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(54) **AEROSPACE INTELLIGENT WINDOW SYSTEM**

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(*) Notice: Subject to any disclaimer, the term of this patent is extended or adjusted under 35 U.S.C. 154(b) by 103 days.

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(57) **ABSTRACT**

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(65) **Prior Publication Data**

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A network system for monitoring and storing data of aircraft intelligent windows or windshields to provide useable life of, provide real life performance of, and/or measure characteristics and/or properties of, the windshields forwards the data from sensors mounting the windshield to a window sensing hub having a microprocessor programed to receive and process the data to determine the performance of the windshield and formatting the data in accordance to a preset program, wherein the program includes providing data from the sensors that measures characteristics and properties of the windshield that are active during the period in which the data is taken. An aircraft central maintenance system connected to the window sensing hub receives the formatted information from the window sensing hub and unfiltered or unformatted information, wherein the unfiltered information from the windshield is acted on by the central maintenance system to provide an estimated useable life of the windshield.

Related U.S. Application Data

(60) Provisional application No. 61/678,315, filed on Aug. 1, 2012.

(51) **Int. Cl.**

B64C 1/14 (2006.01)
B64D 45/00 (2006.01)

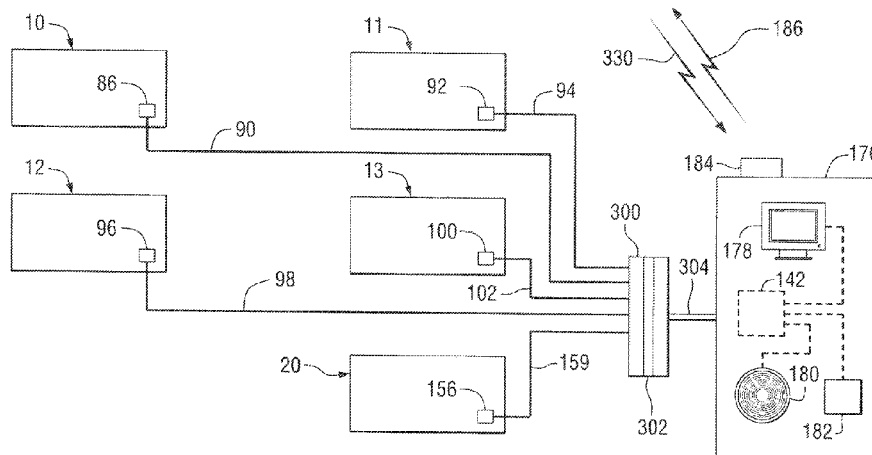
(52) **U.S. Cl.**

CPC **B64C 1/1484** (2013.01); **B64C 1/1476** (2013.01); **B64C 1/1492** (2013.01); **B64D 2045/0085** (2013.01)

(58) **Field of Classification Search**

None
See application file for complete search history.

6 Claims, 6 Drawing Sheets



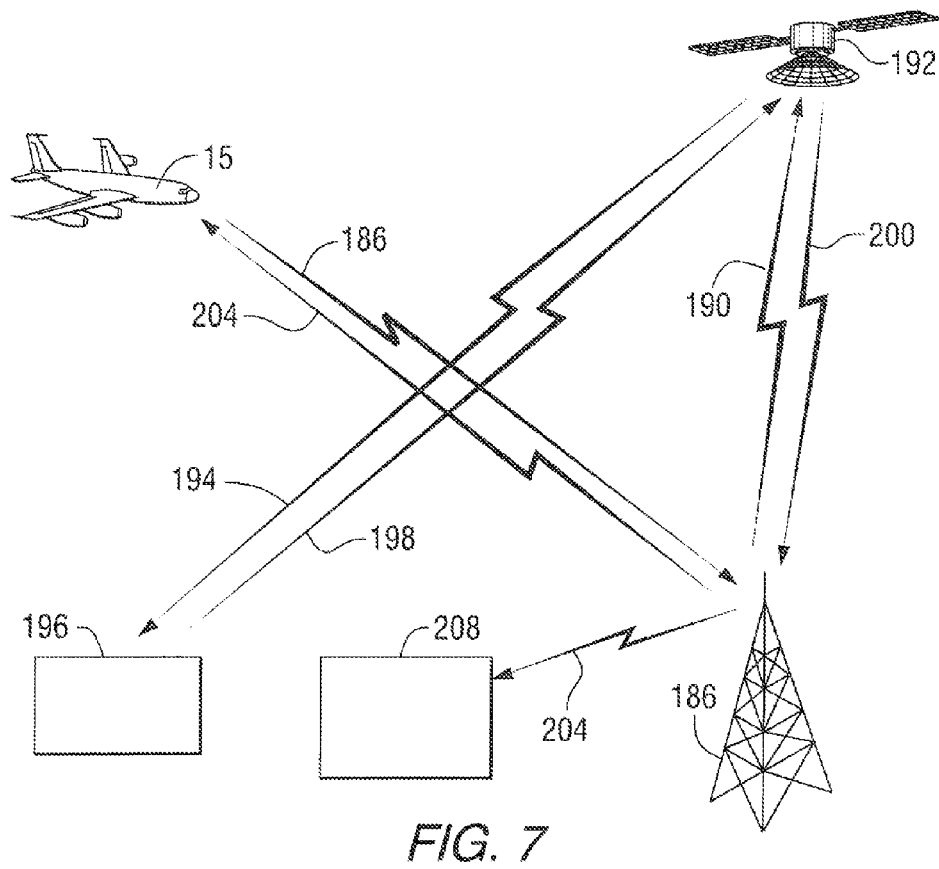
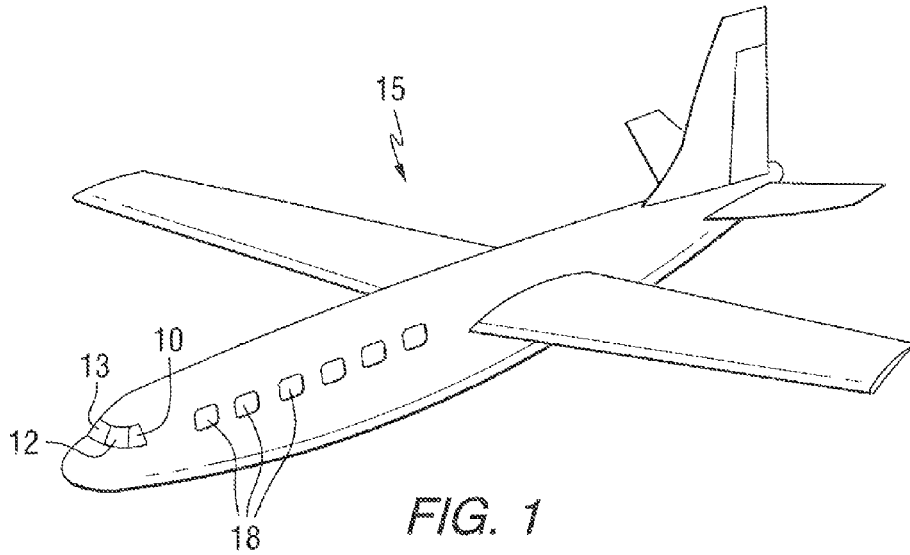
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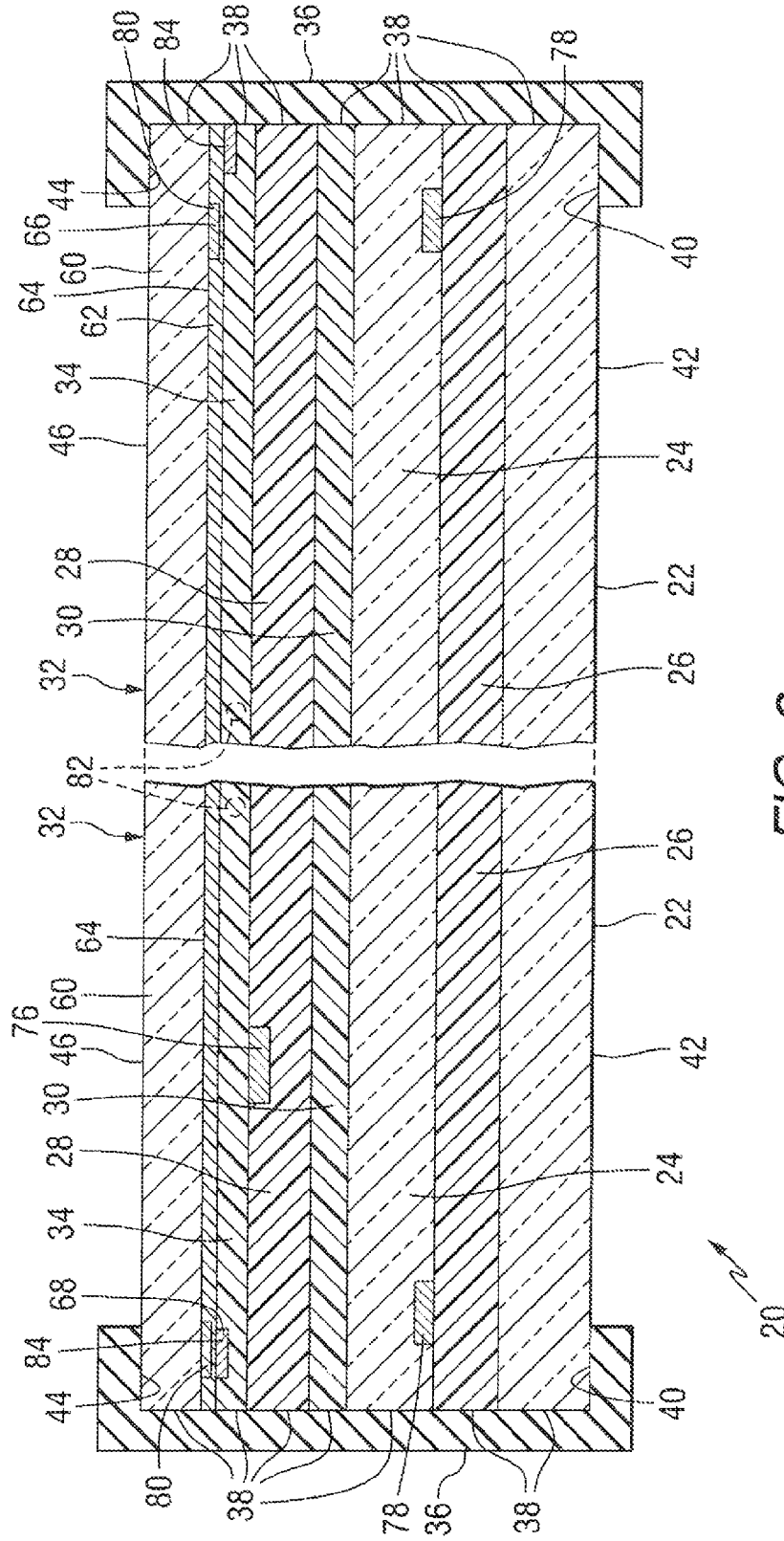


FIG. 2

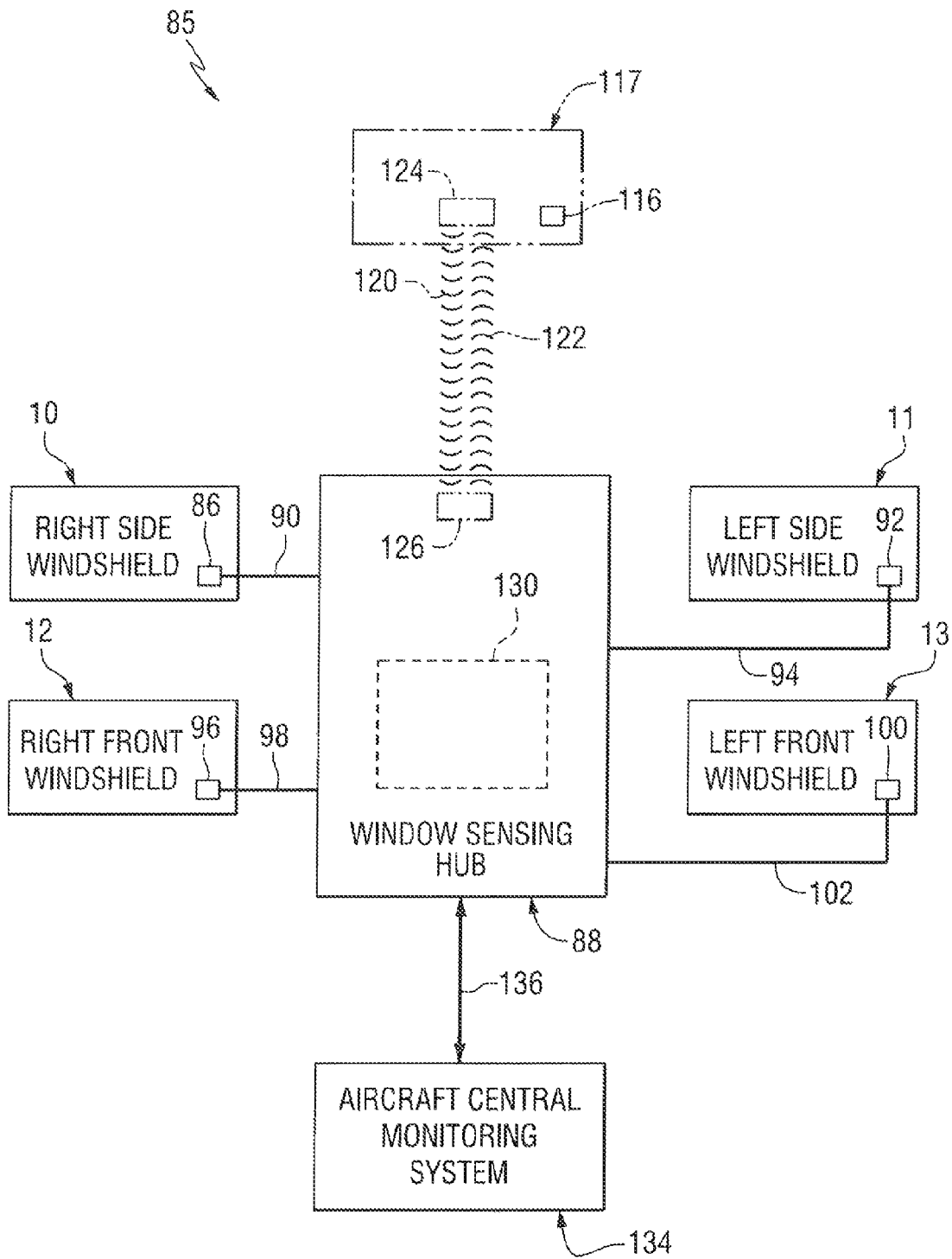


FIG. 3

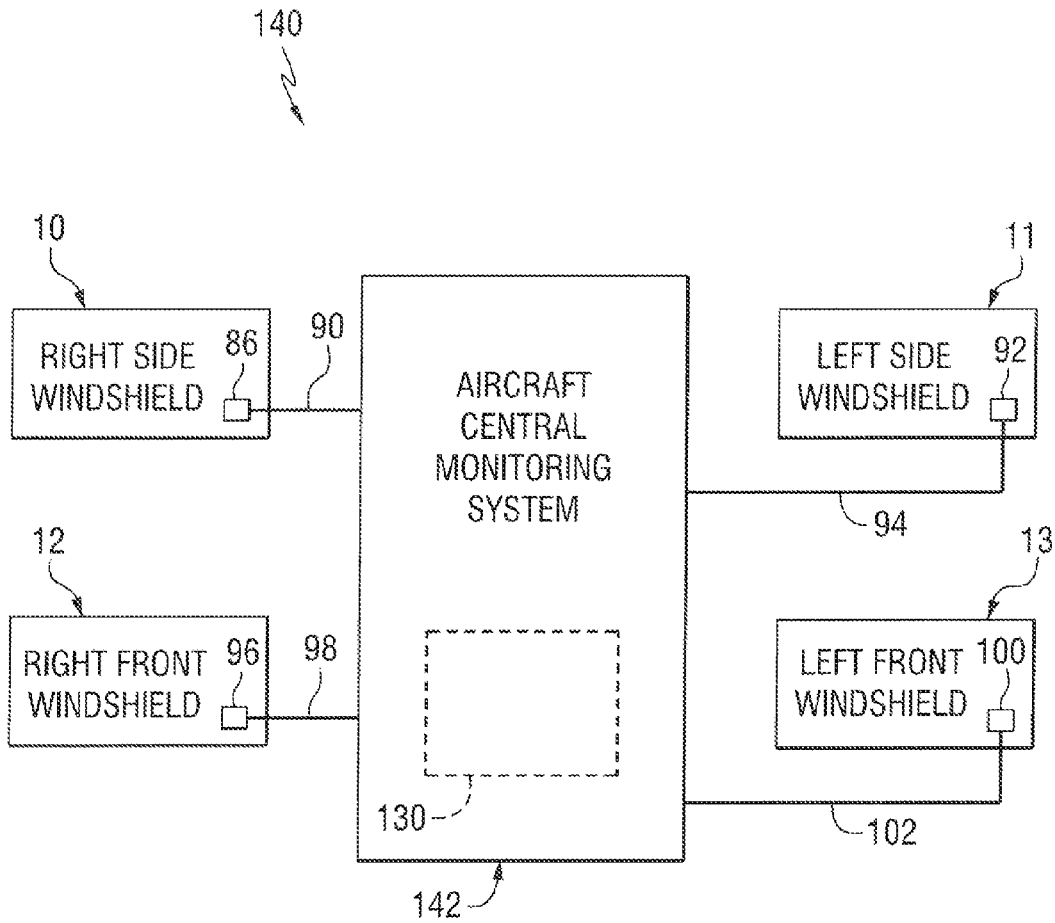


FIG. 4

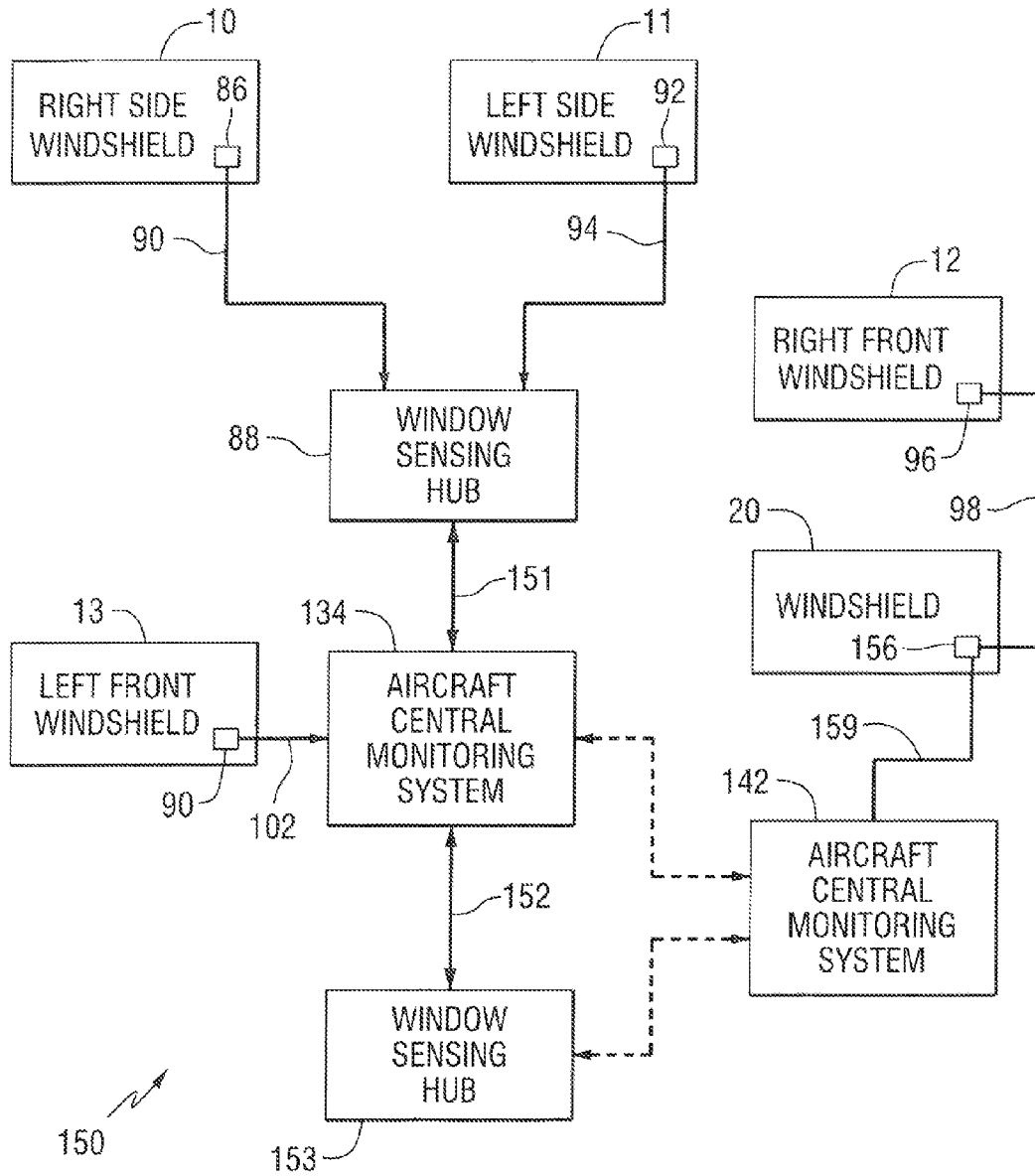


FIG. 5

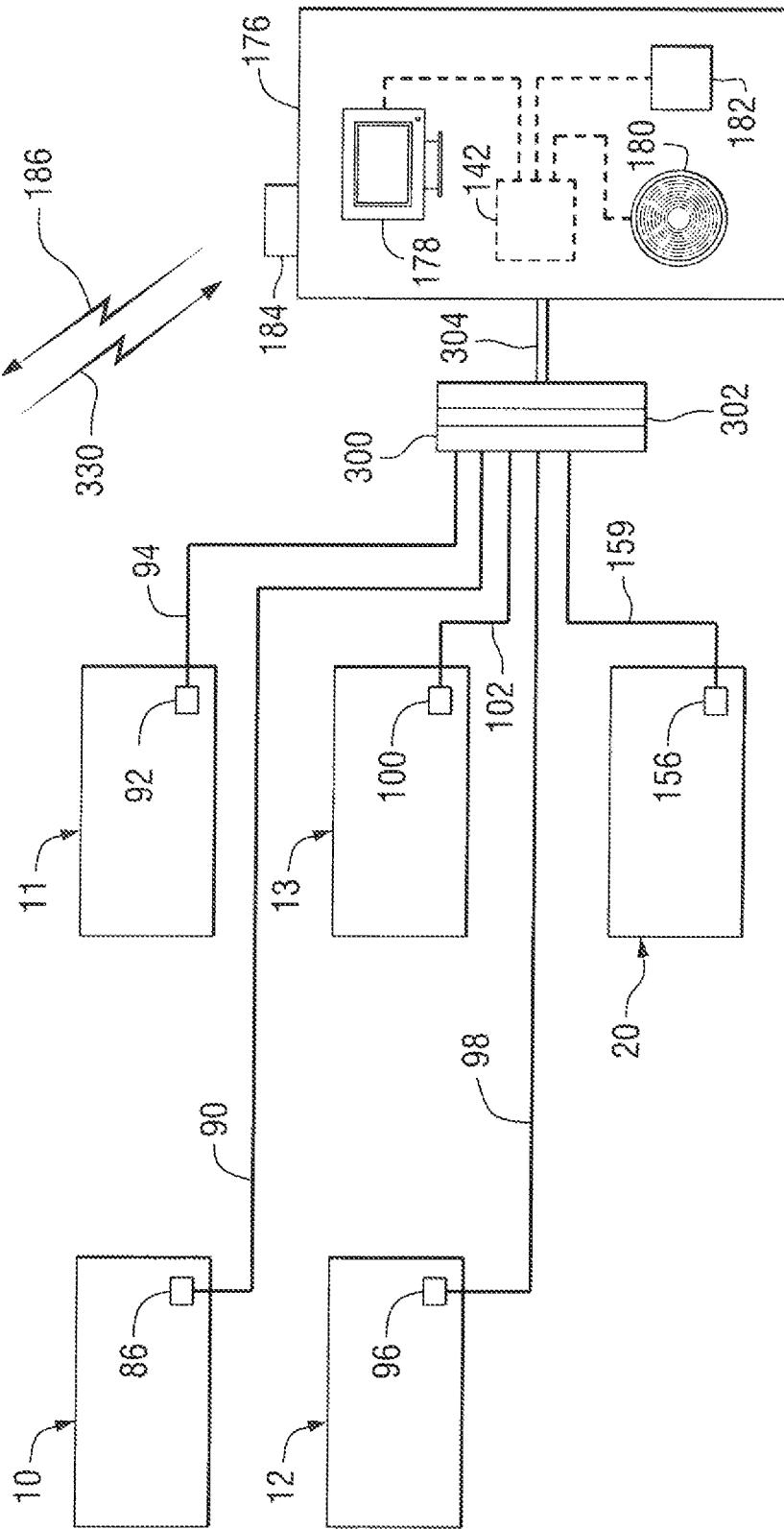


FIG. 6

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**AEROSPACE INTELLIGENT WINDOW
SYSTEM**

RELATED APPLICATION

This application claims the benefit of the filing date of U.S. Provisional Application Ser. No. 61/678,315 filed on Aug. 1, 2012 in the names of Yu Jiao, Nicolas Duarte and Monroe A. Stone for an "Aerospace Intelligent Window System". Application Ser. No. 61/678,315 in its entirety is hereby incorporated by reference.

BACKGROUND OF THE INVENTION

1. Field of the Invention

This invention relates to an aerospace intelligent window system that includes performance data of one or more currently mounted windows in an aircraft (present performance data), and performance data of one or more previously mounted windows in the same aircraft and/or other aircrafts (past performance data) wherein the present and/or past performance data is used to determine life expectance of the one or more currently mounted windows.

2. Presently Available Technology

The present technology relating to sensors for aircraft windows, e.g. an aircraft windshield provides for mounting one or more sensors on an aircraft window and connecting the sensor output to a monitoring system mounted on the aircraft to provide instant performance data of selected properties or characteristics of the window. For a detailed discussion of sensors monitoring performance of selected properties and/or characteristics of aircraft windows reference is directed to U.S. Patent Application Publication Nos. 2010/0163675 A1, and 2013/075,531A1. The sensors, e.g. an impact sensor, a rupture sensor, an arc sensor, a temperature sensor and/or a moisture sensor, mounted on the aircraft window provide information relating to the performance of the window to determine if the performance is operating within acceptable limits. When the window performance is outside of acceptable limits, the window is repaired or replaced, e.g. as disclosed in U.S. Pat. No. 8,155,816 B2. U.S. Pat. No. 8,155,816 B2, and U.S. Patent Application Publication Nos. 2010/0183675 A1 and 2013/075,531A1 in their entirety are hereby incorporated by reference.

Although the present system to monitor performance of an aircraft window is acceptable, there are limitations. More particularly, one limitation of the presently available systems is that the data from the sensors provide information regarding actual performance of the aircraft window, e.g. but not limited to an aircraft windshield but little, if any, information regarding the useable life or life expectancy of the aircraft window. Another limitation of the present system is that the data is presented as information for each aircraft window as a separate unit, and the interaction between aircraft windows is not fully taken into account. As can now be appreciated by those skilled in the art, it would be advantageous to consider interconnect output of the sensors of aircraft windows mounted in the body of an airplane to form a network of windows. In this manner the performance of the network of windows can be monitored, and the present and past performance of the network of windows of the aircraft can be considered to determine useable life or life expectance of the windows of the network of windows and determine if the window failure is a result of the window construction or an indication that a portion of the aircraft

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body and/or the window mount surrounding the aircraft window is causing the window to perform outside of an acceptable range.

SUMMARY OF THE INVENTION

This invention relates to a network system for monitoring and storing performance data of a transparency to provide estimated useable life of the transparency and/or to provide real life performance of the transparency. The transparency includes but is not limited to a transparent sheet having a sensor group secured on a surface of the sheet to measure predetermined characteristics and/or properties of the transparency, wherein the sensor group includes, but is not limited to at least one sensor selected from the family of sensors including, but not limited to an arc sensor for measuring arcing of an electrically heatable member mounted on a surface of the transparent sheet, hereinafter referred to as an "arc sensor"; a heat sensor for measuring temperature of the heatable member, hereinafter referred to as a "heat sensor"; a moisture sensor for measuring moisture content on a surface of the transparent sheet, hereinafter referred to as a "moisture sensor"; an impact sensor for measuring force of impact of objects hitting a surface of the transparent sheet, hereinafter referred to as an "impact sensor", and a sensor for identifying fractures in and/or on the transparent sheet, hereinafter referred to as a "crack sensor".

The network system includes, but is not limited to a window sensing hub comprising a microprocessor to receive output of the sensor group of each of the transparencies, wherein the microprocessor is programed to receive data from the sensor group of the transparency providing information on the performance of the property and characteristic of the transparency associated with the sensor group and formatting the received data in accordance to a preset program, wherein the program includes at least providing data from the sensor group of the transparent sheet that measure characteristics and properties of the transparency that are active during the period in which the data taken is of interest, and a central maintenance system connected to the window sensing hub to receive the formatted information from the window sensing hub and unfiltered information, wherein the unfiltered information from the transparent sheet is acted on by the central maintenance system to provide an estimated useable life of the transparency.

This invention further relates to a transparency network inspection system, including, but not limited to a plurality of transparencies, each transparency comprising a pair of sheets laminated together and a sensor group to measure predetermined characteristics and/or properties of the transparency, wherein the sensor group comprises at least one sensor selected from the family of sensors comprising an arc sensor for measuring arcing of an electrically heatable member; a heat sensor for measuring temperature of the heatable member; a moisture sensor for measuring moisture content between the sheets of the transparency, an impact sensor for measuring force of impact of objects hitting outer surface of the transparency, and a fracture sensor for identifying fractures in a sheet of the transparency, and a central monitoring system comprising a microprocessor to receive output of the sensor group of each of the transparencies, wherein the microprocessor is programed to receive data from the sensor group of each transparency providing information on the performance of the property and characteristic of the transparency associated with the sensor group and formatting the received data in accordance to a preset

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program, wherein the program includes at least providing data from sensors that measure characteristics and properties of the transparency that are active during the period in which the data taken is of interest, and to act on the formatted information and unfiltered information to provide an estimated useable life of the transparency.

BRIEF DESCRIPTION OF THE DRAWINGS

FIG. 1 is an isometric view of an aircraft that can be used in the practice of the invention.

FIG. 2 is a cross sectional view of a non-limiting embodiment of an aircraft intelligent window used in the practice of the invention,

FIG. 3 is a non-limiting embodiment of a schematic of a network of aircraft intelligent windows of the invention.

FIG. 4 is another non-limiting embodiment of a schematic of a network of aircraft intelligent windows of the invention.

FIG. 5 is still another non-limiting embodiment of a schematic of a network of aircraft intelligent windows of the invention,

FIG. 6 is a schematic of a system of the invention for monitoring the output signals of sensors or detectors of the aircraft intelligent windows in accordance to the teachings of the invention,

FIG. 7 is a schematic of a system for scheduling repairs to, or replacements of, aircraft intelligent windows that are performing outside of acceptable limits.

DETAILED DESCRIPTION OF THE INVENTION

As used herein, spatial or directional terms, such as “left”, “right”, “inner”, “outer”, “above”, “below”, and the like, relate to the invention as it is shown in the drawing figures. However, it is to be understood that the invention can assume various alternative orientations and, accordingly, such terms are not to be considered as limiting. Further, as used herein, all numbers expressing dimensions, physical characteristics, processing parameters, quantities of ingredients, reaction conditions, and the like, used in the specification and claims are to be understood as being modified in all instances by the term “about”. Accordingly, unless indicated to the contrary, the numerical values set forth in the following specification and claims can vary depending upon the desired properties sought to be obtained by the present invention. At the very least, and not as an attempt to limit the application of the doctrine of equivalents to the scope of the claims, each numerical value should at least be construed in light of the number of reported significant digits and by applying ordinary rounding techniques. Moreover, all ranges disclosed herein are to be understood to encompass the beginning and ending range values and any and all subranges subsumed therein. For example, a stated range of “1 to 10” should be considered to include any and all subranges between (and inclusive of) the minimum value of 1 and the maximum value of 10; that is, all subranges beginning with a minimum value of 1 or more and ending with a maximum value of 10 or less, e.g., 1 to 3.3, 4.7 to 7.5, 5.5 to 10, and the like. Further, as used herein, the terms “formed over”, “applies over”, “deposited over”, or “provided over” mean formed, applied, deposited, or provided on but not necessarily in contact with the surface. For example, a coating layer “formed over” a substrate does not preclude the presence of one or more other coating layers or films of the same or different composition located between the formed coating layer and the substrate.

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Before discussing non-limiting embodiments of the invention, it is understood that the invention is not limited in its application to the details of the particular non-limiting embodiments shown and discussed herein since the invention is capable of other embodiments. Further, the terminology used herein to discuss the invention is for the purpose of description and is not of limitation. Still further, unless indicated otherwise in the following discussion, like numbers refer to like elements. The non-limited embodiments of the invention discussed herein are directed to an intelligent window system for a vehicle, e.g. but not limited to an aircraft windshield that includes, among other things, a plurality of windows; selected ones of the plurality of windows having one or more sensors to monitor the performance of properties of the window, and an aircraft central monitoring system (hereinafter also referred to as an aircraft “CMS”, or an “ACMS”) to receive data from the sensors to estimate useable life of the selected ones of the plurality of windows having a sensor.

With reference to FIG. 1, non-limiting embodiments of the invention will be directed to an aircraft laminated transparency, e.g. but not limited to a right side windshield 10, a left side windshield 11 (numbered but not shown in FIG. 1), a right front windshield 12, and a left front windshield 13, of aircraft 15. The invention, however, is not limited to any particular type of aircraft transparency, and the invention contemplates the practice of the invention on aircraft cabin windows 18, e.g. but not limited to aircraft windows of the type having a medium responsive to electric stimuli to increase or decrease visible transmission, e.g. but not limited to the type of window disclosed in U.S. Published Patent application 2007/0002422A1 and on aircraft windows of the type having an insulated air space between a pair of laminated sheets. The entire disclosure of U.S. Published Patent Application 2007/0002422A1 is hereby incorporated by reference. Further, the invention can be practiced on commercial and residential windows, e.g. but not limited to types disclosed in U.S. Pat. No. 5,675,944, which patent in its entirety is hereby incorporated by reference; a window for any type of land vehicle; a canopy, and windshield for any type of air and space vehicle, a window for any above or below water vessel, and a window for a viewing side or door for any type of containers, for example but not limited to a refrigerator, cabinet and/or oven door.

Shown in FIG. 2 is a non-limiting embodiment of an aircraft intelligent window (hereinafter also referred to as “AIW”) 20, e.g. but not limited to a laminated aircraft windshield 20 that can be used in the practice of the invention. The AIW or windshield 20 includes a first glass sheet 22 secured to a second glass sheet 24 by a first interlayer 26; the second sheet 24 secured to a second vinyl-interlayer or sheet 28 by a first urethane interlayer 30, and the second vinyl-interlayer 28 secured to a heatable member 32 by a second urethane interlayer 34. An edge member or moisture barrier 36 of the type used in the art, e.g. but not limited to a silicone rubber or other flexible durable moisture resistant or impervious material is secured to (1) peripheral edge 38 of the windshield 20, i.e. the peripheral edge 38 of the first and second sheets 22, 24; of the first and second vinyl-interlayers 26, 28; of the first and second urethane interlayers 30, 34 and of the heatable member 32; (2) margins or marginal edges 40 of outer surface 42 of the windshield, i.e. the margins 40 of the outer surface 42 of the first glass sheet 22 of the windshield 20, and (3) margins or marginal edges 44 of outer surface 46 of the windshield 20, i.e. margins of the outer surface 46 of the heatable member 32.

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As is appreciated by those skilled in the art, and not limiting to the invention, the first and second glass sheets **22**, **24**; the first and second vinyl-interlayers **26**, **28** and the first urethane interlayer **30** form the structural part, or inner segment, of the windshield **20** and the outer surface **42** of the windshield **20** faces the interior of the aircraft **14**, and the second urethane layer **34** and the heatable member **32** form the non-structural part, or outer segment, of the windshield **20**, and the surface **46** of the windshield **20** faces the exterior of the aircraft **15**. The heatable member **32** provides heat to remove fog from, and/or to melt ice on, the outer surface **46** of the windshield **20**.

The invention is not limited to the construction of the AIW or windshield **20** and any of the constructions of aircraft transparencies used in the art can be used in the practice of the invention. For example and not limiting to the invention, the windshield **20** can include a construction wherein the vinyl interlayer **28** and the urethane interlayer **30** are omitted, and/or the sheets **22** and **24** are plastic sheets.

Further, the invention is not limited to the design and/or construction of the heatable member **32**, and any electrically conductive heatable member used in the art to heat a surface of a sheet to melt ice on, and/or remove fog from the surface of the sheet can be used in the practice of the invention. In general, the heatable member **32** includes a glass sheet **60** having a conductive coating **62** applied to surface **64** of the glass sheet **60**, and a pair of spaced bus bars **66**, **68** in electrical contact with the conductive coating **62**. The invention is not limited to the composition of the conductive coating **62**, for example and not limiting to the invention the conductive coating **62** can be made from any suitable electrically conductive material. Non-limiting embodiments of conductive coatings that can be used in the practice of the invention include, but are not limited to, a pyrolytic deposited fluorine doped tin oxide film of the type sold by PPG Industries, Inc. under the registered trademark NESAS; a magnetron sputter deposited tin doped indium oxide film of the type sold by PPG Industries, Inc. under the registered trademark NESATRON; a coating made up of one or more magnetron sputter deposited films, the films including, but not limited to a metal film, e.g. silver between metal oxide films, e.g. zinc oxide and/or zinc stannate, each of which may be applied sequentially by magnetron sputtering, e.g. as disclosed in U.S. Pat. Nos. 4,610,771; 4,806,220 and 5,821,001, the disclosures of which in their entirety are hereby incorporated by reference. The invention also contemplates a heatable member **32** using spaced electrically conductive wires. Heatable members to remove ice and snow from the aircraft windshield are well known in the art, and no further discussion is deemed necessary.

Generally the glass sheets **22**, **24** and **60** of the windshield **20** are clear chemically strengthened lithium containing glass sheets; however, the invention is not limited thereto, and the glass sheets can be heat strengthened or heat tempered conventional soda-lime-silicate glass or borosilicate glass sheets. Further as is appreciated, the invention is not limited to the number of glass sheets, vinyl interlayers or urethane interlayers that make up the windshield **20** and the windshield **20** can have any number of sheets and/or interlayers.

Still further, the invention is not limited to the construction of the windshield **20** and any of the constructions of aircraft transparencies used in the art can be used in the practice of the invention. For example and not limiting to the invention, the windshield **20** can include a construction wherein the second vinyl-interlayer **28** and the first urethane interlayer **30** are omitted, and/or the glass sheets **22** and **24**

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are plastic sheets. Further, the cross section of the window **20** shown in FIG. 1 shows flat or non-shaped sheets, the invention is not limited thereto, and the window **20** can have a contour to match the contour of the outer surface of the aircraft in which the window is mounted, e.g. the aircraft **15** shown in FIG. 1.

In addition, the invention is not limited to the material of the layers or sheets of the transparency, and the layers or sheets can be made of, but not limited to, cured and uncured plastic sheets; annealed, heat strengthened, and heat and chemically strengthened, clear, colored, coated and uncoated glass sheets. Still further the invention can be practiced on windows having opaque sheets, e.g. but not limited to wood and metal sheets, and glass sheets having an opaque coating, and combinations thereof. In the preferred practice of the invention, the sheets **22**, **24** and **60** are transparent clear glass sheets. By "clear glass" is meant non-tinted or non-colored glass. The glass sheets can be conventional float glass, and can be of any composition having any optical properties, e.g., any value of visible transmission, ultraviolet transmission, infrared transmission, and/or total solar energy transmission. By "float glass" is meant glass formed by a conventional float process. Examples of float glass processes are disclosed in U.S. Pat. Nos. 4,744,809 and 8,094,942, which patents in their entirety are hereby incorporated by reference.

In the preferred practice of the invention, the AIW or the windshield **20** has one or more sensors to monitor one or more properties of the window to determine if the window is operating within an acceptable performance range, and to take appropriate action based on the performance of the window as indicated by the sensor. In one non-limiting embodiment of the invention, the sensors are selected from the group of (A) an impact sensor **76** that generates a signal when an object hits or impacts the windshield, e.g. but not limiting to the invention, hits the outer surface **48** of the windshield **20** during take off; (B) a rupture or crack detector, or sensor **78** that generates a signal when a portion of the sheet cracks; (C) an arc sensor **80** which generates a signal indicating that the heatable member is arcing which indicates that the heatable member **32** has or is developing a defect; (D) a sensor or detector **82** to measure the temperature of the conductive coating **62** of the heatable member **32** to prevent over heating of the heatable member **32**, and (E) a moisture sensor **84** to indicate moisture penetration through or around the moisture seal **36** and/or between the sheets of the laminated windshield.

The term "aircraft intelligent window" as used herein is an aircraft window having one or more sensors or detectors to measure performance of a property or characteristic of the window and forward a signal having the data to a processor.

The impact sensor **76**, the rupture and crack sensor **78**, the arc sensor **80**, the temperature measuring sensor **82** and the moisture sensor **84** are shown in FIG. 2 as blocks without showing specific designs, components and/or operation of the sensors because the invention is not limited to the design or operation of the sensors **76**, **78**, **80**, **82** and **84**, and sensors well known in the art can be used in the practice of the invention, e.g. sensors that can be used in the practice of the invention are disclosed in detail in U.S. Pat. No. 8,155,816 B2, and U.S. Patent Application Publication Nos. 2010/0163675 A1 and 2013/075,531, and no further discussion is deemed necessary.

Non-limiting embodiments of the invention will be discussed using the right side windshield **10**, the left side windshield **11**, the right front windshield **12**, and the left front windshield **13**, of the aircraft **15**. As can be appreci-

ated, each of the right side windshield **10**, the left side windshield **11**, the right front windshield **12**, and the left front windshield **13**, of the aircraft **15** can have the design of the windshield **20** shown in FIG. **2** and discussed above, or selected ones of the right side windshield **10**, the left side windshield **11**, the right front windshield **12**, and the left front windshield **13**, of the aircraft **15** can have the design of the windshield **20** and the remaining ones of the right side windshield **10**, the left side windshield **11**, the right front windshield **12**, and the left front windshield **13**, of the aircraft **15** can have different designs, e.g. but not limited any prior art design of an aircraft windshield.

The discussion is now directed to non-limiting embodiments of aircraft window sensing networks (hereinafter also referred to as "AWSN") of the invention to receive input from selected ones of the sensors **76**, **78**, **80**, **82** and **84** of the aircraft intelligent windows or windshields **10-13**, and compare the present output of the sensors to previous output of the sensors and/or the output from different sensors of different windows measuring the same parameter to estimate life expectancy or remaining useable life of the aircraft intelligent window for each of the properties measured by the sensors. In the preferred practice of the invention, the aircraft intelligent windshields **10-13** are considered windshields of an aircraft window sensing network (hereinafter also referred to as "AWSN") discussed in detail below instead of individual independent operating windshields. In this manner, individual performance of the windshields can be considered, and performance of the windshield operating as a windshield of a network can be considered. As will be appreciated, monitoring a network of windows instead of one window may provide information to indicate that the aircraft opening or window mount may be contributing to a window defect, or the design of the window may be useable for one location but not for a different location. By way of illustration and not limiting to the discussion, if past performance and present performance of a window in a network of windows that usually develops defects, e.g. moisture penetration is in "position x" of the aircraft, the type of failure of the windows (moisture penetration) in "position x" may be an indication that a window design is not useable in "position x".

In the preferred practice of the non-limiting embodiments of the invention, the sensor **76** provides data relating to impacts to the windshield; the sensor **78** provides data relating to the presence of cracks in the glass and plastic sheets; the sensor **80** provides data relating to arcing of the heatable member **32**; the sensor **82** provides data relating to the temperature of the heatable member **32**, and the sensor **84** provides data relating to moisture penetration. The sensors used in the practice of the invention include, but are not limited to the sensors and detectors disclosed in U.S. Patent Application Publication Nos. 2010/0163675 A1 and 2013/075531.

In one non-limiting embodiment of the invention, each of the aircraft intelligent windows monitored have the same sensors, and in another non-limiting embodiment of the invention, the aircraft intelligent windows have sensors that measure a property that is expected to be effected by the continued operation of the aircraft having the aircraft intelligent window. By way of illustration and not limiting to the invention, aircraft intelligent windows at the rear of the aircraft may not have impact sensors because impacts to the windows at the rear of the plane are less likely to have impacts during take off and landings than the front wind-

shields of the aircraft. The sensors mounted on the aircraft intelligent window are collectively referred to as a group of sensors or sensor group.

Shown in FIG. **3** is a non-limiting embodiment of an aircraft window sensing network or AWSN **85** of the invention. The AWSN **85** includes, but is not limited to the right side windshield **10** having sensor group **86** connected to a window sensing hub ("WSH") **88** by way of cable **90**; the left side windshield **11** having sensor group **92** connected to the WSH **88** by way of cable **94**; the right front windshield **12** having sensor group **96** connected to the WSH **88** by way of cable **98**; the left front windshield having sensor group **100** connected to the WSH **88** by way of cable **102**, and AIW **117** having sensor group **116**. The communication between the WSH **88** and the sensor groups **86**, **92**, **96** and **100** of the AIWs **10-13**, respectively, is provided by cables **90**, **94**, **98** and **102**, respectively, and the sensor group **116** of the windshield **117** is connected to the WSH **88** by a transmitter and antenna combination **124** mounted on the windshield **117**, and a transmitter and antenna combination **126** is positioned in the WSH **88**.

As can now be appreciated by those skilled in the art, passing information between the sensor groups **86**, **92**, **96**, **100**, and **116**, and the WSH **88** can be by wireless transmission as shown for windshield **117** or can be by wire or cable transmission as shown for the AIWs **10-13** as discussed above. Wireless communication and wire communication to pass information between two or more locations or positions is well known in the art and no further discussion is deemed necessary. For additional discussion directed to wireless communication and wire communication to pass information between two or more locations, reference to U.S. Published Patent Publication No 2013/0075531 is recommended. Based on the present discussion, it can now be appreciated that the invention is not limited to the method of passing the information between the sensor groups **86**, **92**, **96**, **100**, **104**, **110** and **116**, and the WSH **88**.

The windshield **117** was added to the AWSN **85** to illustrate a non-limiting embodiment of a wireless connection of the invention. The windshield **117** is not shown in the other non-limiting embodiments of the invention, however, it is understood that wireless connections can be used with other non-limiting embodiments of the invention. Further, unless indicated otherwise, the discussion directed one or more of the IAWs **10-13** is applicable to the IAW **117**.

With continued reference to FIG. **3**, the WSH **88** includes a microprocessor **130** to process the information received from sensor groups **86**, **92**, **96**, **100** and **116**, to determine the performance of properties of each of the windows of interest. In one non-limiting embodiment of the invention, the WSH **88** includes software and data to format the information from the sensors groups **86**, **92**, **96**, **100** and **116** to present a representation of the performance of the AIWs **10-13** and **117**. The invention is not limited to the formatting of the information received by the WSH **88**, e.g. in one non-limiting embodiment of the invention, the data from the sensor groups **86**, **92**, **96**, **100** and **116** received by the WSH **88** is formatted to show performance of the AIWs **10-13** and **117** in the categories of windows performing at the bottom quartile, the top quartile and the middle half of the acceptable range for each of the properties of the AIW being monitored. In another non-limiting embodiment of the invention, the formatting includes assigning windows to networks of windows, e.g. a network of side windshields, and a network of front windshields, and noting and comparing their performance.

As can be appreciated, the invention is not limited to a microprocessor and any equipment for processing information can be used in the practice of the invention, e.g. but not limited to a fully programmable gate array (also known in the art as “FPGA”) and/or an application specific integrated circuit.

In one non-limiting embodiment of the invention, and as shown in FIG. 3, the WSH 88 can be an independent unit used to monitor and control the aircraft window system, or the WSH 88 can be connected to an aircraft central monitoring system 134 (hereinafter also referred to as “aircraft CMS 134”) of the aircraft 15 by wire or cable 136, and/or wireless. The aircraft CMS 134 includes software and historical data to provide expected useable life of the windshields 10-13 and 117, through a mathematical framework and when necessary, set up repair or replacement of the aircraft intelligent window, e.g. but not limited to the procedure disclosed in U.S. Pat. No. 8,155,816. When the windshield sensing hub 88 is used with the aircraft CMS 134, the formatted information is forwarded from the WSH 88 to the aircraft CMS 134 of the aircraft by the cable 136. The aircraft CMS 134 acts on the formatted information to determine health and useable life of the AIW 10-13 and 117, e.g. as discussed in detail below.

As can now be appreciated, the invention is not limited to the number of windows connected to the window sensing hub 88, and all the windows of the aircraft 15 can be connected to the WSH 88. Further, the invention is not limited to the number of sensors provided on each of the AIW 10-13 and 117 of the aircraft 15.

An advantage of the aircraft window sensing network 85 shown in FIG. 3 is the capability of setting up the microprocessor 130 of the WSH 88 to format only data of interest and forwarding the formatted data of interest to the aircraft central monitoring system 134 to determine the health and useable life of the AIWs of interest, e.g. but not limited to windshields 10-13 and 117. The filtered data is stored for reference, e.g. but not limited to the invention to determine expected life expectancy of the AIWs 10-13 and 117 as discussed below. By way of illustration and not limiting to the invention, the microprocessor 130 of the WSH 88 is programmed based on a mathematical model to format the data of interest from the data collected on the activity of the aircraft. For example and not limiting to the discussion. If the plane is scheduled for take off, before take off, data from all the sensor groups 88, 92, 96, 100 and 116 of the aircraft windshields 10-13 and 117 is collected by the aircraft central monitoring system, or aircraft CMS 134 and evaluated by a model based microprocessor 130 to show performance of the aircraft intelligent windshields 12-13 and 117, and the resulting data sent to the aircraft CMS 134 to determine expected useable life of the AIWs 10-13 and 117, if the expected useable life of an AIW 12-13 and 117 is less than the flight time to the initial destination or less than a scheduled flight time to an airport having an AIW to replace the AIW of interest, the AIW of interest is replaced before take off.

In another non-limiting embodiment of the invention, during take off of the aircraft 15, the microprocessor 130 of the WSH 88 is programmed to process data from the impact sensor and the crack sensor of the windshields 10-13 and 117, and to forward the data from the impact sensor and the crack sensor of the windshields to the aircraft CMS 134 to determine any change in the expected useable life of the AIWs 10-13 and 117.

In still another non-limiting embodiment of the invention, during flight time, the microprocessor 130 is programmed to

evaluate data from the sensors at a frequency based on its importance during the flight. For example and not limiting to the discussion, the temperature sensor and the arc sensor are checked at the highest frequency; the crack or rupture sensor and the moisture sensor are measured at a frequency less than the frequency check of the arc and temperature sensors, and the impact sensor is measured at a frequency less than the frequency check of the moisture sensor, and the crack sensor of the AIWs 10-13 and 117.

Further, in another non-limiting embodiment of the invention, during landing of the aircraft, the microprocessor 130 of the WSH 88 is programmed to process data from the impact sensor and the crack sensor and to forward the data from the impact sensor and the crack sensor to the ACMS 134 to be part of the window history database and with a mathematical model to determine any change in the expected useable life of the AIWs 10-13 and 117.

The microprocessor 130 of the WSH 88 can also be programmed to display immediately any sensor reading that has a drop in performance of a predetermined percent, e.g. but not limited to 25% or more drop in performance. Data collected and not used during take-off, flight and landing, is forwarded to a storage facility and used to determine life expectancy as discussed below.

Shown in FIG. 4 is another non-limited embodiment of an aircraft window sensing network of the invention identified by the number 140. The aircraft window sensing network 140 includes, but is not limited to, the windshields 10-13 having sensor groups 86, 92, 96, and 100, respectively, connected to aircraft central monitoring system 142 by the wires 90, 94, 98, and 102, respectively. In this non-limiting embodiment of the invention, the aircraft central monitoring system 142 can include the formatting features of the WSH 88 e.g. but not limited to having the microprocessor 130 positioned in, or being part of, the aircraft central monitoring system 142. The aircraft central monitoring system 142 of the aircraft window-sensing network 140 shown in FIG. 4 operates in a similar manner as the WSH 88 and aircraft CMS 134 of the aircraft window sensing network 85 shown in FIG. 3.

FIG. 5 shows still another non-limiting embodiment of an aircraft window-sensing network (“AWSN”) of the invention designated by the number 150. In general, the aircraft window-sensing network 150 includes, but is not limited to, any combination of single or multiple units, including any or all of the AIW 10-13, and 20 (see FIGS. 1-4), the WSH 88 (see FIG. 3), and/or the aircraft CMS 134 (FIG. 3) and the aircraft central monitoring system 142 (FIG. 4). In the non-limiting embodiment of the invention shown in FIG. 5, the aircraft window sensing network 150 includes, but is not limited to, the sensor group 86 of the right side window 10 and the sensor group 92 of the left side window 11 connected to the WSH 88 by wires 90 and 94, respectively. The sensor 92 of the left front window 13 is connected to the aircraft CMS 134 by cable 102, and the aircraft CMS 134 and the WSH 88 are connected by cable 151 to exchange information between the aircraft CMS 134 and the WSH 88 relating to performance of the windows, e.g. but not limited to aircraft history and maintenance of the aircraft intelligent windows 10, 11 and 13. The aircraft central monitoring system 142 is connected by cable 152 to a window sensing HUB 153 to exchange information between the aircraft central monitoring system 142 and the window sensing hub 153 relating to performance of the windows 10, 11 and 13. With continued reference to FIG. 5, the sensor 96 of the windshield 12 is connected by wire 98 to sensor 156 of the window 20, and the sensors 96 and 156 are connected to the

aircraft central monitoring system **142** by cable **159**. As can be appreciated, the information from the sensor **96** of the window **12** can be integrated with the information of the sensor **156** of the window **20** and forwarded to the aircraft central monitoring system **142** by the cable **159**, or the information from the sensor **96** of the window **12** can be forwarded by cable **98** to the window **20** and forwarded with the information from the sensor **156** by cable **159** to the aircraft central monitoring system **142**.

The AWSN **150** of FIG. **5** provides the option to compare the performance of the AIWs **10** and **11** to one another, and to compare the performance of the AIW **13** to the AIWs **10** and **11** as individual windows or as a group of windows, or to the combination of windows and to compare the performance of the AIW **12** and **20** to one another. Optionally and not limiting to the invention the aircraft central monitoring system **142** can be connected to the aircraft CMS **134** and/or the window sensing hub **153** by wires **160** and **162**, respectively (shown in phantom). In this manner, the performance of the windows can be compared to one another. For example and not limiting to the invention, the output of the arc sensor and temperature sensor for the windows **10**, **11** and **13** can be compared to one another to see if the deterioration of the AIW follows a pattern or is random. If a failure pattern is followed, the failure of the AIW may be due to an effect acting on the window as contrasted to an effect of the window.

As can now be appreciated, the invention is not limited to the connections of the intelligent aircraft windows ("IAW") **10-13** as shown in FIGS. **3-5**, and the invention contemplates any IAW connection configurations, but not limited to IAW directly or indirectly connected to the WSH or the aircraft CMS; IAW inter-connected to another IAW for information sharing or networking, and any combination of IAWs inter-connected first and then connected as a group to either AWH or integrated aircraft CMS. Further, connections of the intelligent aircraft windows **10-13** to the aircraft central monitoring systems **134** and **142** in FIGS. **3-5**, respectively, and the invention contemplates any individual IAW directly integrated into the aircraft central monitoring system to be part of the aircraft maintenance and aircraft reliability calculation/predictions, any combination of IAWs can be inter-connected and/or grouped together and then connected to the aircraft central monitoring system as part of the aircraft history and maintenance recording system, and the Window Sensing Hub (WSH) integrated into the aircraft central monitoring system as part of the central monitoring system for the aircraft monitoring system.

As discussed above, any central monitoring system integration from AIW system to aircraft central monitoring system can be made either by wired connection or wireless connection when acceptable. In the practice of the invention, every AIW can be independently used to shutdown the window heat controller as disclosed in U.S. Published Patent Application Serial No. 2013/075,531 or other window related controllers as disclosed in U.S. Patent Application Publication No 2010/0163675 A1. The invention is not limited to reasons to shut down the sensors of the intelligent aircraft windows **10-13** and **20** (shown only in FIG. **5**, and the windows can be shut down for any reason including, but not limited to, a window operating outside of an acceptable range for the property being monitored, or the sensor of the window is not needed because the property being monitored is not active, e.g. a sensor for impact monitoring can be needed on take off but may not necessarily be needed when the aircraft is in flight.

The non-limited embodiments of the invention provide, among other things, the opportunity to make an intelligent decision for the aircraft control actions, such as shutdown the intelligent aircraft window heat controller, alarm the pilot, data input to the aircraft ACMS based on the window conditions, such as, the window life prediction, window longevity, window arcing condition, window moisture ingress, window impact status, and other window related measurement, will be instituted either by the aircraft ACMS system, by WSH with connection to the aircraft control system input/output modules system, or individually in any AIW with direct or indirect connection to the aircraft control system input/output modules system.

The discussion is now directed to the use of the networks to monitor the performance of the windows having the sensors. The network shown in FIG. **3** uses the WSH as a filter to take all of the information and data provided by the sensor groups to study those properties of interest for the activity of the aircraft. By way of illustration and not limiting to the invention, during take off the information from the impact sensors is monitored to identify any impact having force sufficient to cause a sheet of the windshield to fracture. After take off the output of the crack sensor and the impact sensor are saved into the WSH and evaluated by a math model to determine if a reading is outside of normal low impact. With the plane in flight the monitoring of the impact sensor is reduced to read impact data only if a reading is outside of an acceptable range.

Consider the arcing sensor and the temperature sensor. For planes flying in high temperature zones, the heatable member may have limited use. More particularly, the data from the temperature sensor and the arc sensor can be taken for historical purposes and not processed because the heatable member may be turned off. As can now be appreciated, the output of the arc sensor and the temperature sensor is only used when the data indicates that the heatable member is arcing and/or the temperature sensor indicates a temperature outside an acceptable range.

As can be appreciated, the invention is not limited to the software program to evaluate the performance of the windshield, and to determine and/or estimate useable life of the windshields, and any of the types known and/or used in the art can be used in the practice of the invention Control System

With reference to FIG. **6**, there is shown one non-limiting embodiment of the invention to monitor the performance of the AIW **10-13** and **20** shown in FIG. **5** of the invention and to timely schedule maintenance of, e.g. repairs to, or replacement of, transparencies, e.g. AIWs that are performing outside acceptable limits and/or have a short life expectancy. The sensor groups **86**, **92**, **96**, and **100**, of the IAWs **10-13** of the network **140** shown in FIG. **4** are connected by wires **90**, **94**, **98**, and **102**, respectively to the aircraft central monitoring system **142**, mounted in a console **176** mounted in the airplane **15** (FIG. **1**). The aircraft central monitoring system **142** is connected to a monitor **178** to provide visual display, and speaker **180** to provide audible information regarding the performance of the AIWs **10-13**. The console **176** can include an alarm **182** to bring attention to the monitor **178**. Placing the console **176** in the aircraft **15** provides the personnel within the aircraft **15** with real time performance of the AIW **10-13**

With reference to FIGS. **6** and **7**, in another non-limiting embodiment of the invention, the console **176** has a wireless transmitter and receiver **184**; the transmitter **184** transmits signals **186** to a transmitting tower **188**. The signals **186** carry data on the performance of the AIWs **10-13**. The tower

188 transmits a signal 190 carrying the data on the performance of the AIW 10-13 to a satellite 192. The satellite 192 transmits a signal 194 carrying the data on the performance of the AIWs 10-136 to a control center 196. The data received is studied and the appropriate action to be taken is scheduled. In one non-limiting embodiment of the invention, based on the information received, personnel at the control center 196 determine what action, if any, is needed. If action such as repairs to the AIW or replacement of the AIW is needed, a signal 198 providing a repair schedule is transmitted to the satellite 192. The satellite 192 transmits a signal 200 having the repair schedule to the tower 188. The tower 188 transmits a signal 204 having the repair schedule to the console 176 and to a maintenance center 208 geographically close to the designated repair location (usually the next scheduled stop for the aircraft) to arrange to have all parts, equipment and personal need at the designated repair location.

In still another non-limiting embodiment of the invention, if the data from the sensor groups indicate that an AIW has to be replaced, the repair schedule can include shipment of an AIW replacement windshield to the next scheduled stop of the aircraft; if the AIW has to be replaced with some urgency, the repair schedule would include a change to the flight plan to land immediately and an AIW scheduled to be delivered to the repair area. The passengers can optionally be transferred to another plane or wait until the repair is completed. If a repair is scheduled, and the repair can be made without removing the unacceptable AIW, the repair schedule can provide for personnel and repair parts to be provided at the designated repair location.

As can be appreciated, the invention is not limited to wireless transmission of signals carrying information and the transmission can be made by land-lines. Further, the signals can be transmitted between locations solely by satellite, or solely by transmission towers, and by combinations thereof.

The invention is not limited to the embodiments of the invention presented and discussed above which are presented for illustration purposes only, and the scope of the invention is only limited by the scope of the following claims and any additional claims that are added to applications having direct or indirect lineage to this application.

What is claimed is:

1. A network system for monitoring and storing performance data of more than one transparency of a vehicle, wherein the vehicle comprises one of a land vehicle, an air vehicle, a space vehicle, and a water vessel, to provide estimated useable life of selected ones of the transparencies of the vehicle and to provide real life performance of the selected ones of the transparencies, wherein the selected ones of the transparencies each comprises a transparent sheet having a sensor group co-acting with the transparent sheet to measure predetermined characteristics and/or properties of the transparent sheet of the transparency, wherein the sensor group comprises at least one sensor selected from the family of sensors comprising an arc sensor for measuring arcing of an electrically heatable member mounted on a surface of the transparent sheet, hereinafter referred to as an "arc sensor"; a heat sensor for measuring temperature of the heatable member, hereinafter referred to as a "heat sensor"; a moisture sensor for measuring moisture content on a surface of the transparent sheet, hereinafter referred to as an

"moisture sensor"; an impact sensor for measuring force of impact of objects hitting one or more of the selected ones of the transparencies, hereinafter referred to as an "impact sensor"; and a sensor for identifying fractures in and/or on the transparent sheet, hereinafter referred to as an "crack sensor", the network system comprising:

a window sensing hub comprising a microprocessor programmed to receive output of selected ones of the sensor group of selected ones of the transparencies, wherein the microprocessor is programmed on command to receive data from the sensor group of selected ones of the transparencies, wherein the data provides information on the performance of the property and characteristic of the sensor group of selected ones of the transparencies associated with the sensor group and formatting the received data in accordance to a preset program of the microprocessor, wherein the preset program includes at least providing data from the sensor group of the selected ones of the transparencies that measure characteristics and properties of the transparency that are active during the period in which the data taken is of interest, and

a central monitoring system connected to the microprocessor of the window sensing hub to receive the formatted information from the window sensing hub and raw sensor data, wherein the raw sensor data from the selected ones of transparencies are acted on by the central monitoring system to provide the real life performance of the selected ones of the transparencies and an estimated useable life of the selected ones of the transparencies, wherein the repair and replacement of the selected ones of the transparencies is based on (1) the real life performance of the selected ones of the transparencies and (2) the estimated useable life of the selected ones of the transparencies, wherein the estimated useable life of the transparencies is determined from the real life performance of the selected ones of the transparencies, performance history of the selected ones of the transparencies, and output from sensors of other transparencies of the vehicle, wherein the sensors of the other transparencies measure the same properties and/or characteristics as the at least one sensor of the sensor groups of the selected ones of the transparencies.

2. The system according to claim 1, wherein the selected ones of the transparencies of the vehicle are each aircraft windshields having glass and plastic sheets laminated together.

3. The system according to claim 1, wherein the selected ones of the transparencies are selected from aircraft windows, automotive windows, and windows for underwater vehicles.

4. The system according to claim 3, wherein the transparencies of the aircraft are aircraft windshields.

5. The system according to claim 1, wherein the window sensing hub and/or the central monitoring system is connected to a monitor, an alarm, and/or a voice activated alarm.

6. The system according to claim 1, wherein the selected ones of the transparencies comprise a laminated aircraft windshield and the sensors are mounted between outer surfaces of the laminated windshield and/or on periphery of the laminated windshield.