**, the storage tank **88**, and the carbon sequestration system **90**. As discussed above, the EG treatment system **82** may be controlled to purify and/or separate the exhaust gas **42** into one or more gas streams **95**, such as the CO₂ rich, N₂ lean stream **96**, the intermediate concentration CO₂, N₂ stream **97**, and the CO₂ lean, N₂ rich stream **98**. In addition to controls for the exhaust gas **42, 60, and 95**, the controls **124, 126, and 128** may execute one or more instructions to maintain the mechanical power **72** within a suitable power range, or maintain the electrical power **74** within a suitable frequency and power range.

[0050] FIG. 3 is a diagram of embodiment of the system **10**, further illustrating details of the SEGR gas turbine system **52** for use with the hydrocarbon production system **12** and/or other systems **84**. In the illustrated embodiment, the SEGR gas turbine system **52** includes a gas turbine engine **150** coupled to the EG processing system **54**. The illustrated gas turbine engine **150** includes a compressor section **152**, a combustor section **154**, and an expander section or turbine section **156**. The compressor section **152** includes one or more exhaust gas compressors or compressor stages **158**, such as 1 to 20 stages of rotary compressor blades disposed in a series arrangement. Likewise, the combustor section **154** includes one or more combustors **160**, such as 1 to 20 combustors **160** distributed circumferentially about a rotational axis **162** of the SEGR gas turbine system **52**. Furthermore, each combustor **160** may include one or more fuel nozzles **164** configured to inject the exhaust gas **66**, the oxidant **68**, and/or the fuel **70**. For example, a head end portion **166** of each combustor **160** may house **1, 2, 3, 4, 5, 6**, or more fuel nozzles **164**, which may inject streams or mixtures of the exhaust gas **66**, the oxidant **68**, and/or the fuel **70** into a combustion portion **168** (e.g., combustion chamber) of the combustor **160**.

[0051] The fuel nozzles **164** may include any combination of premix fuel nozzles **164** (e.g., configured to premix the oxidant **68** and fuel **70** for generation of an oxidant/fuel premix flame) and/or diffusion fuel nozzles **164** (e.g., configured to inject separate flows of the oxidant **68** and fuel **70** for generation of an oxidant/fuel diffusion flame). Embodiments of the premix fuel nozzles **164** may include swirl vanes, mixing chambers, or other features to internally mix the oxidant **68** and fuel **70** within the nozzles **164**, prior to injection and combustion in the combustion chamber **168**. The premix fuel nozzles **164** also may receive at least some partially

mixed oxidant **68** and fuel **70**. In certain embodiments, each diffusion fuel nozzle **164** may isolate flows of the oxidant **68** and the fuel **70** until the point of injection, while also isolating flows of one or more diluents (e.g., the exhaust gas **66**, steam, nitrogen, or another inert gas) until the point of injection. In other embodiments, each diffusion fuel nozzle **164** may isolate flows of the oxidant **68** and the fuel **70** until the point of injection, while partially mixing one or more diluents (e.g., the exhaust gas **66**, steam, nitrogen, or another inert gas) with the oxidant **68** and/or the fuel **70** prior to the point of injection. In addition, one or more diluents (e.g., the exhaust gas **66**, steam, nitrogen, or another inert gas) may be injected into the combustor (e.g., into the hot products of combustion) either at or downstream from the combustion zone, thereby helping to reduce the temperature of the hot products of combustion and reduce emissions of NO_x (e.g., NO and NO₂). Regardless of the type of fuel nozzle **164**, the SEGR gas turbine system **52** may be controlled to provide substantially stoichiometric combustion of the oxidant **68** and fuel **70**.

[0052] In diffusion combustion embodiments using the diffusion fuel nozzles **164**, the fuel **70** and oxidant **68** generally do not mix upstream from the diffusion flame, but rather the fuel **70** and oxidant **68** mix and react directly at the flame surface and/or the flame surface exists at the location of mixing between the fuel **70** and oxidant **68**. In particular, the fuel **70** and oxidant **68** separately approach the flame surface (or diffusion boundary/interface), and then diffuse (e.g., via molecular and viscous diffusion) along the flame surface (or diffusion boundary/interface) to generate the diffusion flame. It is noteworthy that the fuel **70** and oxidant **68** may be at a substantially stoichiometric ratio along this flame surface (or diffusion boundary/interface), which may result in a greater flame temperature (e.g., a peak flame temperature) along this flame surface. The stoichiometric fuel/oxidant ratio generally results in a greater flame temperature (e.g., a peak flame temperature), as compared with a fuel-lean or fuel-rich fuel/oxidant ratio. As a result, the diffusion flame may be substantially more stable than a premix flame, because the diffusion of fuel **70** and oxidant **68** helps to maintain a stoichiometric ratio (and greater temperature) along the flame surface. Although greater flame temperatures can also lead to greater exhaust emissions, such as NO_x emissions, the disclosed embodiments use one or more diluents to help control the temperature and emissions while still avoiding any premixing of the fuel **70** and oxidant **68**. For example, the disclosed embodiments may introduce one or more diluents separate from the fuel **70** and oxidant **68** (e.g., after the point of combustion and/or downstream from the diffusion flame), thereby helping to reduce the temperature and reduce the emissions (e.g., NO_x emissions) produced by the diffusion flame.

[0053] In operation, as illustrated, the compressor section **152** receives and compresses the exhaust gas **66** from the EG processing system **54**, and outputs a compressed exhaust gas **170** to each of the combustors **160** in the combustor section **154**. Upon combustion of the fuel **60**, oxidant **68**, and exhaust gas **170** within each combustor **160**, additional exhaust gas or products of combustion **172** (i.e., combustion gas) is routed into the turbine section **156**. Similar to the compressor section **152**, the turbine section **156** includes one or more turbines or turbine stages **174**, which may include a series of rotary turbine blades. These turbine blades are then driven by the products of combustion **172** generated in the combustor section **154**, thereby driving rotation of a shaft **176** coupled to the

machinery 106. Again, the machinery 106 may include a variety of equipment coupled to either end of the SEGR gas turbine system 52, such as machinery 106, 178 coupled to the turbine section 156 and/or machinery 106, 180 coupled to the compressor section 152. In certain embodiments, the machinery 106, 178, 180 may include one or more electrical generators, oxidant compressors for the oxidant 68, fuel pumps for the fuel 70, gear boxes, or additional drives (e.g. steam turbine 104, electrical motor, etc.) coupled to the SEGR gas turbine system 52. Non-limiting examples are discussed in further detail below with reference to TABLE 1. As illustrated, the turbine section 156 outputs the exhaust gas 60 to recirculate along the exhaust recirculation path 110 from an exhaust outlet 182 of the turbine section 156 to an exhaust inlet 184 into the compressor section 152. Along the exhaust recirculation path 110, the exhaust gas 60 passes through the EG processing system 54 (e.g., the HRSG 56 and/or the EGR system 58) as discussed in detail above.

[0054] Again, each combustor 160 in the combustor section 154 receives, mixes, and stoichiometrically combusts the compressed exhaust gas 170, the oxidant 68, and the fuel 70 to produce the additional exhaust gas or products of combustion 172 to drive the turbine section 156. In certain embodiments, the oxidant 68 is compressed by an oxidant compression system 186, such as a main oxidant compression (MOC) system (e.g., a main air compression (MAC) system) having one or more oxidant compressors (MOCs). The oxidant compression system 186 includes an oxidant compressor 188 coupled to a drive 190. For example, the drive 190 may include an electric motor, a combustion engine, or any combination thereof. In certain embodiments, the drive 190 may be a turbine engine, such as the gas turbine engine 150. Accordingly, the oxidant compression system 186 may be an integral part of the machinery 106. In other words, the compressor 188 may be directly or indirectly driven by the mechanical power 72 supplied by the shaft 176 of the gas turbine engine 150. In such an embodiment, the drive 190 may be excluded, because the compressor 188 relies on the power output from the turbine engine 150. However, in certain embodiments employing more than one oxidant compressor is employed, a first oxidant compressor (e.g., a low pressure (LP) oxidant compressor) may be driven by the drive 190 while the shaft 176 drives a second oxidant compressor (e.g., a high pressure (HP) oxidant compressor), or vice versa. For example, in another embodiment, the HP MOC is driven by the drive 190 and the LP oxidant compressor is driven by the shaft 176. In the illustrated embodiment, the oxidant compression system 186 is separate from the machinery 106. In each of these embodiments, the compression system 186 compresses and supplies the oxidant 68 to the fuel nozzles 164 and the combustors 160. Accordingly, some or all of the machinery 106, 178, 180 may be configured to increase the operational efficiency of the compression system 186 (e.g., the compressor 188 and/or additional compressors).

[0055] The variety of components of the machinery 106, indicated by element numbers 106A, 106B, 106C, 106D, 106E, and 106F, may be disposed along the line of the shaft 176 and/or parallel to the line of the shaft 176 in one or more series arrangements, parallel arrangements, or any combination of series and parallel arrangements. For example, the machinery 106, 178, 180 (e.g., 106A through 106F) may include any series and/or parallel arrangement, in any order, of: one or more gearboxes (e.g., parallel shaft, epicyclic gearboxes), one or more compressors (e.g., oxidant compressors,

booster compressors such as EG booster compressors), one or more power generation units (e.g., electrical generators), one or more drives (e.g., steam turbine engines, electrical motors), heat exchange units (e.g., direct or indirect heat exchangers), clutches, or any combination thereof. The compressors may include axial compressors, radial or centrifugal compressors, or any combination thereof, each having one or more compression stages. Regarding the heat exchangers, direct heat exchangers may include spray coolers (e.g., spray intercoolers), which inject a liquid spray into a gas flow (e.g., oxidant flow) for direct cooling of the gas flow. Indirect heat exchangers may include at least one wall (e.g., a shell and tube heat exchanger) separating first and second flows, such as a fluid flow (e.g., oxidant flow) separated from a coolant flow (e.g., water, air, refrigerant, or any other liquid or gas coolant), wherein the coolant flow transfers heat from the fluid flow without any direct contact. Examples of indirect heat exchangers include intercooler heat exchangers and heat recovery units, such as heat recovery steam generators. The heat exchangers also may include heaters. As discussed in further detail below, each of these machinery components may be used in various combinations as indicated by the non-limiting examples set forth in TABLE 1.

[0056] Generally, the machinery 106, 178, 180 may be configured to increase the efficiency of the compression system 186 by, for example, adjusting operational speeds of one or more oxidant compressors in the system 186, facilitating compression of the oxidant 68 through cooling, and/or extraction of surplus power. The disclosed embodiments are intended to include any and all permutations of the foregoing components in the machinery 106, 178, 180 in series and parallel arrangements, wherein one, more than one, all, or none of the components derive power from the shaft 176. As illustrated below, TABLE 1 depicts some non-limiting examples of arrangements of the machinery 106, 178, 180 disposed proximate and/or coupled to the compressor and turbine sections 152, 156.

TABLE 1

106A	106B	106C	106D	106E	106F
MOC	GEN				
MOC	GBX	GEN			
LP	HP	GEN			
MOC	MOC				
HP	GBX	LP	GEN		
MOC		MOC			
MOC	GBX	GEN			
MOC					
HP	GBX	GEN	LP		
MOC			MOC		
MOC	GBX	GEN			
MOC	GBX	DRV			
DRV	GBX	LP	HP	GBX	GEN
		MOC	MOC		
DRV	GBX	HP	LP	GEN	
		MOC	MOC		
HP	GBX	LP	GEN		
MOC	CLR	MOC			
HP	GBX	LP	GBX	GEN	
MOC	CLR	MOC			
HP	GBX	LP	GEN		
MOC	HTR	MOC			
	STGN				
MOC	GEN	DRV			
MOC	DRV	GEN			
DRV	MOC	GEN			

TABLE 1-continued

106A	106B	106C	106D	106E	106F
DRV	CLU	MOC	GEN		
DRV	CLU	MOC	GBX	GEN	

[0057] As illustrated above in TABLE 1, a cooling unit is represented as CLR, a clutch is represented as CLU, a drive is represented by DRV, a gearbox is represented as GBX, a generator is represented by GEN, a heating unit is represented by HTR, a main oxidant compressor unit is represented by MOC, with low pressure and high pressure variants being represented as LP MOC and HP MOC, respectively, and a steam generator unit is represented as STGN. Although TABLE 1 illustrates the machinery **106, 178, 180** in sequence toward the compressor section **152** or the turbine section **156**, TABLE 1 is also intended to cover the reverse sequence of the machinery **106, 178, 180**. In TABLE 1, any cell including two or more components is intended to cover a parallel arrangement of the components. TABLE 1 is not intended to exclude any non-illustrated permutations of the machinery **106, 178, 180**. These components of the machinery **106, 178, 180** may enable feedback control of temperature, pressure, and flow rate of the oxidant **68** sent to the gas turbine engine **150**. As discussed in further detail below, the oxidant **68** and the fuel **70** may be supplied to the gas turbine engine **150** at locations specifically selected to facilitate isolation and extraction of the compressed exhaust gas **170** without any oxidant **68** or fuel **70** degrading the quality of the exhaust gas **170**.

[0058] The EG supply system **78**, as illustrated in FIG. 3, is disposed between the gas turbine engine **150** and the target systems (e.g., the hydrocarbon production system **12** and the other systems **84**). In particular, the EG supply system **78**, e.g., the EG extraction system (EGES) **80**, may be coupled to the gas turbine engine **150** at one or more extraction points **76** along the compressor section **152**, the combustor section **154**, and/or the turbine section **156**. For example, the extraction points **76** may be located between adjacent compressor stages, such as 2, 3, 4, 5, 6, 7, 8, 9, or 10 interstage extraction points **76** between compressor stages. Each of these interstage extraction points **76** provides a different temperature and pressure of the extracted exhaust gas **42**. Similarly, the extraction points **76** may be located between adjacent turbine stages, such as 2, 3, 4, 5, 6, 7, 8, 9, or 10 interstage extraction points **76** between turbine stages. Each of these interstage extraction points **76** provides a different temperature and pressure of the extracted exhaust gas **42**. By further example, the extraction points **76** may be located at a multitude of locations throughout the combustor section **154**, which may provide different temperatures, pressures, flow rates, and gas compositions. Each of these extraction points **76** may include an EG extraction conduit, one or more valves, sensors, and controls, which may be used to selectively control the flow of the extracted exhaust gas **42** to the EG supply system **78**.

[0059] The extracted exhaust gas **42**, which is distributed by the EG supply system **78**, has a controlled composition suitable for the target systems (e.g., the hydrocarbon produc-

tion system **12** and the other systems **84**). For example, at each of these extraction points **76**, the exhaust gas **170** may be substantially isolated from injection points (or flows) of the oxidant **68** and the fuel **70**. In other words, the EG supply system **78** may be specifically designed to extract the exhaust gas **170** from the gas turbine engine **150** without any added oxidant **68** or fuel **70**. Furthermore, in view of the stoichiometric combustion in each of the combustors **160**, the extracted exhaust gas **42** may be substantially free of oxygen and fuel. The EG supply system **78** may route the extracted exhaust gas **42** directly or indirectly to the hydrocarbon production system **12** and/or other systems **84** for use in various processes, such as enhanced oil recovery, carbon sequestration, storage, or transport to an offsite location. However, in certain embodiments, the EG supply system **78** includes the EG treatment system (EGTS) **82** for further treatment of the exhaust gas **42**, prior to use with the target systems. For example, the EG treatment system **82** may purify and/or separate the exhaust gas **42** into one or more streams **95**, such as the CO₂ rich, N₂ lean stream **96**, the intermediate concentration CO₂, N₂ stream **97**, and the CO₂ lean, N₂ rich stream **98**. These treated exhaust gas streams **95** may be used individually, or in any combination, with the hydrocarbon production system **12** and the other systems **84** (e.g., the pipeline **86**, the storage tank **88**, and the carbon sequestration system **90**).

[0060] Similar to the exhaust gas treatments performed in the EG supply system **78**, the EG processing system **54** may include a plurality of exhaust gas (EG) treatment components **192**, such as indicated by element numbers **194, 196, 198, 200, 202, 204, 206, 208, and 210**. These EG treatment components **192** (e.g., **194** through **210**) may be disposed along the exhaust recirculation path **110** in one or more series arrangements, parallel arrangements, or any combination of series and parallel arrangements. For example, the EG treatment components **192** (e.g., **194** through **210**) may include any series and/or parallel arrangement, in any order, of: one or more heat exchangers (e.g., heat recovery units such as heat recovery steam generators, condensers, coolers, or heaters), catalyst systems (e.g., oxidation catalyst systems), particulate and/or water removal systems (e.g., inertial separators, coalescing filters, water impermeable filters, and other filters), chemical injection systems, solvent based treatment systems (e.g., absorbers, flash tanks, etc.), carbon capture systems, gas separation systems, gas purification systems, and/or a solvent based treatment system, or any combination thereof. In certain embodiments, the catalyst systems may include an oxidation catalyst, a carbon monoxide reduction catalyst, a nitrogen oxides reduction catalyst, an aluminum oxide, a zirconium oxide, a silicone oxide, a titanium oxide, a platinum oxide, a palladium oxide, a cobalt oxide, or a mixed metal oxide, or a combination thereof. The disclosed embodiments are intended to include any and all permutations of the foregoing components **192** in series and parallel arrangements. As illustrated below, TABLE 2 depicts some non-limiting examples of arrangements of the components **192** along the exhaust recirculation path **110**.

TABLE 2

194	196	198	200	202	204	206	208	210
CU	HRU	BB	MRU	PRU				
CU	HRU	HRU	BB	MRU	PRU	DIL		
CU	HRSG	HRSG	BB	MRU	PRU			

TABLE 2-continued

194	196	198	200	202	204	206	208	210
OCU	HRU	OCU	HRU	OCU	BB	MRU	PRU	
HRU	HRU	BB	MRU	PRU				
CU	CU							
HRSG	HRSG	BB	MRU	PRU	DIL			
OCU	OCU							
OCU	HRSG	OCU	HRSG	OCU	BB	MRU	PRU	DIL
	OCU		OCU					
OCU	HRSG	HRSG	BB	COND	INER	WFIL	CFIL	DIL
	ST	ST						
OCU	OCU	BB	COND	INER	FIL	DIL		
HRSG	HRSG							
ST	ST							
OCU	HRSG	HRSG	OCU	BB	MRU	MRU	PRU	PRU
	ST	ST			HE	WFIL	INER	FIL
					COND			CFIL
CU	HRU	HRU	HRU	BB	MRU	PRU	PRU	DIL
	COND	COND	COND		HE	INER	FIL	
					COND		CFIL	
					WFIL			

[0061] As illustrated above in TABLE 2, a catalyst unit is represented by CU, an oxidation catalyst unit is represented by OCU, a booster blower is represented by BB, a heat exchanger is represented by HX, a heat recovery unit is represented by HRU, a heat recovery steam generator is represented by HRSG, a condenser is represented by COND, a steam turbine is represented by ST, a particulate removal unit is represented by PRU, a moisture removal unit is represented by MRU, a filter is represented by FIL, a coalescing filter is represented by CFIL, a water impermeable filter is represented by WFIL, an inertial separator is represented by INER, and a diluent supply system (e.g., steam, nitrogen, or other inert gas) is represented by DIL. Although TABLE 2 illustrates the components 192 in sequence from the exhaust outlet 182 of the turbine section 156 toward the exhaust inlet 184 of the compressor section 152, TABLE 2 is also intended to cover the reverse sequence of the illustrated components 192. In TABLE 2, any cell including two or more components is intended to cover an integrated unit with the components, a parallel arrangement of the components, or any combination thereof. Furthermore, in context of TABLE 2, the HRU, the HRSG, and the COND are examples of the HE; the HRSG is an example of the HRU; the COND, WFIL, and CFIL are examples of the WRU; the INER, FIL, WFIL, and CFIL are examples of the PRU; and the WFIL and CFIL are examples of the FIL. Again, TABLE 2 is not intended to exclude any non-illustrated permutations of the components 192. In certain embodiments, the illustrated components 192 (e.g., 194 through 210) may be partially or completely integrated within the HRSG 56, the EGR system 58, or any combination thereof. These EG treatment components 192 may enable feedback control of temperature, pressure, flow rate, and gas composition, while also removing moisture and particulates from the exhaust gas 60. Furthermore, the treated exhaust gas 60 may be extracted at one or more extraction points 76 for use in the EG supply system 78 and/or recirculated to the exhaust inlet 184 of the compressor section 152.

[0062] As the treated, recirculated exhaust gas 66 passes through the compressor section 152, the SEGR gas turbine system 52 may bleed off a portion of the compressed exhaust gas along one or more lines 212 (e.g., bleed conduits or bypass conduits). Each line 212 may route the exhaust gas into one or more heat exchangers 214 (e.g., cooling units),

thereby cooling the exhaust gas for recirculation back into the SEGR gas turbine system 52. For example, after passing through the heat exchanger 214, a portion of the cooled exhaust gas may be routed to the turbine section 156 along line 212 for cooling and/or sealing of the turbine casing, turbine shrouds, bearings, and other components. In such an embodiment, the SEGR gas turbine system 52 does not route any oxidant 68 (or other potential contaminants) through the turbine section 156 for cooling and/or sealing purposes, and thus any leakage of the cooled exhaust gas will not contaminate the hot products of combustion (e.g., working exhaust gas) flowing through and driving the turbine stages of the turbine section 156. By further example, after passing through the heat exchanger 214, a portion of the cooled exhaust gas may be routed along line 216 (e.g., return conduit) to an upstream compressor stage of the compressor section 152, thereby improving the efficiency of compression by the compressor section 152. In such an embodiment, the heat exchanger 214 may be configured as an interstage cooling unit for the compressor section 152. In this manner, the cooled exhaust gas helps to increase the operational efficiency of the SEGR gas turbine system 52, while simultaneously helping to maintain the purity of the exhaust gas (e.g., substantially free of oxidant and fuel).

[0063] FIG. 4 is a flow chart of an embodiment of an operational process 220 of the system 10 illustrated in FIGS. 1-3. In certain embodiments, the process 220 may be a computer implemented process, which accesses one or more instructions stored on the memory 122 and executes the instructions on the processor 120 of the controller 118 shown in FIG. 2. For example, each step in the process 220 may include instructions executable by the controller 118 of the control system 100 described with reference to FIG. 2.

[0064] The process 220 may begin by initiating a startup mode of the SEGR gas turbine system 52 of FIGS. 1-3, as indicated by block 222. For example, the startup mode may involve a gradual ramp up of the SEGR gas turbine system 52 to maintain thermal gradients, vibration, and clearance (e.g., between rotating and stationary parts) within acceptable thresholds. For example, during the startup mode 222, the process 220 may begin to supply a compressed oxidant 68 to the combustors 160 and the fuel nozzles 164 of the combustor section 154, as indicated by block 224. In certain embodi-

ments, the compressed oxidant may include a compressed air, oxygen, oxygen-enriched air, oxygen-reduced air, oxygen-nitrogen mixtures, or any combination thereof. For example, the oxidant **68** may be compressed by the oxidant compression system **186** illustrated in FIG. 3. The process **220** also may begin to supply fuel to the combustors **160** and the fuel nozzles **164** during the startup mode **222**, as indicated by block **226**. During the startup mode **222**, the process **220** also may begin to supply exhaust gas (as available) to the combustors **160** and the fuel nozzles **164**, as indicated by block **228**. For example, the fuel nozzles **164** may produce one or more diffusion flames, premix flames, or a combination of diffusion and premix flames. During the startup mode **222**, the exhaust gas **60** being generated by the gas turbine engine **156** may be insufficient or unstable in quantity and/or quality. Accordingly, during the startup mode, the process **220** may supply the exhaust gas **66** from one or more storage units (e.g., storage tank **88**), the pipeline **86**, other SEGR gas turbine systems **52**, or other exhaust gas sources.

[0065] The process **220** may then combust a mixture of the compressed oxidant, fuel, and exhaust gas in the combustors **160** to produce hot combustion gas **172**, as indicated by block **230** by the one or more diffusion flames, premix flames, or a combination of diffusion and premix flames. In particular, the process **220** may be controlled by the control system **100** of FIG. 2 to facilitate stoichiometric combustion (e.g., stoichiometric diffusion combustion, premix combustion, or both) of the mixture in the combustors **160** of the combustor section **154**. However, during the startup mode **222**, it may be particularly difficult to maintain stoichiometric combustion of the mixture (and thus low levels of oxidant and unburnt fuel may be present in the hot combustion gas **172**). As a result, in the startup mode **222**, the hot combustion gas **172** may have greater amounts of residual oxidant **68** and/or fuel **70** than during a steady state mode as discussed in further detail below. For this reason, the process **220** may execute one or more control instructions to reduce or eliminate the residual oxidant **68** and/or fuel **70** in the hot combustion gas **172** during the startup mode.

[0066] The process **220** then drives the turbine section **156** with the hot combustion gas **172**, as indicated by block **232**. For example, the hot combustion gas **172** may drive one or more turbine stages **174** disposed within the turbine section **156**. Downstream of the turbine section **156**, the process **220** may treat the exhaust gas **60** from the final turbine stage **174**, as indicated by block **234**. For example, the exhaust gas treatment **234** may include filtration, catalytic reaction of any residual oxidant **68** and/or fuel **70**, chemical treatment, heat recovery with the HRSG **56**, and so forth. The process **220** may also recirculate at least some of the exhaust gas **60** back to the compressor section **152** of the SEGR gas turbine system **52**, as indicated by block **236**. For example, the exhaust gas recirculation **236** may involve passage through the exhaust recirculation path **110** having the EG processing system **54** as illustrated in FIGS. 1-3.

[0067] In turn, the recirculated exhaust gas **66** may be compressed in the compressor section **152**, as indicated by block **238**. For example, the SEGR gas turbine system **52** may sequentially compress the recirculated exhaust gas **66** in one or more compressor stages **158** of the compressor section **152**. Subsequently, the compressed exhaust gas **170** may be supplied to the combustors **160** and fuel nozzles **164**, as indicated by block **228**. Steps **230**, **232**, **234**, **236**, and **238** may then repeat, until the process **220** eventually transitions

to a steady state mode, as indicated by block **240**. Upon the transition **240**, the process **220** may continue to perform the steps **224** through **238**, but may also begin to extract the exhaust gas **42** via the EG supply system **78**, as indicated by block **242**. For example, the exhaust gas **42** may be extracted from one or more extraction points **76** along the compressor section **152**, the combustor section **154**, and the turbine section **156** as indicated in FIG. 3. In turn, the process **220** may supply the extracted exhaust gas **42** from the EG supply system **78** to the hydrocarbon production system **12**, as indicated by block **244**. The hydrocarbon production system **12** may then inject the exhaust gas **42** into the earth **32** for enhanced oil recovery, as indicated by block **246**. For example, the extracted exhaust gas **42** may be used by the exhaust gas injection EOR system **112** of the EOR system **18** illustrated in FIGS. 1-3.

[0068] In some embodiments of the SEGR gas turbine system **52**, the fuel-to-oxidant ratio (FOR) for the one or more combustors **160** is controlled independently from control of the exhaust gas-to oxidant ratio (EGOR) for the one or more combustors **160**. FIG. 5 illustrates an embodiment of the SEGR gas turbine system **52** with one combustor **160** shown for clarity, however the SEGR gas turbine system **52** may include multiple (e.g., **2**, **3**, **4**, **8**, **12**, **16**, or more) combustors **160**. The combustor **160** receives an oxidant flow **260** at the head end **166** of the combustor **160** via an oxidant control system **262**. As discussed below, a fuel control system **264** controls a fuel flow **266** to the one or more fuel nozzles **164** of the combustor **160**. As may be appreciated, the oxidant control system **262** may include an industrial controller having a processor, memory, and instructions stored on memory and executable by the processor to perform control functions via valves and/or actuators to control the oxidant flow **260** to the one or more fuel nozzles **164** of each combustor **160**. Likewise, the fuel control system **264** may include an industrial controller having a processor, memory, and instructions stored on memory and executable by the processor to perform control functions via valves and/or actuators as discussed below to control the fuel flow **266** to the one or more fuel nozzles **164** of each combustor **160**. The controller **118** may control the FOR of the combustor **160** by controlling the oxidant flow **260** and/or the fuel flow **266**. The controller **118** may control the FOR to be approximately the stoichiometric fuel-to-oxidant ratio (FOR_{ST}) during steady state operation, thereby controlling the equivalence ratio to approximately 1.0, between approximately 0.90 to 1.10, or between approximately 0.95 to 1.05.

[0069] The controller **118** may be coupled to the oxidant control system **262** and to the fuel control system **264** via control lines **268**, and the controller **118** may be coupled to an oxidant sensing system **270** and/or to a fuel sensing system **272** via signal lines **273**. The oxidant sensing system **270** may include an industrial controller having a processor, a memory, and instructions stored on the memory and executable by the processor to perform sensing functions via the one or more sensors or measurement devices to monitor the oxidant flow **260**. Likewise, the fuel sensing system **272** may include an industrial controller having a processor, a memory, and instructions stored on the memory and executable by the processor to perform sensing functions via the one or more sensors or measurement devices to monitor the fuel flow **266**. The oxidant sensing system **270** and the fuel sensing system **272** may each include one or more sensors or measurement devices to measure composition, flow rate, temperature, and/

or pressure of the respective flows. For example, the one or more sensors of the oxidant sensing system 270 may include, but is not limited to, zirconia-based, electrochemical-based, infrared-based, ultrasonic-based, and laser-based oxygen sensors. Flow rate sensors or measurement devices of the oxidant sensing system 270 and/or fuel sensing system 272 may independently include one or more of any suitable type of flow analysis devices, including but not limited to, thermal mass flow meters, Coriolis mass flow meters, mass flow controllers, rotary piston meters, gear meters, Venturi meters, orifice plate meters, Dall tube meters, Pitot tube meters, multi-hole pressure probe meters, cone meters, optical flow meters, electromagnetic flow meters, or ultrasonic flow meters. Temperature sensors or measurement devices of the oxidant sensing system 270 and/or fuel sensing system 272 may independently include one or more of any suitable type of temperature analysis devices, including but not limited to, thermocouples, thermistors, resistance temperature detector, or any combination thereof. One or more combustion product sensors 274 may sense the composition, flow rate, temperature, and/or pressure of the exhaust gas 60 transmitted from the turbine section 156 to the exhaust gas processing system 54. Additionally, or in the alternative, the one or more combustion product sensors 274 may sense the composition, flow rate, temperature, and/or pressure of the combustion gases 172 transmitted from the one or more combustors 160 to the turbine section 156. The one or more combustion product sensors 274 may include, but are not limited to, lambda sensors. For example, the combustion product sensor 274 may include, but is not limited to, zirconia-based, electrochemical-based, infrared-based, ultrasonic-based, and laser-based oxygen sensors.

[0070] The controller 118 may control the oxidant flow 260 and the fuel flow 266 based at least in part on sensor feedback from the oxidant sensing system 270, the fuel sensing system 272, the combustion product sensor 274, or any combination thereof. In some embodiments, the controller 118 utilizes a feed forward control based on the compositions of the fuel 70 and the oxidant 68, the current flow rates of the fuel and/or the oxidant, and a desired FOR for a desired operating load and/or operating speed. Additionally, or in the alternative, the controller 118 utilizes a feedback control based on the composition of the combustion gas 172 as determined by the combustion product sensor 274 and/or a composition of a recirculation flow 276 of exhaust gas 60 as determined by a sensor 278. For example, the disclosed embodiments described herein may be used in combination with any and all of the embodiments set forth in U.S. Application No. 61/747, 194, filed on Dec. 28, 2012, and entitled "GAS TURBINE COMBUSTOR CONTROL SYSTEM," which is hereby incorporated by reference in its entirety.

[0071] The oxidant flow 260 is supplied to the head end 166 of the combustor 160, and the fuel flow 266 is supplied to the one or more fuel nozzles 164 of the combustor 160 at the head end 166. The one or more fuel nozzles 164 may be premix fuel nozzles, diffusion fuel nozzles, or any combination thereof. The recirculation flow 276 or exhaust gas 60 may be supplied to a recirculation inlet 277 at the head end 166 or the combustor portion 168 of the combustor 160. The recirculation flow 276 may include the compressed exhaust gas 170 from the compressor section 152. In some embodiments the recirculation flow 276 mixes with the oxidant flow 260 and the fuel flow 266 in the head end 166, thereby diluting the oxidant concentration in the mixture. This recirculation flow 276 may

increase the combustion of residual oxidant and/or residual fuel within the recirculation flow 276 within the combustor 160. Additionally, or in the alternative, the recirculation flow 276 flows about the combustor 160 within a flow sleeve 280 to cool a combustor liner 282. Moreover, the recirculation flow 276 may increase the quantity or ratio of carbon dioxide within the recirculation flow 276 that may be extracted, such as for carbon sequestration, refrigeration system, or other target system. The recirculation flow 276 within the flow sleeve 280 may enter the combustor 160 as shown by the dashed arrows through dilution holes to mix with and to cool the combustion products 172 within the combustor portion 168. In some embodiments, cooling the combustion products 172 with the recirculation flow 276 reduces NO_x emissions. The recirculation flow 276 may flow to the head end 166 to mix with the oxidant 68 and the fuel 70, as described above.

[0072] The combustion products 172 from the combustion of the fuel flow 266 and the oxidant flow 260 and a portion of the recirculation flow 276 are directed to the turbine section 156 of the SEGR gas turbine system 52, which may drive the machinery 106 and/or the compressor section 152 via the shaft 176. The exhaust gas 60 exiting the turbine section 156 flows to the exhaust gas processing system 54, which recirculates the exhaust gas 60 to the compressor section 152. As discussed above, the exhaust gas processing system 54 may extract heat from the exhaust gas 60, such as via the HRSG 56. The exhaust gas processing system 54 directs the cooled exhaust gas 60 to the compressor section 152 for compression and recirculation to the one or more combustors 160. In some embodiments, the controller 118 is coupled to one or more sets of inlet guide vanes 284 to control the pressure and/or the flow rate of the recirculation flow 276 from the compressor section 152 to the one or more combustors 160.

[0073] As discussed herein, the recirculation flow 276 supplied to the one or more combustors 160 includes a first portion 288 extracted to an extraction manifold 286 via one or more extraction ports 287, and a second portion that mixes with the oxidant 68 and the fuel 70 in the one or more combustors 160. The second portion mixed with the oxidant 68 and the fuel 70 flows to the turbine section 156 as the combustion gases 172. From the extraction manifold 286, the first portion 288 of the recirculation flow 276 may be subdivided into a third portion 289 and a bypass flow 292. The extraction manifold 286 is coupled to the one or more combustors 160. The extraction manifold 286 may be fluidly coupled to one or more exhaust gas supply systems 78, which receive the third portion 289 of the recirculation flow 276. As described above with FIGS. 1-4, the one or more exhaust gas supply systems 78 may utilize the third portion 289 (e.g., exhaust gas 42) in an enhanced oil recovery system 18 and/or may supply the third portion 289 to a pipeline 86, storage tank 88, or to a carbon sequestration system 90. The quantity of the recirculation flow 276 supplied to the one or more exhaust gas supply systems 78 as the third portion 289 may vary based at least in part on the exhaust gas demands of the EOR system 18 or other systems 84. For example, the flow rate of the third portion 289 may be a fixed or variable flow rate. In some embodiments, the first portion 288 is extracted from the flow sleeve 280 about the combustor 160 prior to mixing with the oxidant flow 260 and/or the fuel flow 266. The recirculation flow 276 may flow upstream within the flow sleeve 280 relative to the flow of combustion gases 172. The first portion 288 of the recirculation flow 276 may have less than approximately 10, 20, 30, 40, 50, 60, 70, 80, 90, 100, 200, 300, 400,

500, 1000, 2000, 3000, 4000, or 5000 parts per million by volume (ppmv) of oxidant (e.g., oxygen), unburnt fuel or hydrocarbons (e.g., HCs), nitrogen oxides (e.g., NO_x), carbon monoxide (CO), sulfur oxides (e.g., SO_x), hydrogen, and other products of incomplete combustion. For example, the first portion 288 may have a concentration of less than approximately 10, 20, 30, 40, 50, 75, or 100 ppmv of oxidant. In some embodiments, the first portion 288 may include part of the combustion gas flow 172.

[0074] The controller 118 may control the EGOR for a combustor 160 by adjusting the recirculation flow 276 supplied to the combustor 160 and/or by adjusting the first portion 288 of the recirculation flow 276 extracted from the combustor 160. The controller 118 may adjust the recirculation flow 276 supplied to the combustor 160 by adjusting (e.g., opening, closing) the inlet guide vanes 284 and/or adjusting (e.g., increasing, decreasing) the amount of exhaust gas 60 from the exhaust gas processing system 54 that is recirculated to the compressor section 152. Opening the inlet guide vanes 284 may increase the oxidant 68 within the recirculation flow 276, thereby decreasing the EGOR, and closing the inlet guide vanes 284 may decrease the oxidant 68 within the recirculation flow 276, thereby increasing the EGOR. Increasing the amount of exhaust gas 60 from the exhaust gas processing system 54 (e.g., via opening valves) may increase the exhaust gas 60 within the recirculation flow 276, thereby increasing the EGOR, and decreasing the amount of exhaust gas 60 from the exhaust gas processing system 54 (e.g., via closing valves) may decrease the exhaust gas 60 within the recirculation flow 276, thereby decreasing the EGOR. In some embodiments, the controller 118 may control the EGOR by controlling the first portion 288 extracted from the combustor 160, thereby controlling the remainder of the recirculation flow 276 in the combustor 160 that mixes with the oxidant flow 260 and/or the fuel flow 266. The controller 118 may adjust (e.g., increase, decrease) the first portion 288 of the recirculation flow 276 extracted from the combustor 160 by adjusting a turbine bypass valve 290. The turbine bypass valve 290 is fluidly coupled between the extraction manifold 286 and the exhaust gas flow 60 downstream of the turbine section 256. The controller 118 controls the turbine bypass valve 290 to adjust (e.g., increase, decrease) the bypass flow 292 of the first portion 288 that bypasses the turbine section 156 and joins the exhaust gas flow 60. Where a flow rate of the third portion 289 is substantially maintained, closing the turbine bypass valve 290 to decrease the bypass flow 292 may increase the EGOR in the one or more combustors 160 because less of the recirculation flow 276 is extracted prior to mixing with the oxidant 68. Likewise, opening the turbine bypass valve 290 to increase the bypass flow 292 while maintaining the flow rate of the third portion 289 may decrease the EGOR in the one or more combustors 160 because more of the recirculation flow 276 is extracted from the one or more combustors 160. As may be appreciated, adjustments to the flow rate of the third portion 289 may have a longer response time than a response time (e.g., less than 5, 4, 3, 2, or 1 second) for the turbine bypass valve 290. The controller 118 may control the flow rate of the third portion 289 via a control valve (e.g., ball valve, butterfly valve) of the extraction manifold 286, the exhaust gas supply system 78, or between the extraction manifold 286 and the exhaust gas supply system 78.

[0075] One or more turbine bypass valves 290 controlled by the controller 118 may be arranged to control (e.g.,

increase, decrease) one or more bypass flows 292 around the turbine section 156. In some embodiments, one extraction manifold 286 may receive the first portion 288 of the recirculation flow 276 from each of the one or more combustors 160, and a corresponding turbine bypass valve 290 may be fluidly coupled to the one extraction manifold 286. On other embodiments, multiple extraction manifolds 286 may receive the first portion 288 of the recirculation flow 276 from one or more combustors 160. For example, a first extraction manifold 286 may supply its third portion 289 to the enhanced oil recovery system 18, and a second extraction manifold 286 may supply its respective third portion 289 to a carbon sequestration system 90. One or more of the multiple extraction manifolds 286 may be coupled to a turbine bypass valve 290, and each turbine bypass valve 290 may control a respective bypass flow 292 that joins the exhaust gas flow 60 from the turbine section 156.

[0076] The bypass flow 292 may be recirculated as the recirculation flow 276 with at least some of the exhaust gas 60 from the turbine section 156. In some embodiments, the bypass flow 292 may merge with the exhaust gas 60 downstream of the exhaust gas processing system 54. The bypass flow 292 may be at a different pressure and/or temperature than the exhaust gas 60. For example, the bypass flow 292 may be at a higher pressure than the exhaust gas 60 because the turbine section 256 did not expand the bypass flow 292. The bypass flow 292 may be at a lower temperature than the exhaust gas 60 because the exhaust gas 60 may include combustion products that reacted within the combustor 160 more recently than combustion products of the bypass flow 292. Accordingly, recirculation of the bypass flow 292 may reduce the work performed by the compressor section 152 to direct the recirculation flow 276 to the one or more combustors 160. In some embodiments, the turbine bypass valve 290 enables the controller 118 to control (e.g., increase, decrease) the EGOR, while utilizing the bypass flow 292 within the SEGR gas turbine system 52 for later recirculation and/or extraction.

[0077] The controller 118 may control the inlet guide vanes 284 and/or the turbine bypass valve 290 to control (e.g., increase, decrease) the EGOR to a desired value and to maintain combustor stability within an operating margin. Adjustment to the inlet guide vanes 284 may be relatively slow (e.g., between 3 and 5 seconds), which may be less suitable for responding to rapid changes in the EGOR than adjustment to the turbine bypass valve 290. The bypass flow 292 may be more rapidly adjusted by the controller 118 via the turbine bypass valve 290. Indeed, adjustments to the bypass flow 292 may more rapidly address various changes in the EGOR than adjustments to the inlet guide vanes 284. In some embodiments, a bypass flow sensor 293 may measure the temperature, pressure, flow rate, and/or composition of the bypass flow 292 through the turbine bypass valve 290.

[0078] In some embodiments, the controller 118 may adjust (e.g., increase, decrease) the bulk EGOR for the one or more combustors 160 together, such as via the turbine bypass valve 290, whereas the controller 118 may adjust (e.g., increase, decrease) the FOR of the one or more combustors 160 individually (e.g., can-to-can adjustment). The controller 118 may control (e.g., increase, decrease) the EGOR, as discussed above, based at least in part on the composition of the combustion gases 172 as measured by the combustion product sensor 274, the flow rate of the oxidant flow 260, the flow rate of the fuel flow 266, a flow rate of the first portion 288 of the recirculation flow 276, a flow rate of the bypass

flow 292, or a flow rate of the third portion 289, or any combination thereof. In some embodiments, an extraction flow sensor 294 may measure the temperature, pressure, flow rate, and/or composition of the first portion 288 extracted from the one or more combustors 160.

[0079] FIG. 6 illustrates a chart 300 of a combustor operability limit 302 for an embodiment of a SEGR gas turbine system 52. The combustor operability limit 302 is a curve corresponding to the equivalence ratio 304 and a combustor operability value 306 of the SEGR gas turbine system 52 with certain combustion sustainability characteristics. The combustor operability value 306 may be based at least in part on properties of the non-fuel flow (e.g., exhaust gas 66, oxidant 68) into the combustor 160. The properties that may affect the combustor operability value 306 may include, but are not limited to, temperature, pressure, or oxidant concentration, or any combination thereof. The combustor operability value is proportional to the EGOR for each of the one or more combustors 160. Operating points 308 plotted on the chart 300 correspond to the equivalence ratio 304 and the combustor operability 306 during operation of the combustor 160. As may be appreciated, operating points 308 below the combustor operability limit 302 correspond to sustainable combustion of the fuel flow 266 and oxidant flow 260 with the recirculation flow 276. Combustion at operating points beyond the combustor operability limit 302 (as shown in the cross-hatched area 310) may have an increased probability and/or occurrences of flame blowout, flame flashback, auto-ignition of fuel flow 266 and oxidant flow 260, or induction of harmonic vibrations, or any combination thereof. In some embodiments, operating points 308 in area 310 may correspond to oxidant concentrations in the combustor 160 less than a threshold concentration for a desired level of sustained combustion. The combustor operability limit 302 and area 310 may vary between gas turbine systems due to differences in component configurations and geometries of the gas turbine systems, loading of the gas turbine systems, ambient conditions, and so forth. In some embodiments, the combustor operability limit 302 is determined for each SEGR gas turbine system 52 and/or for each combustor 160. Moreover, the combustor operability limit 302 may be based at least in part on sustainability thresholds as determined by the use and operation of the gas turbine system. That is, operators of the SEGR gas turbine system 52 may determine the combustor operability limit 302 based at least in part on weighting different factors, such as base load, emissions, combustion stability, and so forth. For example, a first gas turbine system in which emissions and flame stability are valued relatively highly (e.g., for driving a substantially constant load and/or for maintaining emissions below a desired level) may have the first combustor operability limit 302. A second gas turbine system for which emissions and flame stability are valued relatively less (e.g., for driving a periodic load) may have the second combustor operability limit 312.

[0080] An operating curve 314 illustrates an example of multiple operating points 308 of the SEGR gas turbine system 52 from a startup point to a steady-state operation point. As may be appreciated, the SEGR gas turbine system 52 described herein is not to be limited to the operating curve 314. In some embodiments, each combustor 160 of the SEGR gas turbine system 52 may have a separate operating curve 314, and each combustor 160 may have a separate combustor operability limit 302. Accordingly, the controller 118 may adjust the EGOR and/or the FOR for each combustor 160 to

maintain the operating point 308 at any time below the respective combustor operability limit 302. At a starting point 316 of a start-up sequence, the equivalence ratio 304 may be lean (e.g., approximately 0.5) such that the actual oxidant flow is greater than a stoichiometric oxidant flow. As may be appreciated, an external driver (e.g., electric motor, reciprocating engine) coupled to the gas turbine system 52 may drive the compressor section 152 and the turbine section 156 of the gas turbine system 52 at a low speed (e.g., less than 20% load speed) during a start-up sequence. At the starting point 316, components of the gas turbine system may be cool (e.g., approximately ambient temperature, between approximately 0° C. to approximately 45° C.) relative to peak operating temperatures. Fuel and oxidant introduced into the combustor 160 react with one another within the combustor 160, thereby heating components of the gas turbine system and generating exhaust gas 66. During a first startup operating region 318, the oxidant flow to the combustor 160 increases as the operating speed of the SEGR gas turbine system 52 increases, thereby decreasing the equivalence ratio 304. For example, the increasing speed of the compressor section 152 may increase the oxidant supplied to the combustor 160, thereby decreasing the equivalence ratio.

[0081] At a second operating point 320, the quantity of exhaust gas recirculated to the combustor 160 may increase. For example, components of the gas turbine system may warm up and/or the completeness of combustion of the fuel and the oxidant within the combustor 160 may increase, thereby reducing the oxidant concentration in the recirculated exhaust gas. The increased exhaust gas (e.g., carbon dioxide) and/or decreased oxidant concentration increases the combustor operability 306. Additionally, or in the alternative, the controller 118 may increase the exhaust gas in the combustor 160 via closing the turbine bypass valve 290 to control the products of combustion (e.g., emissions, oxidant, unburnt fuel) in the exhaust gas. In some embodiments, the operating speed of the gas turbine system may be between approximately 40 to 60 percent of full operating speed when operating in a second startup operating region 322.

[0082] When the operating curve 314 nears the combustor operability limit 302 (e.g., at a third operating point 324), the controller 118 increases the FOR and/or decreases the EGOR to maintain combustor stability. For example, the controller 118 may open the turbine bypass valve 290 to decrease the EGOR in the combustor 160, thereby reducing the combustor operability 306. Additionally, or in the alternative, the controller 118 increases the fuel flow to the combustor 160 as the components of the gas turbine system warm. In a third operating region 326, the controller 118 increases the operating speed of the gas turbine system and increases the fuel flow to the combustor 160 to prepare the gas turbine system for driving the load or machinery 106. A fourth operating region 328 illustrates a loading sequence in which the controller 118 controls the load on the gas turbine system, such as when the gas turbine system is connected to an electric grid. A fifth operating region 330 illustrates the combustor operability 306 and the equivalence ratio 304 for the SEGR gas turbine system 52 when operating at a substantially steady-state combustion equivalence ratio while driving the coupled machinery 106. As may be appreciated, the equivalence ratio 304 illustrated for the fifth operating region 330 is a rich value of approximately 1.02, which is between approximately 0.90 to 1.10 and approximately 0.95 to 1.05. The controller 118 may adjust the exhaust gas recirculation and combustor operabil-

ity **306** while substantially maintaining the equivalence ratio **304** in the fifth operating region **330**. As discussed below, the controller **118** may adjust (e.g., increase, decrease) the exhaust gas recirculation, and thereby the combustor operability **306**, to control the residual fuel, oxidant, and other emissions levels in the exhaust gas. As may be appreciated, the operating points in the fifth operating region **330** may have a greater margin from the combustor operability limit **302** than operating points in the second or third operating regions **322**, **326**.

[0083] As discussed herein, the controller **118** may control the EGOR independent from control of the FOR to maintain the operating point below the combustor operability limit **302**. The controller **118** may adjust the combustor operability **306** by adjusting the EGOR value via the turbine bypass valve **290** and/or the inlet guide vanes **284**. Because the EGOR value is proportional to the combustor operability **306**, the controller **118** may adjust the operating point **308** along the vertical axis of the chart **300** by controlling the EGOR. The controller **118** may adjust the equivalence ratio **304** by adjusting the FOR via controlling the fuel flow **266** and/or the oxidant flow **260**. Increasing the fuel flow **266** and/or decreasing the oxidant flow **260** causes the flow mixture within the combustor **160** to become more rich, corresponding to moving the operating point **308** to the right. Increasing the oxidant flow **260** and/or decreasing the fuel flow **266** causes the flow mixture within the combustor **160** to become more lean, corresponding to moving the operating point **308** to the left.

[0084] As may be appreciated, adjustments to the oxidant flow may affect the FOR and the EGOR. For example, increasing the oxidant flow **260** alone may decrease the equivalence ratio **304** and decrease the combustor operability **306**. Accordingly, the controller **118** may independently control the FOR and the EGOR by separately controlling the fuel flow and the exhaust gas recirculation flow. As discussed above, the controller **118** may control the EGOR and the combustor operability **306** via controlling the turbine bypass valve **290** and/or the inlet guide vanes **284**. Also, the controller **118** may control the FOR and the equivalence ratio **304** for one or more of the combustors **160** via the fuel control system **264** to the one or more combustors **160**.

[0085] FIG. 7 illustrates an embodiment of the fuel control system **264** coupled to multiple fuel nozzles **164**. In some embodiments, the SEGR gas turbine system **52** may have one or more groupings of fuel nozzles **164**, where each grouping includes one or more sets of fuel nozzles **164**. For example, the groupings may be combustor cans **350** arranged within an annulus **352** and/or in an annular arrangement, where fuel nozzles **164** of each combustor can **350** supply the fuel flow **266** that mixes with and combusts with the oxidant flow **260**. In some embodiments, the recirculation flow **276** may be separately supplied to the combustor can **350**. Each combustor can **350** may have a first nozzle set **354** of one or more fuel nozzles **164** and a second nozzle set **358** of one or more fuel nozzles **164**. Each combustor can **350** may include more than two nozzle sets. The one or more fuel nozzles **164** may be arranged in concentric rows, such that the first nozzle set **354** is a set of one or more inner fuel nozzles **164** arranged proximate to an axis **356** of the combustor can **350**, and the second nozzle set **358** is a set of one or more outer fuel nozzles **164** arranged proximate to a perimeter **360** of the combustor can **350**. Alternative embodiments of the first nozzle set **354** may include the one or more outer fuel nozzles **164** or another arrangement of the fuel nozzles **164**, such as the fuel nozzles

164 within a sector or quadrant of the combustor can **350**. That is, the first nozzle set **354** and the second nozzle set **358** may include, but are not limited to, the inner and outer arrangement discussed below.

[0086] The controller **118** controls the fuel control system **264** to supply the fuel flow to each of the one or more fuel nozzles **164**. In some embodiments, the controller **118** controls a fuel shut-off valve **362** coupled to a fuel supply **364**. As may be appreciated, the fuel shut-off valve **362** may be opened when the gas turbine system is in operation, and may be closed during a shutdown of the gas turbine system. A speed-ratio control valve **366** may provide a relatively gross control of the fuel flow to nozzle manifolds (e.g., first nozzle set manifold **378**, second nozzle set manifold **380**) and to the one or more fuel nozzles **164** based at least in part on the speed of the SEGR gas turbine system **52**.

[0087] In some embodiments, a first fuel supply **368** supplies a first fuel flow to the first set **354** of fuel nozzles (e.g., inner fuel nozzles), and a second fuel supply **370** supplies a second fuel flow to the second set **358** of fuel nozzles (e.g., outer fuel nozzles). The second fuel supply **370** may be coupled to a respective second fuel shut-off valve **372** and speed-ratio control valve **374** controlled by the controller **118**. The first fuel flow from may have a different heating value (e.g., a higher heating value) than the second fuel flow. For example, the first fuel flow may be supplied as a pilot fuel during a start up sequence, and the first fuel flow may be decreased during steady state operation while the second fuel flow, which has a relatively lower heating value, may be increased.

[0088] The first set **354** of fuel nozzles (e.g., inner fuel nozzles) of each of the combustor cans **350** may be coupled to a first nozzle set manifold **378**, and the second set **358** of fuel nozzles (e.g., outer fuel nozzles) of each of the combustor cans **350** may be coupled to a second nozzle set manifold **380**. In some embodiments, the first set **354** and the second set **358** of fuel nozzles utilize a common fuel supply (e.g., first fuel supply **368**). The controller **118** may control a distribution (e.g., flow ratio) of the fuel flow between the first nozzle set manifold **378** and the second nozzle set manifold **380** via distribution control valves **381**. For example, the controller **118** may direct more fuel flow to the second nozzle set manifold **380** than to the first nozzle set manifold **378** because there are more fuel nozzles **164** (e.g., outer fuel nozzles) in the second set **358** than in the first set **354** (e.g., inner fuel nozzles).

[0089] First fuel lines **384** couple the first set **354** of fuel nozzles **164** of each combustor can **350** to the first nozzle set manifold **378**, and second fuel lines **386** couple the second set **358** of fuel nozzles **164** of each combustor can **350** to the second nozzle set manifold **380**. For example, a first fuel line **384** supplies fuel to the inner (e.g., center) fuel nozzle **164** of a first combustor can **388**, and a second fuel line **386** supplies fuel to the outer (e.g., perimeter) fuel nozzles **164** of the first combustor can **388**. For clarity of illustration, FIG. 7 depicts only the first fuel lines **384** supplying the inner fuel nozzles **354** of combustor cans **390**, and only the second fuel lines **386** supplying combustor cans **392**; however, embodiments of each of the combustor cans **350** may include first fuel lines **384** and second fuel lines **386** configured to supply fuel to each of the fuel nozzles **164**, as shown for the first combustor can **388**.

[0090] The controller **118** may control (e.g., increase, decrease) the fuel flow along a fuel line (e.g., first fuel line **384**

and/or second fuel line 386) via controlling (e.g., incrementally opening or closing) a trim valve 382 arranged along the fuel line. That is, each trim valve 382 may be fluidly coupled between the respective nozzle set manifold and the set of fuel nozzles. In some embodiments, trim valves 382 may be disposed along the first fuel lines 384 to control (e.g., increase, decrease) the fuel flow to the first set 354 of fuel nozzles of each combustor can 350. Additionally, or in the alternative, trim valves 382 may be disposed along the second fuel lines 386 to control (e.g., increase, decrease) the fuel flow to the second set 358 of fuel nozzles of each combustor can 350. As the second fuel lines 386 may supply fuel to a greater number of fuel nozzles (e.g., outer fuel nozzles) per combustor can 350 than the first fuel lines 384, control of trim valves 382 along the second fuel lines 386 may enable a coarse or bulk level of control of the FOR for the combustor can 350. Control (e.g., incrementally opening or closing) of trim valves 382 along the first fuel lines 384 may enable a fine level of control of the FOR for the combustor can 350.

[0091] The controller 118 may differentially control the trim valves 382 to supply different fuel flows to the sets of nozzles of different combustor cans 350. For example, the controller 118 may increase the fuel flow to the first set 354 and/or the second set 358 of fuel nozzles 164 of the first combustor can 388, and decrease the fuel flow to the first set 354 and/or the second set 358 of fuel nozzles 164 of a second combustor can 394. The controller 118 may differentially control the trim valves 382 to a set of fuel nozzles based at least in part on the position of the respective combustor can 350 within the SEGR gas turbine system 52, a fuel composition, a composition of combustion gases from the respective combustor can 350, the geometry of the combustor can 350 and/or the fuel lines, or any combination thereof. Differential control of the trim valves 382 along the first fuel lines 384 and/or the second fuel lines 386 enables the controller 118 to control the fuel flow that mixes with the oxidant flow in each combustor can 350, thereby controlling the FOR for each combustor can 350. In some embodiments, sensors 396 along the fuel lines (e.g., first fuel lines 384, second fuel lines 386) may provide sensor feedback to the controller 118 that is related to the fuel flow through the respective fuel lines. For example, the sensors 396 may measure the pressure, temperature, flow rate, and/or composition of the fuel flow along the respective fuel lines.

[0092] FIG. 8A illustrates a computer-implemented method 420 for controlling the FOR, and FIG. 8B illustrates a computer-implemented method 440 for controlling the EGOR, where the FOR control method 420 is independent of the EGOR control method 440. As discussed above, the oxidant control system 262 and the fuel control system 264 may each have an industrial controller with a processor, a memory, and instructions stored on the memory and executable by the processor to control (e.g., increase, decrease) the oxidant flow 260 and/or the fuel flow 266 to the one or more fuel nozzles 164 of each combustor 160. The controller 118 may implement methods 420 and 440 via control of the oxidant control system 262 and/or the fuel control system 264. In the FOR control method 420, the controller 118 determines (block 422) a target fuel flow rate based at least in part on several inputs, such as a fuel flow rate to one or more fuel nozzles 164 of the respective combustor 160, the fuel composition, an oxidant flow rate to the one or more fuel nozzles of the respective combustor, and a desired equivalence ratio for the respective combustor 160 and/or the SEGR gas turbine sys-

tem 52. The controller 118 measures (block 424) the fuel flow rate and measures (block 426) the fuel composition to determine the current flow rate and composition of the fuel flow. In some embodiments, the fuel flow rate may be based at least in part on feedback from one or more flow meters in the flow sensing system 272, from the sensors 396 of the fuel control system 264, from the position of the trim valves 382, or from the position of the distribution control valves 381, or any combination thereof. The controller 118 may measure (block 428) the oxidant flow via the oxidant sensing system 270 to determine the oxidant flow rate. The controller 118 receives (block 430) the desired equivalence ratio via an operator interface or a set of instructions (e.g., code, software) configured to operate the SEGR gas turbine system 52. In some embodiments, the desired equivalence ratio may be based at least in part on an operating condition of the gas turbine system. For example, the desired equivalence ratio of the gas turbine system during a start-up sequence or no-load condition may be less than the desired equivalence ratio of the gas turbine system while operating at a steady state condition driving a load.

[0093] The controller 118 determines (block 422) the target fuel flow rate for each of the one or more combustors based at least in part on the current measured fuel flow rate, the fuel composition, the oxidant flow rate, and the desired equivalence ratio for the respective combustor. The controller 118 regulates (block 432) the fuel flow to the one or more fuel nozzles of a combustor to the target fuel flow rate to adjust the FOR so that the current equivalence ratio of the combustors is approximately equal to the desired equivalence ratio (e.g., between approximately 0.90 to 1.10, between approximately 0.95 to 1.05) for the combustor. That is, the controller 118 may adjust the FOR the one or more fuel nozzles of a combustor so that the current equivalence ratio is within a deviation of approximately 1, 2, 3, 4, 5, 6, 7, 8, 9, or 10 percent of the desired equivalence ratio. The controller 118 may control the fuel flow rate to the target fuel flow rate via control of the distribution control valves 381 and/or the trim valves 382. In some embodiments, the controller 118 determines (block 422) the target fuel flow rate for multiple combustors based at least in part on a bulk fuel flow rate, bulk fuel composition, bulk oxidant flow rate, and bulk desired equivalence ratio for the multiple combustors. The controller 118 regulates (block 432) the fuel flow to the fuel nozzles of the multiple combustors to the target fuel flow rate to adjust the FOR so that the current bulk equivalence ratio for the multiple combustors is approximately equal to the bulk desired equivalence ratio (e.g., between approximately 0.90 to 1.10, between approximately 0.95 to 1.05) for the multiple combustors. That is, the controller 118 may adjust the FOR for the fuel nozzles of the multiple combustors so that the current bulk equivalence ratio is within a deviation of approximately 1, 2, 3, 4, 5, 6, 7, 8, 9, or 10 percent of the bulk desired equivalence ratio for the multiple combustors.

[0094] FIG. 8B illustrates a computer-implemented method 440 for controlling the EGOR independent of the FOR control method 420. The controller 118 determines (block 442) a target bypass flow rate from the extraction manifold based at least in part on several inputs, such as an oxidant flow rate to one or more fuel nozzles 164 of the respective combustor 160, the flow rate of the extracted first portion 288, and a desired EGOR for the respective combustor 160 and/or the SEGR gas turbine system 52. The controller 118 may measure (block 428) the oxidant flow via the

oxidant sensing system 270 to determine the oxidant flow rate. The controller 118 determines (block 444) the extraction flow rate of the first portion 288. For example, the controller 118 may determine the extraction flow rate based at least in part on the bypass flow sensor 293, the extraction flow sensor 294, or the exhaust gas supply system 78, or any combination thereof. The controller 118 receives (block 446) the desired EGOR via an operator interface or a set of instructions (e.g., code, software) configured to operate the SEGR gas turbine system 52. In some embodiments, the desired EGOR is based at least in part on the equivalence ratio, such that the operating point (e.g., relationship between the combustor operability and the equivalence ratio) is below the combustor operability limit, as discussed above with FIG. 6. Upon determination of the target bypass flow rate, the controller 118 controls the bypass flow rate to the determined target bypass flow rate via control of the turbine bypass valve 290 and/or the inlet guide vanes 284, thereby enabling the controller 118 to regulate (block 448) the EGOR for the one or more combustors.

[0095] The desired equivalence ratio and the desired EGOR may be determined such that the operating point is below the combustor operability limit. The desired FOR may be based at least in part on the current EGOR, and the desired EGOR may be based at least in part on the current FOR. However, while the desired values for the FOR and the EGOR may be based at least in part on each other, the controls to regulate the FOR via adjustment of the fuel flow rates and to regulate the EGOR via adjustment of the turbine bypass flow are generally independent of each other. For example, regulation of the fuel flow to the first and/or the second nozzle set of a combustor can to control the FOR does not affect the EGOR for the combustor can. Likewise, regulation of the bypass flow to control the EGOR for the one or more combustors does not affect the FOR. The oxidant flow may be controlled based at least in part on the loading on the gas turbine system, and the controller adjusts the FOR and the EGOR accordingly to maintain the operating point for the one or more combustors below the combustor operability limit.

[0096] This written description uses examples to disclose the invention, including the best mode, and also to enable any person skilled in the art to practice the invention, including making and using any devices or systems and performing any incorporated methods. The patentable scope of the invention is defined by the claims, and may include other examples that occur to those skilled in the art. Such other examples are intended to be within the scope of the claims if they have structural elements that do not differ from the literal language of the claims, or if they include equivalent structural elements with insubstantial differences from the literal language of the claims.

ADDITIONAL EMBODIMENTS

[0097] The present embodiments provide a system and method for controlling combustion and emissions in a gas turbine engine with exhaust recirculation. It should be noted that any one or a combination of the features described above may be utilized in any suitable combination. Indeed, all permutations of such combinations are presently contemplated. By way of example, the following clauses are offered as further description of the present disclosure:

Embodiment 1

[0098] A system comprises a fuel control system configured to control a fuel flow to one or more combustors; an

oxidant control system configured to control an oxidant flow to each combustor of the one or more combustors, wherein the oxidant flow is configured to at least partially react with the fuel flow within the one or more combustors to form an exhaust gas flow; an exhaust gas system configured to direct a recirculation flow of the exhaust gas flow to each combustor of the one or more combustors; and a controller coupled to the fuel control system, the oxidant control system, and the exhaust gas system, wherein the controller is configured to independently control a fuel-to-oxidant ratio (FOR) and an exhaust gas-to-oxidant ratio (EGOR), the FOR comprises the fuel flow divided by the oxidant flow, and the EGOR comprises the recirculation flow divided by the oxidant flow.

Embodiment 2

[0099] The system of embodiment 1, wherein each combustor of the one or more combustors comprises a plurality of fuel nozzles, and the fuel control system comprises a first fuel trim valve coupled to a first set of fuel nozzles of the plurality of fuel nozzles, wherein the fuel control system is configured to adjust a first fuel flow to the first set of fuel nozzles via the first fuel trim valve to control the FOR.

Embodiment 3

[0100] The system of embodiment 2, wherein the fuel control system comprises a second fuel trim valve coupled to a second set of fuel nozzles of the plurality of fuel nozzles, wherein the fuel control system is configured to adjust a second fuel flow to the second set of fuel nozzles via the second fuel trim valve to control the FOR.

Embodiment 4

[0101] The system of embodiment 3, comprising a first fuel supply coupled to the first fuel trim valve, and a second fuel supply coupled to the second fuel trim valve.

Embodiment 5

[0102] The system of embodiments 2, 3, or 4, wherein the first set of fuel nozzles comprises one or more inner fuel nozzles, and a second set of fuel nozzles comprises a plurality of outer fuel nozzles disposed about the one or more inner fuel nozzles.

Embodiment 6

[0103] The system of any preceding embodiment, wherein the controller is configured to differentially control the FOR to each combustor of the one or more combustors based at least in part on an arrangement of the one or more combustors within a gas turbine engine.

Embodiment 7

[0104] The system of any preceding embodiment, comprising a turbine configured to receive the exhaust gas flow from the one or more combustors; and an exhaust extraction system configured to extract a first portion of the recirculation flow from the one or more combustors, wherein the exhaust extraction system comprises a turbine bypass valve configured to adjust a bypass flow of the first portion to bypass the turbine and to join the recirculation flow, wherein the turbine bypass valve is coupled to the controller, the controller is configured to control the EGOR based at least in part on control of the

turbine bypass valve, and the recirculation flow comprises the bypass flow and a second portion of the exhaust gas flow through the turbine.

Embodiment 8

[0105] The system of embodiment 7, wherein the exhaust extraction system is configured to direct a third portion of the recirculation flow from the one or more combustors to an exhaust gas supply system for transmission to a hydrocarbon production system, a pipeline, a storage tank, or a carbon sequestration system, or any combination thereof.

Embodiment 9

[0106] The system of any preceding embodiment, wherein the recirculation flow comprises less than approximately 100 parts per million by volume of oxygen.

Embodiment 10

[0107] The system of any preceding embodiment, wherein the controller is configured to control the oxidant flow to the one or more combustors based at least in part on a load on the system.

Embodiment 11

[0108] The system of any preceding embodiment, wherein the controller is configured to control the fuel flow to the one or more combustors based at least in part on a desired equivalence ratio of the one or more combustors.

Embodiment 12

[0109] The system of any preceding embodiment, wherein the controller is configured to control the recirculation flow based at least in part on a relationship between the FOR, the EGOR, and a combustor operability limit corresponding to the FOR and the EGOR.

Embodiment 13

[0110] The system of any preceding embodiment, comprising a gas turbine engine having the one or more combustors, a turbine driven by the exhaust gas flow from the one or more combustors, and an exhaust gas compressor driven by the turbine, wherein the exhaust gas compressor is configured to compress and to route the exhaust gas flow to the one or more combustors, and the gas turbine engine is a stoichiometric exhaust gas recirculation (SEGR) gas turbine engine.

Embodiment 14

[0111] A system comprising a combustor, wherein the combustor includes an oxidant inlet configured to receive an oxidant flow; a plurality of fuel nozzles, wherein each fuel nozzle of the plurality of fuel nozzles is configured to receive a fuel flow; a first fuel trim valve configured to control the fuel flow to at least one fuel nozzle of the plurality of fuel nozzles, wherein the first fuel trim valve is controlled based at least in part on a fuel-to-oxidant ratio (FOR); a recirculation inlet configured to receive a recirculation flow from a recirculation system, wherein at least the oxidant flow and the fuel flow are configured to combust within the combustor and form an exhaust gas flow; and one or more extraction ports configured to extract a first portion of the recirculation flow to an extraction manifold. The system also includes a turbine configured

to receive the exhaust gas flow and a second portion of the recirculation flow from the combustor and the exhaust gas flow, to drive a load, and to direct the second portion of the exhaust gas flow to the recirculation system; and a turbine bypass valve configured to extract a bypass flow from the first portion of the recirculation flow to the extraction manifold, wherein the turbine bypass valve is configured to direct the bypass flow to the recirculation system based at least in part on an exhaust-gas-to-oxidant ratio (EGOR), wherein the recirculation flow comprises the second portion of the exhaust gas flow and the bypass flow, and the turbine bypass valve is controlled independently of the first fuel trim valve.

Embodiment 15

[0112] The system of embodiment 14, wherein the plurality of fuel nozzles comprises a first set of one or more inner fuel nozzles and a second set of one or more outer fuel nozzles disposed around the one or more inner fuel nozzles of the plurality of fuel nozzles the first fuel trim valve is configured to control the fuel flow to one or more inner fuel nozzles, and the combustor comprises a second fuel trim valve configured to control the fuel flow to the second set of one or more outer fuel nozzles of the plurality of fuel nozzles.

Embodiment 16

[0113] The system of embodiment 15, wherein the first fuel trim valve is fluidly coupled to a first fuel supply, and the second fuel trim valve is fluidly coupled to a second fuel supply different from the first fuel supply.

Embodiment 17

[0114] The system of embodiment 14, 15, or 16, comprising the extraction manifold coupled between the combustor and the turbine bypass valve, wherein the extraction manifold is configured to direct a third portion of the first portion of the recirculation flow to a hydrocarbon production system, a pipeline, a storage tank, or a carbon sequestration system, or any combination thereof.

Embodiment 18

[0115] The system of embodiment 14, 15, 16, or 17, comprising the recirculation system, wherein the recirculation system comprises an exhaust gas compressor having a plurality of inlet guide vanes, and the inlet guide vanes are controlled based at least in part on the EGOR.

Embodiment 19

[0116] A method of operating an exhaust gas recirculation (EGR) gas turbine engine, comprising: controlling a fuel-to-oxidant ratio (FOR) to a combustor based at least in part on a desired equivalence ratio and a load on the EGR gas turbine engine; combusting fuel and oxidant in the combustor to form exhaust gas; recirculating a recirculation portion of the exhaust gas to the combustor; and controlling an exhaust-gas-to-oxidant ratio (EGOR) based at least in part on an operability limit corresponding to the FOR.

Embodiment 20

[0117] The method of embodiment 19, wherein controlling the FOR comprises adjusting a fuel flow to one or more fuel nozzles of a plurality of fuel nozzles.

Embodiment 21

[0118] The method of embodiment 20, wherein controlling the FOR comprises differentially adjusting a fuel flow to one or more inner fuel nozzles of the plurality of fuel nozzles relative to a plurality of outer fuel nozzles of the plurality of fuel nozzles, wherein the outer fuel nozzles are disposed about the inner fuel nozzles.

Embodiment 22

[0119] The method of embodiment 19, 20, or 21, comprising: extracting a first portion of the recirculation portion of the exhaust gas from the combustor, wherein controlling the EGOR comprises adjusting a bypass valve to control a bypass portion of the first portion that bypasses a turbine of the EGR gas turbine engine; and routing a second portion of the recirculation portion of the exhaust gas from the combustor to the turbine of the EGR gas turbine engine, wherein the recirculation portion comprises the bypass portion and the second portion.

Embodiment 23

[0120] The method of embodiment 19, 20, 21, or 22, comprising controlling the EGOR by adjusting one or more inlet guide vanes to an exhaust gas compressor of the EGR gas turbine engine, wherein the exhaust gas compressor is configured to direct the recirculation portion of the exhaust gas to the combustor.

Embodiment 24

[0121] The method of embodiment 19, 20, 21, 22, or 23, comprising independently controlling the FOR or the EGOR to be less than an operability limit during a startup sequence for the EGR gas turbine engine.

Embodiment 25

[0122] The method of embodiment 19, 20, 21, 22, 23, or 24, comprising controlling the FOR ratio to a corresponding equivalence ratio between approximately 0.95 to 1.05 during a steady state operation.

1. A system comprising:

- a fuel control system configured to control a fuel flow to one or more combustors;
- an oxidant control system configured to control an oxidant flow to each combustor of the one or more combustors, wherein the oxidant flow is configured to at least partially react with the fuel flow within the one or more combustors to form an exhaust gas flow;
- an exhaust gas system configured to direct a recirculation flow of the exhaust gas flow to each combustor of the one or more combustors; and
- a controller coupled to the fuel control system, the oxidant control system, and the exhaust gas system, wherein the controller is configured to independently control a fuel-to-oxidant ratio (FOR) and an exhaust gas-to-oxidant ratio (EGOR), the FOR comprises the fuel flow divided by the oxidant flow, and the EGOR comprises the recirculation flow divided by the oxidant flow.

2. The system of claim 1, wherein each combustor of the one or more combustors comprises a plurality of fuel nozzles, and the fuel control system comprises a first fuel trim valve coupled to a first set of fuel nozzles of the plurality of fuel nozzles, wherein the fuel control system is configured to

adjust a first fuel flow to the first set of fuel nozzles via the first fuel trim valve to control the FOR.

3. The system of claim 2, wherein the fuel control system comprises a second fuel trim valve coupled to a second set of fuel nozzles of the plurality of fuel nozzles, wherein the fuel control system is configured to adjust a second fuel flow to the second set of fuel nozzles via the second fuel trim valve to control the FOR.

4. The system of claim 3, comprising a first fuel supply coupled to the first fuel trim valve, and a second fuel supply coupled to the second fuel trim valve.

5. The system of claim 2, wherein the first set of fuel nozzles comprises one or more inner fuel nozzles, and a second set of fuel nozzles comprises a plurality of outer fuel nozzles disposed about the one or more inner fuel nozzles.

6. The system of claim 1, wherein the controller is configured to differentially control the FOR to each combustor of the one or more combustors based at least in part on an arrangement of the one or more combustors within a gas turbine engine.

7. The system of claim 1, comprising:

a turbine configured to receive the exhaust gas flow from the one or more combustors; and

an exhaust extraction system configured to extract a first portion of the recirculation flow from the one or more combustors, wherein the exhaust extraction system comprises a turbine bypass valve configured to adjust a bypass flow of the first portion to bypass the turbine and to join the recirculation flow, wherein the turbine bypass valve is coupled to the controller, the controller is configured to control the EGOR based at least in part on control of the turbine bypass valve, and the recirculation flow comprises the bypass flow and a second portion of the exhaust gas flow through the turbine.

8. The system of claim 7, wherein the exhaust extraction system is configured to direct a third portion of the recirculation flow from the one or more combustors to an exhaust gas supply system for transmission to a hydrocarbon production system, a pipeline, a storage tank, or a carbon sequestration system, or any combination thereof.

9. The system of claim 1, wherein the recirculation flow comprises less than approximately 100 parts per million by volume of oxygen.

10. The system of claim 1, wherein the controller is configured to control the oxidant flow to the one or more combustors based at least in part on a load on the system.

11. The system of claim 1, wherein the controller is configured to control the fuel flow to the one or more combustors based at least in part on a desired equivalence ratio of the one or more combustors.

12. The system of claim 1, wherein the controller is configured to control the recirculation flow based at least in part on a relationship between the FOR, the EGOR, and a combustor operability limit corresponding to the FOR and the EGOR.

13. The system of claim 1, comprising a gas turbine engine having the one or more combustors, a turbine driven by the exhaust gas flow from the one or more combustors, and an exhaust gas compressor driven by the turbine, wherein the exhaust gas compressor is configured to compress and to route the exhaust gas flow to the one or more combustors, and the gas turbine engine is a stoichiometric exhaust gas recirculation (SEGR) gas turbine engine.

- 14.** A system comprising:
a combustor, comprising:
an oxidant inlet configured to receive an oxidant flow;
a plurality of fuel nozzles, wherein each fuel nozzle of the plurality of fuel nozzles is configured to receive a fuel flow;
a first fuel trim valve configured to control the fuel flow to at least one fuel nozzle of the plurality of fuel nozzles, wherein the first fuel trim valve is controlled based at least in part on a fuel-to-oxidant ratio (FOR);
a recirculation inlet configured to receive a recirculation flow from a recirculation system, wherein at least the oxidant flow and the fuel flow are configured to combust within the combustor and form an exhaust gas flow; and
one or more extraction ports configured to extract a first portion of the recirculation flow to an extraction manifold;
a turbine configured to receive the exhaust gas flow and a second portion of the recirculation flow from the combustor and the exhaust gas flow, to drive a load, and to direct the second portion of the exhaust gas flow to the recirculation system; and
a turbine bypass valve configured to extract a bypass flow from the first portion of the recirculation flow to the extraction manifold, wherein the turbine bypass valve is configured to direct the bypass flow to the recirculation system based at least in part on an exhaust-gas-to-oxidant ratio (EGOR), wherein the recirculation flow comprises the second portion of the exhaust gas flow and the bypass flow, and the turbine bypass valve is controlled independently of the first fuel trim valve.
- 15.** The system of claim **14**, wherein the plurality of fuel nozzles comprises a first set of one or more inner fuel nozzles and a second set of one or more outer fuel nozzles disposed around the one or more inner fuel nozzles of the plurality of fuel nozzles the first fuel trim valve is configured to control the fuel flow to one or more inner fuel nozzles, and the combustor comprises a second fuel trim valve configured to control the fuel flow to the second set of one or more outer fuel nozzles of the plurality of fuel nozzles.
- 16.** The system of claim **15**, wherein the first fuel trim valve is fluidly coupled to a first fuel supply, and the second fuel trim valve is fluidly coupled to a second fuel supply different from the first fuel supply.
- 17.** The system of claim **14**, comprising the extraction manifold coupled between the combustor and the turbine bypass valve, wherein the extraction manifold is configured to direct a third portion of the first portion of the recirculation flow to a hydrocarbon production system, a pipeline, a storage tank, or a carbon sequestration system, or any combination thereof.
- 18.** The system of claim **14**, comprising the recirculation system, wherein the recirculation system comprises an exhaust gas compressor having a plurality of inlet guide vanes, and the inlet guide vanes are controlled based at least in part on the EGOR.
- 19.** A method of operating an exhaust gas recirculation (EGR) gas turbine engine, comprising:
controlling a fuel-to-oxidant ratio (FOR) to a combustor based at least in part on a desired equivalence ratio and a load on the EGR gas turbine engine;
combusting fuel and oxidant in the combustor to form exhaust gas;
recirculating a recirculation portion of the exhaust gas to the combustor; and
controlling an exhaust-gas-to-oxidant ratio (EGOR) based at least in part on an operability limit corresponding to the FOR.
- 20.** The method of claim **19**, wherein controlling the FOR comprises adjusting a fuel flow to one or more fuel nozzles of a plurality of fuel nozzles.
- 21.** The method of claim **20**, wherein controlling the FOR comprises differentially adjusting a fuel flow to one or more inner fuel nozzles of the plurality of fuel nozzles relative to a plurality of outer fuel nozzles of the plurality of fuel nozzles, wherein the outer fuel nozzles are disposed about the inner fuel nozzles.
- 22.** The method of claim **19**, comprising:
extracting a first portion of the recirculation portion of the exhaust gas from the combustor, wherein controlling the EGOR comprises adjusting a bypass valve to control a bypass portion of the first portion that bypasses a turbine of the EGR gas turbine engine; and
routing a second portion of the recirculation portion of the exhaust gas from the combustor to the turbine of the EGR gas turbine engine, wherein the recirculation portion comprises the bypass portion and the second portion.
- 23.** The method of claim **19**, comprising controlling the EGOR by adjusting one or more inlet guide vanes to an exhaust gas compressor of the EGR gas turbine engine, wherein the exhaust gas compressor is configured to direct the recirculation portion of the exhaust gas to the combustor.
- 24.** The method of claim **19**, comprising independently controlling the FOR or the EGOR to be less than an operability limit during a startup sequence for the EGR gas turbine engine.
- 25.** The method of claim **19**, comprising controlling the FOR ratio to a corresponding equivalence ratio between approximately 0.95 to 1.05 during a steady state operation.

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